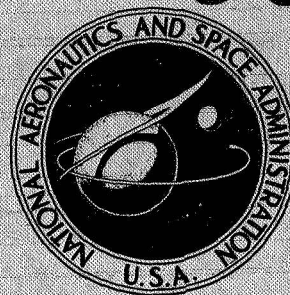


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A COMPUTER PROGRAM FOR CALCULATING
EXTERNAL THERMAL-RADIATION HEAT LOADS
AND TEMPERATURES OF SPACECRAFT
ORBITING THE PLANETS OR THE MOON

by

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NATIONAL AERONAUTICS AND SPACE ADMINISTRATION

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ABSTRACT

A versatile computer program to predict the thermal history of a spacecraft orbiting a celestial body is documented. With this program, all external thermal-radiation heat loads, thin-skin temperatures, or both, are computed for a spinning or oriented spacecraft as a function of orbit position and time. The generalized program applies to any spacecraft configuration. A major feature of the program is its applicability to effects resulting from the extreme surface temperature of the Moon.

Major sections are entitled "Heat-Transfer Theory," "Celestial Mechanics Theory: Coordinate Systems," "Numerical Analysis," "Digital Computer Program," and "Computer Program Application." In addition, sample problems, a complete program listing, and a program user's guide explaining the data input format are included.

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SUMMARY

A computer program to predict the thermal history of a spacecraft in orbit about a celestial body is presented. The program, associated information required for use of the program, and the theory and methods used to develop the program are included. The thermal environment predicted is comprised of all external thermal-radiation heat loads, thin-skin temperatures, or both, for a spinning or oriented space vehicle as a function of orbit position and time. Manned and unmanned satellites and spacecraft or other objects in orbit oriented on the Sun, the Moon, Earth, or a planet other than Earth are within the capability of the program to predict the thermal history of oriented vehicles. A maximum of 200 vehicle surface elements can be analyzed by the computer program. The constant or temperature-dependent thermophysical properties of each element are obtained from one of eight optical-properties tables and from one of eight substrate-properties tables.

The program is generalized so that calculations can be performed for any specified element location. The program does not incorporate details of spacecraft configuration, and the shadowing of one portion of a spacecraft by another spacecraft or by a portion of the same spacecraft is not considered.

A significant feature of the program is the ability to consider the effects of extreme lunar-surface temperature variations on a lunar-orbital spacecraft. The results from a hypothetical lunar-orbital mission confirm the value of this feature of the program by displaying errors of $\pm 100^\circ$ F, on the erroneous assumption that the temperature of the Moon is constant along the surface.

INTRODUCTION

Because of the extreme temperature environment of space and because of the critical limits in the operating temperatures of spacecraft materials, thermal control is an important consideration in the development of spacecraft and components. With the absence of an atmosphere in space, the mode of heat transfer between the spacecraft and its natural environment is thermal radiation. Thus, to insure satisfactory thermal design, a means of accurately determining the spacecraft external radiative environment is required. To determine the environment readily and economically for parametric design analysis, a computer program designed to continuously determine external heat fluxes and temperatures as a vehicle orbits a planet was developed by Midwest Research Institute (MRI) for the NASA Manned Spacecraft Center (MSC) under contract NAS 9-1059. To improve the efficiency, capabilities, and input-output formats of the program, several modifications have been added to the program by NASA MSC. This report is a description of the program developed by MRI with the NASA MSC modifications.

The program, which is documented in the Univac 1108 FORTRAN V language, is a generalized analytical tool capable of determining solar, planetary, and albedo (the solar heat reflected from a planet and its atmosphere) heat fluxes, temperatures, or both. Heat fluxes are obtained for spinning vehicles or for a large number of infinitesimal vehicle surface elements for a vehicle that is planet or sun oriented. To determine the quantity of incident thermal radiation on an elemental area of the vehicle, the angle of incidence is determined first by coordinate transformation. Using the angle of incidence, the configuration factors are obtained from a stored table of radiation-configuration-factor values. The table was developed from a previous study in which the numerical integration of the applicable equations was accomplished. The configuration factors are then used to calculate the component parts of the incident flux. The transient temperatures are calculated by numerical integration of the general differential equation obtained by performing a heat balance about the elemental area. Although the program is capable of analyzing orbits about all planets except Pluto, special emphasis has been placed on problems associated with lunar missions. (Sufficient planetary data are not available to analyze Pluto orbits.) The approach used in analyzing spacecraft heating effects from the extreme surface temperature gradient of the Moon is known, at present, to be applicable only to the Moon. The computer program is written, therefore, so that its use is restricted to lunar orbits. However, if celestial bodies other than the Moon exhibit temperature variations similar to those of the Moon, the program can be modified accordingly, since this part of the program and the discussion within this report are generalized.

SYMBOLS

A	area
a	semimajor axis of orbit ellipse
a_s	semimajor axis of the semiellipse traced by the shadow of the planet

B	boundary on accumulated errors
b	semiminor axis of orbit ellipse
b_s	semiminor axis of the semiellipse traced by the shadow of the planet
C_1, C_2	variable coefficient
C_p	specific heat
c	distance between the center and focus of an ellipse
D	distance between the orbiting vehicle and the center of the planet
DEC	declination (latitude of the Sun with respect to the X_c -, Y_c -, and Z_c -axes)
d	distance between the orbiting vehicle and the planet element
E	eccentric anomaly
E_c, E_p	emissive power, appendix B
E_n	total error caused by numerical integration
e	eccentricity of orbit ellipse
e_{ro}	accumulated effect of round-off error over n steps
$e_{t,a}$	accumulated truncation error over steps 1 to $n-1$
$e_{t,n}$	one-step truncation error (the error caused by approximation over step n)
F_i	radiation configuration factor, $i = 1, 2, 3, 4$
G	gravitational constant
H	altitude
h	vehicle skin thickness
I	intensity of radiation
i	inclination of orbital plane
ℓ	distance between center of areas exchanging radiant energy

M_p	planet mass
P	period (time to complete one orbit)
Q_g	internal heat generation
q	heat flux
R	solar reflectance of celestial body (albedo)
RA	right ascension (longitude of Sun with respect to the X_c -, Y_c -, and Z_c -axes)
r_o	orbit radius
r_p	planet radius
r_s	shadow radius
r_v	radius of a spherical vehicle
S	solar constant
T	temperature
T_m	minimum planet temperature (dark side of planet)
T_p	average planet temperature
t	time
V	volume
X, Y, Z	coordinate axes
α	angle $\leq 180^\circ$ between the planet-Sun line and the X_p -axis
α_p	absorptance of vehicle material with respect to planet-emitted radiation
α_r	absorptance of the receiving surface
α_s	absorptance of vehicle material with respect to solar radiation
β	angle $\leq 180^\circ$ between the planet-Sun line and the Z_p -axis, or the angle between a planet element-Sun line and the planet element normal

γ	angle $\leq 180^\circ$ between the planet-Sun line and the Y_p -axis
$\Delta\theta$	integration step-size as specified in input data; a smaller increment may be chosen by the computer program
δ	angle between the vehicle-Sun line and the vehicle-vehicle element line
ϵ	angle between the vehicle element-vehicle line and the vehicle-planet line
ϵ_e	emittance of emitting surface
ϵ_p	emittance of planet
ϵ_v	emittance of vehicle material
θ	angle between the planet-element line and the planet-vehicle line
θ_s	angle between the Sun-planet line and the planet-vehicle line
Λ'	angle measured from the X_v -axis (towards Y_v) to the projection of the vehicle-vehicle element line on the X_v - Y_v plane
Λ_s	angle measured from the X_v -axis (towards Y_v) to the projection of the Sun-vehicle line on the X_v - Y_v plane
λ	angle between the vehicle-planet line and a vehicle-planet element line
ρ	density
Σ	angle between the X_p -axis and the projection of the Sun-planet line on the X_p - Y_p plane
σ	Stefan-Boltzmann constant
Υ	vernal equinox (an intersection of the Earth equator and the ecliptic)
\emptyset	true anomaly
\emptyset_c	angle between the vehicle-planet-Sun plane and the vehicle element-vehicle-planet plane
\emptyset_{in}	value of \emptyset , where the vehicle passes into the shadow of the planet
\emptyset_{out}	value of \emptyset , where the vehicle passes out of the shadow of the planet

θ_r angle between a normal to dA_r and a line from dA_r to A_e

$$\chi = \frac{X_p}{a}$$

ψ angle shown in figure B-1

Ω longitude of ascending node

Ω' angle between the Z_v -axis and the vehicle-vehicle element line

$\underline{\Omega}$ autumnal equinox (an intersection of the Earth equator and the ecliptic)

Ω_s angle between the Z_v -axis and the Sun-vehicle line

ω argument of perifocus

Subscripts:

c celestial

e emitting body

n specified number of computation steps

o initial condition

p planet

r receiving body

s Sun

v vehicle

ASSUMPTIONS

The formulation of a practical computer program for the calculation of orbital heat fluxes and temperatures requires the use of simplifying assumptions. (These assumptions, none of which appreciably affect the accuracy of the program or the applicability of the program to the intended purpose, are summarized in the following items. It is necessary to evaluate each assumption with respect to application of the program to assure that each assumption is acceptable.

1. Each element is assumed to be thermally isolated from all others; that is, conduction between nodes is not considered. However, the program output allows the

heat loads to each element as a function of time to be loaded conveniently into a heat-conduction program. If desired, the thermal-environment-prediction program may be modified so that the heat loads are punched on cards in a specific format for use as direct input to a heat-conduction program. Programing has been done in FORTRAN V to facilitate such modifications.

2. Conduction and convection between the vehicle and its surroundings are neglected. This assumption is generally valid, since orbits are usually well above any significant atmosphere.

3. Perturbations are neglected. In cases in which perturbations have a significant effect upon the orbit, the mission should be subdivided into two or more segments, and each segment should be run as a separate case. The independently obtained, perturbed orbit parameters in each case are then reentered as new input.

4. The position of the Sun with respect to the orbit is assumed to be fixed. This assumption is reasonable, unless the flight duration becomes an appreciable portion of the planet period. If this occurs, the mission can be subdivided and run as several cases, each with the appropriate coordinates of the Sun specified as input data.

5. The solar constant is assumed to be independent of the vehicle orbit position. Since the semimajor axis of a vehicle orbit is generally small compared to the distance from the Sun to the planet, solar radiation is essentially constant throughout a given orbit.

6. Planet data, such as albedo and mass, are stored internally and are assumed to be invariant. Although these data are essentially constant, the accepted values are occasionally refined. If the program data are to be refined accordingly, a section of the program must be changed and recompiled. This disadvantage of requiring refinement and recompilation is offset by the convenience of not having to include planet properties in the input data each time a case is to be run.

7. The shadow of the planet is assumed to be cylindrically shaped, and penumbral effects are neglected. The method of determining the intersection of the orbit and the shadow is described in appendix A. The resulting errors in Sun-shade point and heat-load determinations are negligible even for large orbits.

8. The thermal radiation discussed in this report is assumed to be diffuse. This assumption is reasonable, except for cases in which the radiation is emitted by a polished vehicle surface.

9. Planet albedo is assumed to be independent of surface position or features. An average value is used because the local values depend on such variables as clouds and other atmospheric conditions that are difficult to predict.

10. Internally generated heat is assumed to be uniformly distributed over the applicable vehicle surface. If this assumption is not reasonable for any case, the internal heat and corresponding surface element should be subdivided until an acceptable assumption is obtained.

11. On the sunlit side, the Moon is assumed to absorb and emit energy as a smooth sphere. With this assumption, the absorbed energy available for emission is proportional to $\cos \beta$, and the lunar-surface temperature is proportional to $\cos^{1/4} \beta$. (A definition of β is provided in appendix B.) Based on a thorough survey of the literature related to the discussion in this report, this assumption was found to be the most realistic and practical approach to handling the extreme temperature gradients over the lunar surface.

12. The temperature T_m of the dark side of the Moon is assumed to be constant. This assumption is considered reasonable, since the heat radiated from the back side of the Moon is less than 1 percent of the heat radiated at the subsolar point, and variations in T_m introduce negligible changes in the heat flux q_p .

13. The vehicle absorptance of reflected solar radiation is assumed to be the same as the vehicle absorptance of direct solar radiation. Although the spectral characteristics of solar radiation are probably changed when solar radiation is reflected, it is believed that this change has little effect on the solar absorptance α_s .

HEAT-TRANSFER THEORY

Fundamental heat-transfer theory has been used to determine the impinging heat loads and the transient surface temperatures of orbiting spacecraft. Heat-transfer theory and equations applicable to the discussion in this report are summarized in the following paragraphs of this section. Detailed derivations are given in appendixes A to C.

Heat Loads

A vehicle orbiting in a vacuum is externally heated by thermal radiation, principally from the planet being orbited and from the Sun. (For convenience, throughout this report, the term "planet" applies to any planet or the Moon.) The amount of heat originating from other celestial bodies is generally negligible. Most of the solar radiation comes directly from the Sun. The remainder is reflected by the planet before impinging upon the vehicle surface and is referred to as albedo heat flux. Solar, planet, and albedo heat fluxes are illustrated in figure 1.

Since an orbiting vehicle may spend a significant portion of each orbit in the shadow of a planet, shielded from both direct and reflected solar radiation, it is necessary to determine the part of the orbit during which the vehicle is shaded. The method used to determine the intersection of the orbit and the shadow is described in appendix A.

The heat emitted by a planet depends on the surface temperature of the planet. If the planet is shrouded with a heavy atmospheric blanket, the surface temperature is relatively uniform because of convection and conductance heat transfer and can be considered constant. It is impractical, if not impossible, to account for temperature

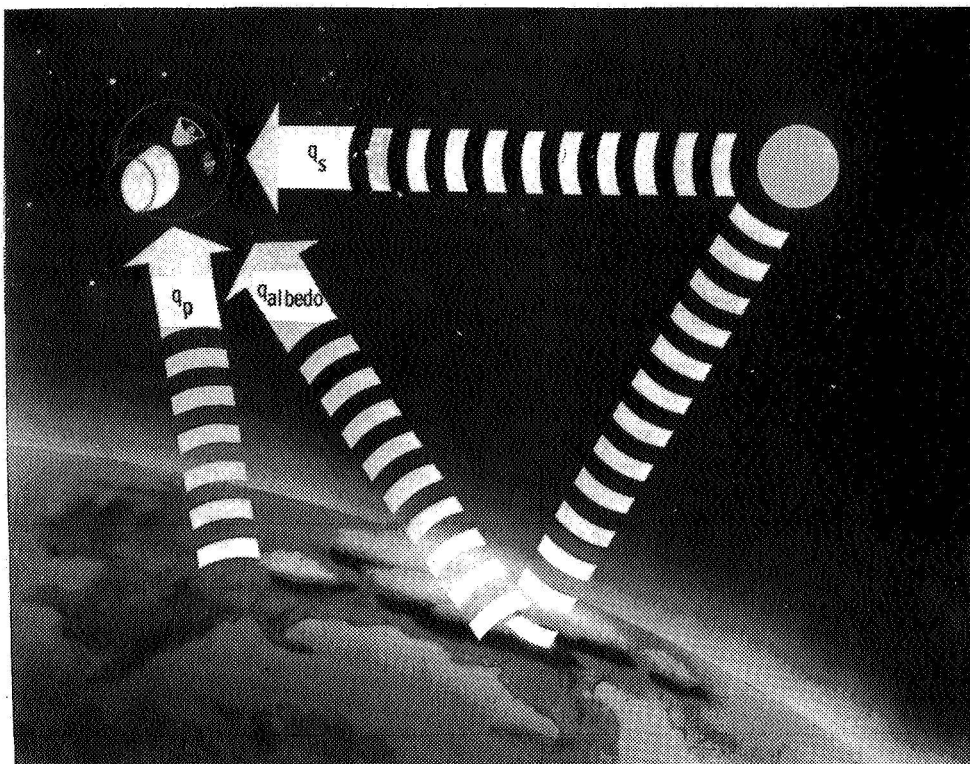


Figure 1. - Principal external heat loads.

deviations from the average, since the deviations are functions of such intangibles as wind, snow, cloud coverage, and atmospheric activity.

If a celestial body (for example, the Moon) has a negligible atmosphere and a nonconducting surface, the surface temperature gradients may be large and should therefore be determined. Fortunately, the factors that cause the extreme variations also make it possible to determine the surface temperature distribution with reasonable accuracy.

External heat loads are summarized for spinning and oriented vehicles as functions of altitude and of the parameters θ_s , Ω' , Λ' , Ω_s , Λ_s , δ , ϵ , and ϕ_c . Complete derivations are given in appendix B.

Radiation impinging on spinning vehicles. - The period of rotation of a spinning vehicle is assumed to be fast enough for the impinging thermal radiation to be uniformly distributed over the vehicle surface. Solar radiation per cross-sectional area A_v striking a spinning vehicle is

$$\frac{q_s}{A_v} = S \quad (1)$$

where the solar constant S is inversely proportional to the square of the distance from the Sun.

Radiation received from a constant-temperature planet ($T_p = \text{constant}$) is given in equation (B16d) of appendix B as

$$\frac{q_p}{A_v} = 2S(1 - R)F_1 \quad T_p = \text{constant} \quad (2a)$$

where

$$F_1 = \frac{1 - \left[1 - \left(\frac{r_p}{D} \right)^2 \right]^{1/2}}{4} \quad (2b)$$

The corresponding heat rate for a variable-temperature planet ($T_p \neq \text{constant}$) is expressed in equation (B30d) as

$$\frac{q_p}{A_v} = 8S(1 - R)F_2 \quad T_p \neq \text{constant} \quad (3a)$$

where

$$F_2 \approx \left\{ \frac{1 - \left[1 - \left(\frac{r_p}{D} \right)^2 \right]^{1/2}}{4} \right\} \cos \theta_s \quad (3b)$$

Equation (3b) closely approximates the exact radiation configuration factor, as demonstrated in figure B-5 of appendix B.

For the condition when $\theta_s > 90^\circ$, F_2 is set equal to zero by the computer program, since the vehicle is in the planet shadow and, therefore, the sun does not emit radiation to the vehicle.

The temperatures of the dark sides of variable-temperature planets are low, but the temperatures are high enough for the emission of some heat. According to equation (B33), this emitted heat can be expressed by

$$\frac{q_p}{A_v} = 8\sigma T_m^4 F_1 \quad T_p \neq \text{constant (dark side of planet)} \quad (4)$$

where F_1 is defined in equation (2b), σ is the Stefan-Boltzmann constant, and T_m is the average dark-side or minimum planet temperature. The value of T_m for the Moon (186° R) is stored internally in the computer program.

The albedo heat flux impinging on a spinning vehicle is given in equation (B38d) as

$$\frac{q_{\text{albedo}}}{A_v} = 8\text{SR}F_2 \quad (5)$$

where F_2 is defined in equation (3b).

Radiation impinging on oriented vehicles. - An object is planet oriented if the line connecting the vehicle and planet center passes through the same vehicle surface element for all positions in the orbit path. Similarly an object is Sun oriented if the vehicle-Sun line passes through the same surface element at all times.

The incident heat flux on an oriented vehicle can vary from one surface position to another. Therefore, expressions for the heat flux to any surface element must be derived. The equations developed in appendix B are applicable to either Sun-oriented or planet-oriented vehicles, provided the required independent variables (for example, δ) are defined for each orientation, as described in the section of this report entitled "Celestial Mechanics Theory: Coordinate Systems." The following heat-flux expressions are developed for a typical surface element located with respect to the vehicle coordinate system by the angles Λ' and Ω' .

Solar radiation impinging on a surface element is

$$\frac{q_s}{A_v} = S \cos \delta \quad \delta \leq 90^\circ \quad (6a)$$

If δ is greater than 90° (that is, $\cos \delta < 0$), the element does not receive radiation from the Sun, therefore

$$\frac{q_s}{A_v} = 0 \quad \delta > 90^\circ \quad (6b)$$

The heat flux from a constant-temperature planet is given in equation (B46c) as

$$\frac{q_p}{A_v} = \frac{S(1 - R)F_4}{4} \quad T_p = \text{constant} \quad (7)$$

where F_4 is a function of altitude H and the variables θ_c and ϵ , which are illustrated in figure 2. Values of F_4 have been evaluated by numerical integration over the applicable ranges of θ_c and ϵ and to an altitude for which the radiation configuration factor, and therefore the planet heat, is negligible (an altitude of approximately five planet radii). (For all values of θ_c and

ϵ , F_4 is set equal to zero when $F_4(H \approx 5r_p)/F_4(H=0) \leq 0.015$). A table of over 2500 values representing the function $F_4 = F(\theta_c, \epsilon, H)$ has been incorporated into the permanent data deck of the computer program.

The heat flux from a variable-temperature planet is given in equation (B42c) as

$$\frac{q_p}{A_v} = S(1 - R)F_3 \quad T_p \neq \text{constant} \quad (8)$$

where F_3 is a function of H , θ_c , ϵ , and θ_s . The variable θ_s is the angle between the vehicle, the planet, and the Sun, as shown in figure 2. In appendix B, the relationship that exists between F_3 and F_4 is described in detail.

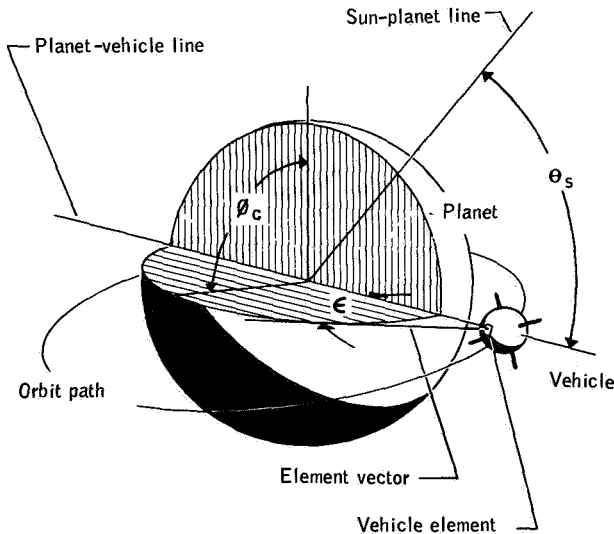


Figure 2. - Angular variables for determining radiation configuration factors of oriented vehicles.

The impinging heat from the dark side of the variable-temperature planet is given in equation (B47) as

$$\frac{q_p}{A_v} = \sigma T_m^4 F_4 \quad T_p \neq \text{constant (dark side of planet)} \quad (9)$$

The albedo heat flux irradiating an oriented-vehicle element is expressed in equation (B50c) as

$$\frac{q_{\text{albedo}}}{A_v} = \text{SRF}_3 \quad (10)$$

Transient Temperatures

The transient temperature of an orbiting vehicle is obtained by conducting a heat balance on the vehicle and by solving the resulting differential equation. For this analysis, each surface element of an oriented vehicle is considered to be thermally isolated from all other elements.

The general governing differential equation is

$$\frac{dT}{dt} = \frac{1}{\rho C_p V} (q_s \alpha_s + q_{\text{albedo}} \alpha_s + q_p \alpha_p + q_{\text{internal}} - q_{\text{out}}) \quad (11)$$

where ρ and C_p are the density and the specific heat, respectively, of the vehicle material and V is the volume being analyzed. The term q_{internal} represents the amount of internally generated heat that is absorbed by the vehicle skin. It is assumed that q_{internal} is dissipated uniformly over the entire external area of the vehicle if it is spinning or over the surface element if the spacecraft is planet or Sun oriented. Internally generated heat is expressed as

$$q_{\text{internal}} = Q_g \quad (12)$$

The negative term q_{out} in equation (11) is the heat emitted by the vehicle and is given by

$$\frac{q_{\text{out}}}{A_v} = \sigma \epsilon_v T^4 \quad (13)$$

where T is the instantaneous surface temperature, ϵ_v is the emittance of the material at temperature T , and A_v is the total emitting-surface area. Introducing the appropriate heat terms into equation (11) gives the governing temperature differential equations as follows. For a spinning vehicle orbiting a constant-temperature planet

$$\frac{dT}{dt} = \frac{1}{\rho C_p h} \left[\frac{S \alpha_s}{4} + 2SR \alpha_s F_2 + \frac{S(1-R) \alpha_p F_1}{2} + Q_g - \sigma \epsilon_v T^4 \right] \quad (14a)$$

where h is skin thickness and α_s and α_p are the solar and planet absorptance of the material, respectively.

For a spinning vehicle orbiting a variable-temperature planet

$$\frac{dT}{dt} = \frac{1}{\rho C_p h} \left[\frac{S \alpha_s}{4} + 2SR \alpha_s F_2 + 2S(1-R) \alpha_p F_2 + Q_g - \sigma \epsilon_v T^4 \right] \quad (14b)$$

or

$$\frac{dT}{dt} = \frac{1}{\rho C_p h} \left(\frac{S \alpha_s}{4} + 2SR \alpha_s F_2 + 2\sigma \alpha_p T_m^4 F_1 + Q_g - \sigma \epsilon_v T^4 \right) \quad (14c)$$

depending on whether the vehicle is above the sunlit or the dark side of the planet, respectively. Near the terminator, the larger value of dT/dt from equations (14b) and (14c) is used.

For an oriented vehicle orbiting a constant-temperature planet

$$\frac{dT}{dt} = \frac{1}{\rho C_p h} \left[S \alpha_s \cos \delta + S R \alpha_s F_3 + \frac{S(1-R) \alpha_p F_4}{4} + Q_g - \sigma \epsilon_v T^4 \right] \quad (14d)$$

For an oriented vehicle orbiting a variable-temperature planet

$$\frac{dT}{dt} = \frac{1}{\rho C_p h} \left[S \alpha_s \cos \delta + S R \alpha_s F_3 + S(1-R) \alpha_p F_3 + Q_g - \sigma \epsilon_v T^4 \right] \quad (14e)$$

or if the oriented vehicle is above the shaded half of the planet

$$\frac{dT}{dt} = \frac{1}{\rho C_p h} \left(S \alpha_s \cos \delta + S R \alpha_s F_3 + \sigma T_m^4 \alpha_p F_4 + Q_g - \sigma \epsilon_v T^4 \right) \quad (14f)$$

Near the terminator, equations (14e) and (14f) are both evaluated, but the smaller value of dT/dt is discarded.

Equations (14a) to (14f) are of the form

$$\frac{dT}{dt} = C_1 - C_2 T^4 \quad (15)$$

where C_1 and C_2 are variable coefficients. Both t and θ must be known in order to evaluate C_1 and C_2 ; thus, an expression relating t and θ is required. The necessary relationship is derived by first writing t as a function of the eccentric anomaly E in accordance with Kepler's laws of planetary motion.

$$t = \frac{(E - e \sin E)P}{2\pi} \quad (16)$$

where t and E are measured from perigee. The period P is

$$P = 2\pi \left(\frac{a^3}{GM_p} \right)^{1/2} \quad (17)$$

where G is a gravitational constant and M_p is the planet mass.

Time can now be expressed as a function of θ by combining equations (16) and (22).

$$t = \frac{2 \tan^{-1} \left[\left(\tan \frac{\theta}{2} \right) \left(\frac{a-c}{b} \right) \right] - e \sin \left\{ 2 \tan^{-1} \left[\left(\tan \frac{\theta}{2} \right) \left(\frac{a-c}{b} \right) \right] \right\}}{2\pi P^{-1}} \quad (18)$$

Equation (18) can be solved readily for t ; however, an iterative solution is required in order to obtain the solution $\theta = f(t)$. Consequently, θ has been selected as the independent variable. This is indicated in equation (15) as

$$\frac{dT}{dt(\theta)} = C_1 - C_2 T^4 \quad (19)$$

Equation (19) does not yield a closed-form solution, but can be solved numerically. The procedure for numerically integrating equation (19) is described in the section of this report entitled "Numerical Analysis."

CELESTIAL MECHANICS THEORY: COORDINATE SYSTEMS

Four locations must be specified for the determination of vehicle heat loads:

1. The location of each vehicle surface element being analyzed (not applicable to spinning satellites)
2. The location of the vehicle with respect to the planet being orbited
3. The celestial location of the vehicle
4. The location of the Sun with respect to the planet being orbited

The required locations can be obtained in terms of vehicle, planet, and celestial coordinate systems. These systems are identified throughout this report by the following notation:

1. Vehicle coordinates — X_v , Y_v , and Z_v
2. Planet coordinates — X_p , Y_p , and Z_p
3. Celestial coordinates — X_c , Y_c , and Z_c

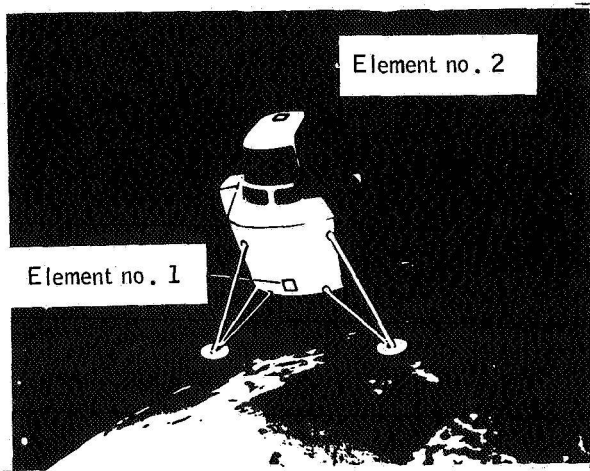
The three coordinate systems are summarized in table D-I of appendix D.

Vehicle Coordinates

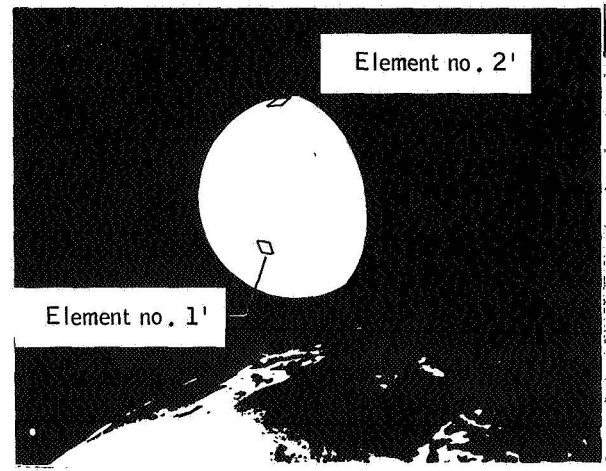
For a spinning vehicle, no distinction is required between surface elements; and coordinates X_v , Y_v , and Z_v are not applicable, since it is assumed that the vehicle is spinning fast enough that impinging thermal radiation is uniformly distributed over the vehicle surface.

For an oriented vehicle, however, the incident heat flux can vary considerably from one surface position to another. For example, element no. 1 on the planet-oriented vehicle shown in figure 3 receives heat emitted by the planet, and element no. 2 does not receive heat emitted by the planet. Consequently, a coordinate system for the location of each element being analyzed is required for the thermal analysis of oriented spacecraft. The vehicle coordinates used are illustrated in figure 3. The complex configuration shown in figure 3(a) is first replaced with a spherical mathematical model, as shown in figure 3(b). Surface element no. 1' and surface element no. 2' on the sphere are selected so that they have the same space orientation as vehicle element no. 1 and vehicle element no. 2. If the orbiting body is planet oriented, surface positions on the sphere are defined with respect to the coordinate system illustrated in figure 3(c). The origin is at the center of the sphere, and the X_v -axis is directed toward the planet center of mass. The Y_v -axis is at right angles to the X_v -axis in the orbital plane, with the positive direction opposite the vehicle velocity vector. The direction of vehicle travel is always in the same direction as the movement when the X_p -axis is rotated into the Y_p -axis through the smallest angle. The Z_v -axis is normal to the orbital plane in a direction such that X_v , Y_v , and Z_v form a right-handed coordinate system. Surface elements are defined by the angles Λ' and Ω' , as shown in figures 3(c) and 3(d). The angle Λ' is measured from the X_v -axis (toward Y_v) to the projection of the element on the X_v - Y_v plane ($0^\circ \leq \Lambda' < 360^\circ$); Ω' is measured from the Z_v -axis to the element ($0^\circ \leq \Omega' \leq 180^\circ$).

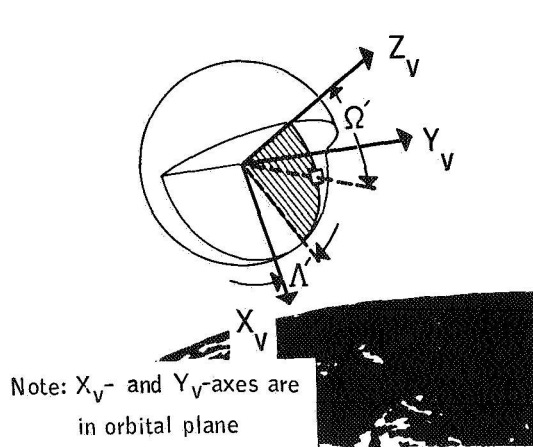
The technique of locating elements on the vehicle with the two angles Λ' and Ω' can be likened to the method used to locate a point on Earth by using the two angles of longitude and latitude. If the orbiting body is Sun oriented, surface points or features are located with respect to the system shown in figure 3(d). The origin is again at the center of the sphere, but the X_v -axis is directed toward the Sun. The Y_v -axis is in the orbital plane at right angles to X_v . The Z_v -axis is always in the same hemisphere as the Z_p -axis and completes the orthogonal system. For the special case in which the projection of the X_v -axis on the orbital plane is a point, the vehicle coordinate system should be set up as described previously. However, to eliminate ambiguity, Y_v is chosen to be in the same direction as Y_p . Surface elements are described by the coordinates Λ' and Ω' , as before.



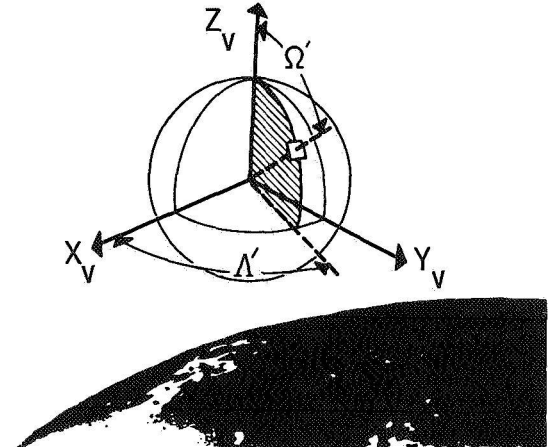
(a) Typical vehicle.



(b) Mathematical model of typical vehicle.



(c) Mathematical model (planet oriented).



(d) Mathematical model (Sun oriented).

Figure 3. - Vehicle coordinate system for typical spacecraft.

Planet Coordinates

Before computing the impinging heat loads emitted or reflected by the planet, the location of the vehicle with respect to the planet being orbited must be specified. The orbits and planets shown in figures 4(a) and 4(b) are the same. However, in figure 4(b), the orbit is shown rotated into the plane of the page. Superimposing the X_p - and Y_p -axes (planet coordinates) onto the orbital plane simplifies the equations of motion by reducing the problem from three dimensions to two. The position of the vehicle can be defined in the two-dimensional system by either polar or Cartesian coordinates.

The polar coordinates of a point on the orbit are the true anomaly θ and the distance D from the principal focus, or center of force, to the vehicle. The distance is given by

$$D = a(1 - e \cos E) \quad (20)$$

where a is the semimajor axis, e is the eccentricity of the orbit, and E is the eccentric anomaly. The variables e and E are formulated as

$$e = \frac{(a^2 - b^2)^{1/2}}{a} = \frac{c}{a} \quad (21)$$

and

$$E = 2 \tan^{-1} \left[\left(\tan \frac{\theta}{2} \right) \left(\frac{a - c}{b} \right) \right] \quad (22)$$

where b is the semiminor axis and c is the distance between the center and the focus of the ellipse.

The Cartesian coordinates X_p and Y_p of a point on the orbit are expressed as

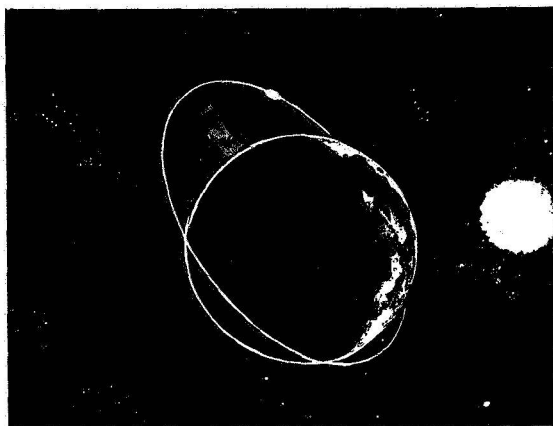
$$X_p = a(\cos E - e) \quad (23)$$

and

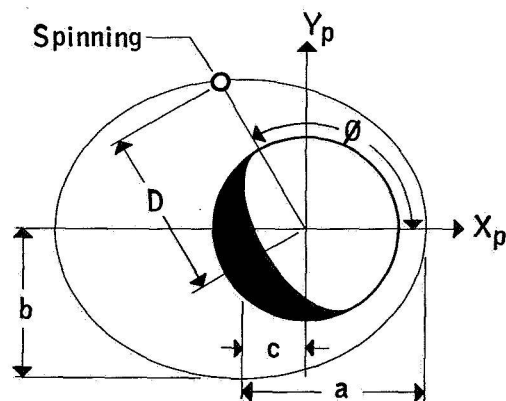
$$Y_p = a(1 - e^2)^{1/2} \sin E \quad (24)$$

Vehicle coordinates for a planet-oriented orbiting body superimposed on the planet coordinate system are shown in figure 4(c). As the vehicle makes one counter-clockwise revolution about the planet, the X_v - and Y_v -axes are similarly rotated about the Z_v -axis.

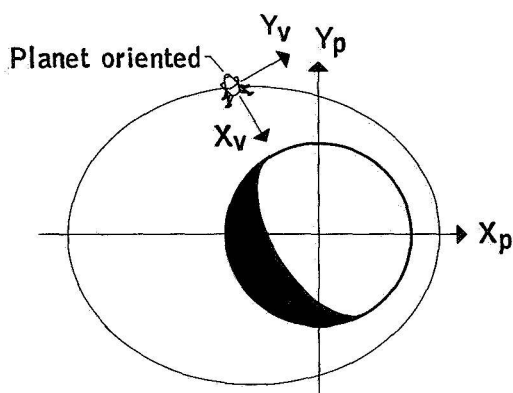
The vehicle and planet coordinates for Sun-oriented satellites are illustrated in figure 4(d). The X_v - Y_v plane is generally not in the X_p - Y_p plane, since X_v is directed toward the Sun. The vehicle coordinates have a fixed orientation in space



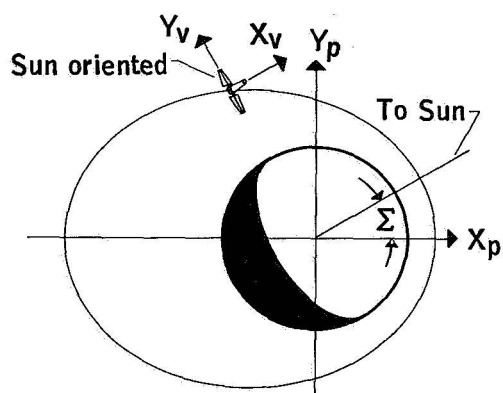
(a) Orbit and planet.



(b) Spinning vehicle.



(c) Planet-oriented vehicle.



(d) Sun-oriented vehicle.

Figure 4. - Planet coordinate systems for spinning and oriented vehicles.

(assuming the planet to be stationary with respect to the Sun, which is essentially true during the time it takes to complete several orbits).

Celestial Coordinates

To determine the contribution of the Sun to spacecraft heating, the celestial location of the vehicle with respect to the celestial coordinates X_c , Y_c , and Z_c must be specified, but first, two astronomical terms, ecliptic and vernal equinox, will be introduced.

Positions in the solar system are commonly defined with respect to the ecliptic and the vernal equinox. The ecliptic is the plane described by Earth as it orbits the Sun. The vernal equinox, symbolized by Υ , is a fixed line from the Earth, directed

toward the Sun at the instant the Sun crosses the Earth equator from south to north in its apparent annual motion along the ecliptic. The direction of the vernal equinox varies slightly (approximately 50.27 seconds of arc per year). Hence, to specify the coordinates of an object, it is necessary to state which vernal equinox is referred to as the principal direction (for example, the equinox of 1950 or 1965).

Thus, the Earth intersects the vernal equinox at the start of spring (in the Northern Hemisphere). The vernal equinox also lies along the line of nodes described by the intersection of the Earth equatorial plane with the ecliptic. The other point at which the Sun crosses the Earth equator, going from the Northern to the Southern Hemisphere, is designated by the symbol $\bar{\Omega}$ and is the beginning of autumn in the Northern Hemisphere (autumnal equinox). At both points, day and night are equal.

The orbit of a vehicle about Earth, a planet other than Earth, or the Moon is conventionally related to the celestial X_c -, Y_c -, and Z_c -axes of the planet by three angles Ω , ω , and i . Before these functions can be defined, it is necessary to identify the geometric relationship between the vehicle orbit and the celestial axes. In figure 5(a), the general orbit of a vehicle is shown, with the intersection of the orbital plane and the planet surface shown in a broken line. The other broken line represents the intersection of the plane containing the celestial X_c - and Y_c -axes with the planet surface. The broken lines cross at two points. (Only one point is visible in fig. 5(a).) The two points define the line of intersection of the orbital plane and the plane containing the X_c - and Y_c -axes. The two intersections are commonly referred to as the ascending node and the descending node, and the line connecting the nodes is called the line of nodes. The vehicle is at the ascending node when it is passing upward (north) through the X_c - Y_c plane.

The orbit and the X_c -, Y_c -, and Z_c -axes are related by Ω , ω , and i , as shown in figure 5(b). The angle Ω is the longitude of the ascending node, measured counter-clockwise in the X_c - Y_c plane from X_c to the line of nodes. The angle ω is the argument of perifocus, measured in the orbital plane in the direction of travel from the ascending node to the perigee. The angle i represents the true inclination between the X_c - Y_c plane and the orbital plane. The axes of the orbital plane are represented by X_p , Y_p , and Z_p , where X_p is directed through the perigee of the orbit, as previously defined.

Any number of reference X_c -, Y_c -, and Z_c -axes can be chosen. The primary consideration in choosing celestial coordinate systems for the analysis in this report was that the systems should be compatible with standard astronomical references in order to minimize input data compilation time and effort by the program user. The geocentric, modified heliocentric, and selenographic systems were selected to describe orbits about Earth, a planet other than Earth, and the Moon, respectively.

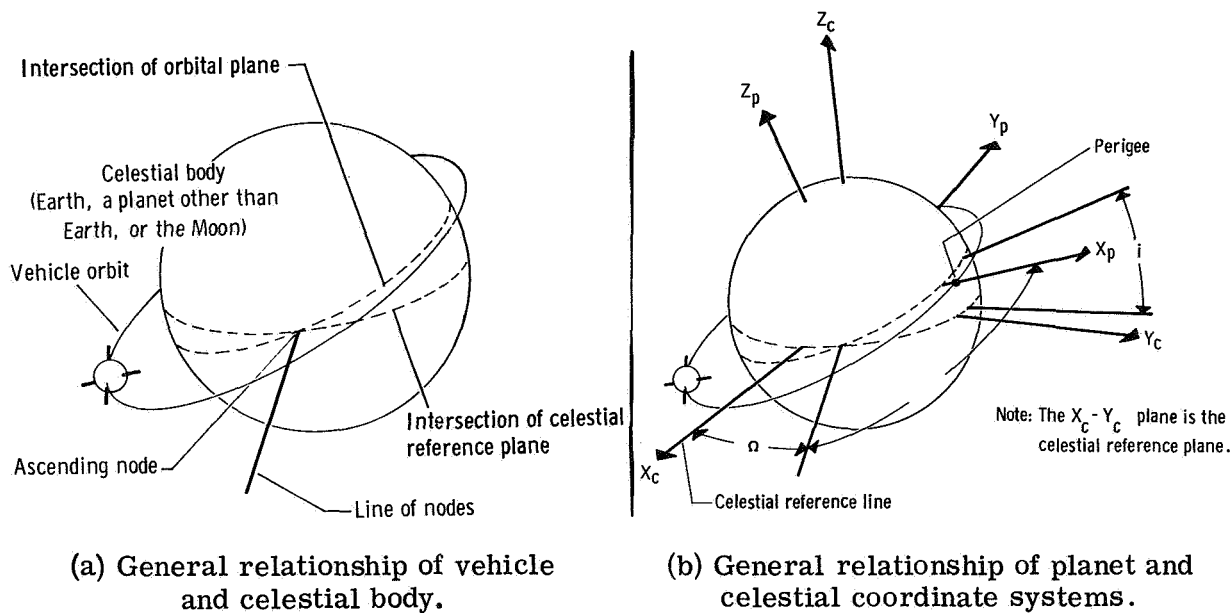


Figure 5. - Relationship between orbit and general celestial coordinate systems.

Conventional geocentric coordinates are employed to define orbits about Earth. In this system, the X_c - Y_c reference plane is the plane of the Earth equator, with the X_c -axis directed along the vernal equinox, as shown in figure 6.

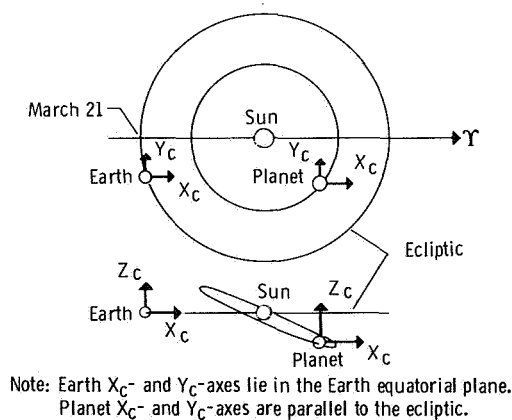


Figure 6. - Celestial coordinate system for Earth (geocentric) and other planets (modified heliocentric).

A modified heliocentric coordinate system is used to define orbits about planets other than Earth. Since the equatorial planes of all planets are not well defined, the X_c - Y_c reference plane has been chosen parallel to the ecliptic, with the X_c -axis directed parallel to the vernal equinox. The X_c -, Y_c -, and Z_c -axes for planets are also illustrated in figure 6.

Conventional selenographic coordinates are employed to define orbits about the Moon. In this system, the X_c - Y_c reference plane is the plane of the Moon equator, and the Z_c -axis extends through the north pole of the Moon (fig. 7). The positive direction of the X_c -axis is from the center of the Moon out through the prime meridian of the Moon. By definition, the prime meridian

of the Moon passes through the mean center of the Moon. The mean center is the point on the lunar surface that is directed toward the center of the Earth when the Moon is at the mean ascending node, and the node coincides with the mean perigee or mean apogee. The longitude is measured as positive toward the west, as seen by an observer on Earth, or in a counter-clockwise direction from the X_c -axis (toward Mare Crisium). The direction of the X_c -axis in space is not fixed, but revolves with the Moon, making one turn every 28 days.

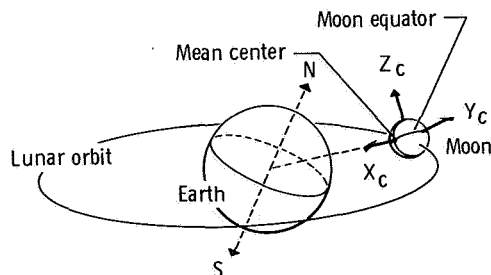


Figure 7. - Celestial coordinate system for the Moon (selenographic).

Position of the Sun

The location of the Sun with respect to the planet being orbited must be specified to complete the definition of Sun, planet, and vehicle relationships. The position of the Sun can be expressed with respect to the celestial body being orbited (in X_c , Y_c , and Z_c coordinates) in terms of right ascension RA and declination DEC, as shown in figure 8. For Earth, RA and DEC can be obtained directly from an ephemeris (ref. 1) for each day of the year. For the Moon, the required coordinates are listed in the ephemeris as latitude and colongitude, where colongitude is

$$\text{colongitude} = 90^\circ - \text{longitude}$$

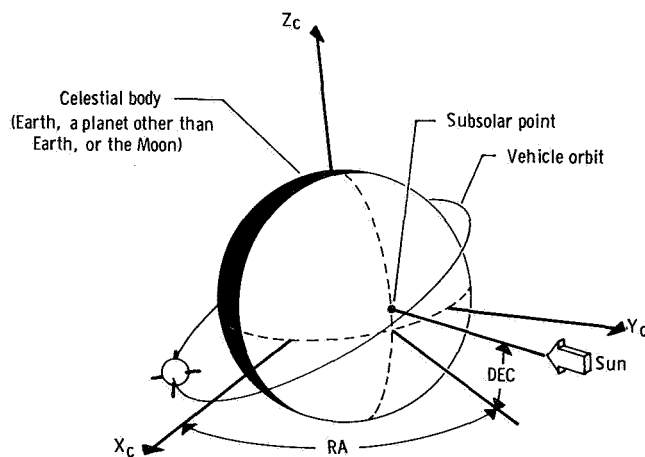


Figure 8. - Position of the Sun relative to the celestial coordinate system.

Longitude must always be positive. Therefore, if colongitude is greater than 90° , the equality becomes $\text{colongitude} = 450^\circ - \text{longitude}$.

For a planet other than Earth, the right ascension and declination of the planet with respect to the Sun are given in heliocentric coordinates in reference 1. The position of the Sun with respect to the modified heliocentric coordinate system can be obtained from these data, as demonstrated in the following examples. If the declination of the planet with respect to the Sun is $+10^\circ$, the declination of the Sun with respect to the planet is -10° . If the right ascension or longitude of the Sun with respect to the planet is 236° , the required value is 56° ($236^\circ + 180^\circ = 416^\circ$; $416^\circ - 360^\circ = 56^\circ$).

The position of the Sun can be expressed with respect to the planet coordinate X_p -, Y_p -, and Z_p -axes in terms of α , β , and γ as shown in figure 9. The coordinates RA and DEC can be transformed into α , β , and γ by performing rotations through the angles Ω , ω , and i , as shown in figures 8 and 9. In the first transformation, the X_c -axis is rotated about the Z_c -axis through an angle Ω into the line of nodes (fig. 5). In the second transformation, the Y_c -axis is rotated about the X_c -axis through the angle i . Finally, the X_c -axis is rotated through an angle ω about the Z_c -axis.

Performing the two transformations gives the following relationships:

$$\begin{aligned}\cos \alpha = & [(\cos \omega)(\cos \Omega) - (\sin \omega)(\sin \Omega)(\cos i)] (\cos RA)(\cos DEC) \\ & + [(\cos \omega)(\sin \Omega) + (\sin \omega)(\cos \Omega)(\cos i)] (\sin RA)(\cos DEC) \\ & + (\sin \omega)(\sin i)(\sin DEC)\end{aligned}\quad (25)$$

$$\begin{aligned}\cos \gamma = & [(-\sin \omega)(\cos \Omega) - (\cos \omega)(\sin \Omega)(\cos i)] (\cos RA)(\cos DEC) \\ & + [(-\sin \Omega)(\sin \omega) + (\cos \Omega)(\cos \omega)(\cos i)] (\sin RA)(\cos DEC) \\ & + (\cos \omega)(\sin i)(\sin DEC)\end{aligned}\quad (26)$$

$$\begin{aligned}\cos \beta = & (\sin \Omega)(\sin i)(\cos RA)(\cos DEC) \\ & + (-\cos \Omega)(\sin i)(\sin RA)(\cos DEC) + (\cos i)(\sin DEC)\end{aligned}\quad (27)$$

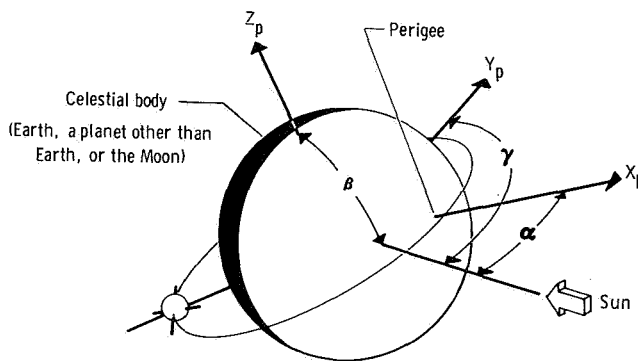


Figure 9. - Position of the Sun relative to the planet coordinate system.

For convenience, the derivations of equations (25) to (27) have been omitted.

After the position of the Sun with respect to the planet and the orbital plane has been obtained, the final step is to determine the position of the Sun with respect to the orbiting spacecraft. In the following paragraphs, the position of the Sun is determined with respect to spinning, planet-oriented, and Sun-oriented vehicles.

Position of the Sun with respect to spinning vehicles. - Some of the solar energy that strikes the planet and the atmosphere of the planet is scattered back into space and impinges on the orbiting vehicle. This radiation (albedo) travels

through the angle θ_s (fig. 2). The sum of the direction cosines of the planet-vehicle line times the sum of the direction cosines of the Sun-planet line is equal to $\cos \theta_s$, which reduces to

$$\cos \theta_s = \frac{X_p}{D} \cos \alpha + \frac{Y_p}{D} \cos \gamma \quad (28)$$

Solar energy directly irradiating the vehicle is the same throughout the orbit (except when the vehicle is shaded by the planet) if the assumption is made that the period of rotation of the spinning sphere is short (that is, that all surface elements continuously receive radiation from the Sun), and if the assumption is made that the Sun is a point source infinitely far away. Therefore, no other angles relating the position of the Sun with respect to the vehicle are required.

Position of the Sun with respect to planet-oriented vehicles. - The relative positions of the Sun, vehicle, vehicle element, and planet are defined by the angles Ω_s , Λ_s , δ (fig. 10), and θ_s (fig. 2). The function $\cos \theta_s$ for a planet-oriented vehicle is the same as $\cos \theta_s$ for a spinning vehicle and is given in equation (28).

Since Z_v is parallel to Z_p and since it is assumed that the Sun is a point source infinitely far away, it follows that

$$\Omega_s = \beta \quad (29)$$

It can also be shown that

$$\Lambda_s = \Sigma - \phi + \pi \quad (30)$$

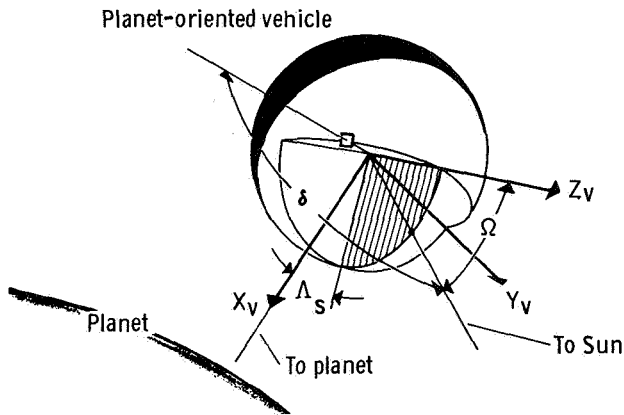


Figure 10. - Position of the Sun relative to the vehicle coordinate system (planet oriented).

where Σ is the true anomaly of the solar projection of the X_p - Y_p plane (fig. 4d). If Λ_s is negative, 2π radians must be added to Λ_s . However, only $\sin \Lambda_s$ and $\cos \Lambda_s$ are required by the analysis in this report; therefore, equation (30) need not be modified for negative values.

The angle δ between the element-vehicle line and the Sun-vehicle line is derived by adding the products of the corresponding direction cosines of the two lines, which gives

$$\begin{aligned} \cos \delta = & \sin \Omega' \cos \Lambda' \sin \Omega_s \cos \Lambda_s \\ & + \sin \Omega' \sin \Lambda_s \sin \Omega_s \sin \Lambda_s + \cos \Omega' \cos \Omega_s \end{aligned} \quad (31)$$

Position of the Sun with respect to Sun-oriented vehicles. - The angle θ_s is independent of vehicle orientation; therefore, $\cos \theta_s$ for a Sun-oriented vehicle is the same as $\cos \theta_s$ for a spinning vehicle, which is given in equation (28).

By definition, the X_v -axis is directed toward the Sun; therefore

$$\Omega_s = \frac{\pi}{2} \quad (32)$$

and

$$\Lambda_s = 0 \quad (33)$$

as shown in figure 11.

The angle δ between the element-vehicle line and the Sun-vehicle line can be derived by combining equations (31), (32), and (33) to obtain

$$\cos \delta = \sin \Omega_s \cos \Lambda_s \quad (34)$$

NUMERICAL ANALYSIS

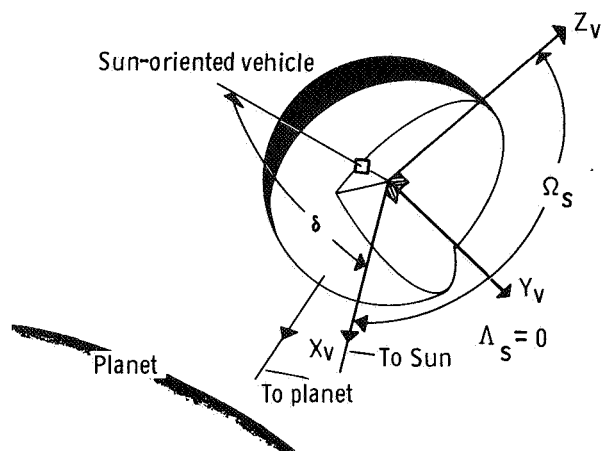


Figure 11. - Position of the Sun relative to the vehicle coordinate system (Sun oriented).

It is important that numerical techniques be both accurate and fast. The accuracy and speed of numerically integrating an equation such as the general thermal expression given in equation (19) are related to the specific method and step-size to be used and to the complexity of the function. Of course, accuracy also depends on proper application of the numerical technique employed.

Method of Integration

Multistep methods require a constant step-size. Although they are generally faster than single-step methods, their use for this specific problem would necessitate a time-consuming iterative

solution of equation (18) to find the true-anomaly increment corresponding to the constant-time increment. Consequently, a single-step integration method is chosen.

An analysis of single-step methods in relation to the general equation to be solved resulted in the selection of the fourth-order Runge-Kutta technique. This method has long been accepted as being the most accurate and the fastest method. The Runge-Kutta technique is well suited as a rapid method for use in equation (19), since C_1 and C_2 need to be calculated only once during each time interval. In the solution of equation (19), the integration speed of the fourth-order Runge-Kutta method is very close to the integration speed of a lower ordered method, or even to the integration speed of the multistep methods that are normally much faster.

Step-Size and Error Analysis

Assuming that the error attributable to a given single-step method is less than the maximum tolerable error, optimization of operating speed can be achieved by investigating the question of maximum step-size.

The computer program has been written so that a step-size $\Delta\theta$ can be input to the computer. The program prints out every $n\Delta\theta$ interval, where n is a specified integer; however, calculations may be based on a smaller, more practical interval for accuracy, speed, and stability. The practicality of $\Delta\theta$ is determined at the start of each interval. Practical step-size and its determination by the program are described in appendix C.

The errors induced by numerical integration can be broken down in the following manner.

$$E_n = e_{t,n} + e_{t,a} + e_{ro} \quad (35)$$

where E_n = total error at $t = n$

$e_{t,n}$ = one-step truncation error (the error caused by approximation over step n)

$e_{t,a}$ = accumulated truncation error over steps 1 to $n - 1$

e_{ro} = accumulated effect of round-off error over n steps

Accumulated errors. - Contrary to the usual case in numerical analysis, the sum of the accumulated errors $e_{t,a}$ and e_{ro} is bounded by some number B , which is much less than the accuracy of the input data for any practical step-size and is therefore negligible regardless of total integration time. The boundary is a result of the stability of the function to be integrated; that is, the solution converges to a specific temperature at a given true anomaly and, on all subsequent orbits, will be virtually at that temperature. Since the temperature at point n in one orbit is the same as the temperature at point n in the next orbit, there is no significant accumulation of errors.

To test this error analysis, a spinning-satellite case was programmed for the IBM 1620 computer so that $e_{t,a}$ and e_{ro} could be analyzed separately. To show that e_{ro} was nearly zero, the program was run using eight-digit and then 12-digit accuracy. The difference in the two solutions occurred in the seventh or eighth decimal place in all the orbits run.

The analysis included the calculation of a function proven to be asymptotic to the accumulated truncation error of the integration (ref. 2). Although the solution of this function was small with respect to the solution of the thermal equation and was oscillatory in nature, it did increase gradually in magnitude. Nevertheless, the stability of the function was generally verified, since asymptotic approximations are not exact.

One-step truncation error. - To study $e_{t,n}$, the IBM 1620 computer analysis for total truncation error was used again. Since the one-step error $e_{t,1}$ made during the first step is generated by the accumulated error function, the analysis necessarily contained the rudiments of a one-step error function. The only modification necessary to generate the one-step errors continuously was to make the computer "think" that each step was the first. This technique was applied to the integration of many differential equations with closed-form solutions and was found to have good accuracy. Although the level of effort did not permit extensive application of this technique to equation (19), the runs that were made indicated that any practical step-size gave an $e_{t,n}$ of less than $0.05^\circ R$.

Since the sum of the accumulated errors is bounded by B , equation (35) can be written as

$$E_n = e_{t,n} + B \quad (36)$$

The results of the error analysis indicate that $e_{t,n} < 0.05^\circ R$ and that $B < 0.1^\circ R$; therefore, the total error attributable to the numerical integration is

$$E_n < 0.15^\circ R \quad (37)$$

which is negligible, considering the accuracy of the input data.

Integration Technique When the Function Is Nondifferentiable

A major contribution of the numerical analysis was the discovery and elimination of a significant error caused by misapplication of the integration techniques employed in previous similar studies. In all standard numerical techniques, the function to be integrated is assumed to be continuously differentiable at every point in the integration interval. This assumption does not apply to equation (19) because of the behavior of the constant C_1 . When the orbiting vehicle enters or leaves the shadow of the planet, C_1 may change suddenly from a relatively large number to almost zero, or vice versa, causing the function to be nondifferentiable at the Sun-shade points. In the IBM 1620 computer analysis, the single-step error in an interval containing a Sun-shade point was revealed to be much larger (up to $10^\circ R$ larger) than the total accumulated error encountered immediately prior to that interval. In support of the stability conjecture, even this large error became negligible after a few integration steps.

The error from misapplication is eliminated by subdividing each interval containing a Sun-shade point so that the nondifferentiable point coincides with the boundary of the interval and thus does not exist within the interval.

DIGITAL COMPUTER PROGRAM

The determination of impinging heat loads, absorbed heat loads, and transient temperatures for up to 200 surface elements of a spacecraft in orbit is an ideal application of a high-speed digital computer. Performing this determination in any other manner would be impractical, at best. Accordingly, a Univac 1108 computer program incorporating the theory and numerical techniques described in the previous sections has been written and validated. Programing was done in FORTRAN V to facilitate any modifications that the program user might wish to perform. Appendixes D to J provide the program user with detailed information on the computer program source and data deck structure.

Summary of Computer Program Capabilities and Applications

The computer program is designed to compute, plot, and print out heat loads and temperatures for a vehicle in orbit about the Moon or about any planet except Pluto. The orbiting vehicle may be spinning or have a fixed orientation with respect to either the Sun or the celestial body orbited. The planet surface temperature, which is used in determining the amount of impinging thermal radiation originating from the planet, is assumed to be constant for all celestial bodies except for the Moon, for which surface temperature may be considered either constant or variable.

The heat loads during one orbit or any fraction of an orbit can be determined by the computer program. When it is necessary to compute temperature, more than one orbit may be specified. However, if all of the temperatures become stable (that is, differ by less than 0.5° R from one orbit to the next), the case being computed is automatically terminated.

The optical properties, specific heat, and density of the vehicle coating(s) and substrate(s) are temperature dependent. Up to 21 data points of material property versus temperature from 0° to $10\,000^\circ \text{ R}$ (the effective temperature of the Sun) may be specified. Absorptance with respect to heat flux radiated by the planet is found as a function of the effective temperature of that part of the planet actually radiating to the element of an orbiting vehicle. A maximum of eight tables of optical properties may be used. Eight tables of specific heat and of density corresponding to eight possible substrate materials for vehicle skin may also be used.

When temperature calculations are desired, up to eight schedules of internal heat versus time can be used. Each data set is input as a table of internal heat and switching time, corresponding to a change to the next value of internal heat. As many as 20 different values (10 duty cycles) may be included in each of the eight tables. If the program runs for more than one orbit, the same sequence of internal heat loads is applied for the successive orbits.

Satellite position in orbit is given by the true anomaly θ measured from perigee. A starting value θ_0 and a step-size $\Delta\theta$ are input. The starting value θ_0 must be between 0° and 360° , and $\Delta\theta$ must be a submultiple of 360° and must be $\geq 2^\circ$. For example, 2° , 5° , 6° , 7.5° , 8° , 9° , or 10° are all acceptable values for $\Delta\theta$, but 7° is not acceptable, since $360/7$ is not an integer.

The step-size $\Delta\theta$ may be subdivided internally to give the optimum interval with respect to accuracy and computer time; however, the input value of $\Delta\theta$ will not be exceeded. For temperature calculations, external heat fluxes are considered constant throughout the specified $\Delta\theta$ interval, provided the vehicle does not pass in or out of the shadow of the planet during the time corresponding to $\Delta\theta$. Furthermore, only one change in internal heat for each element is recognized per interval. Experience indicates that $\Delta\theta = 10^\circ$ is a reasonable upper bound.

The position of the Sun relative to the orbit may be given conventionally as right ascension and declination, both of which can be obtained from an ephemeris; or the position of the Sun may be specified in terms of the angles α , γ , and β relative to the X_p , Y_p , and Z_p coordinate axes.

As many as 200 elements of oriented vehicles may be processed in a single case. Each element has a specified thickness, initial temperature, and position on the vehicle. An element may use one of eight choices for each of the following: optical properties, substrate material, and internal heat duty cycle.

The effects of variations of thickness, initial temperature, material, or internal heat loads can be studied in a single case if a number of elements are chosen with the same positions, but with different values for the other parameters. Thus, a single case may be an entire parametric study in itself. Additional flexibility is provided so that the unchanged values can be represented by blank fields in the input deck if any of the input parameters are to remain the same from one case to another.

General Deck Preparation

Figure 12 is an illustration of the physical sequence of the deck structure required. The name shown on the card preceding each routine (i.e., DECK1) can be related to the actual name of the routine by the following list:

EXEC = PILØT	DECK13 = SUNØR	DECK26 = GEØFAC
DECK1 = HEAD	DECK14 = BETA90	DECK27 = QIIN
DECK2 = TINPUT	DECK15 = WYE	DECK28 = DDVETA
DECK3 = LØØP	DECK16 = SIGBET	DECK29 = DDFERI
DECK4 = TØUT	DECK17 = INTERP	DECK30 = TABLE
DECK5 = HEAT	DECK18 = ARRØUT	DECK31 = MAIN2
DECK6 = FREAD	DECK19 = ARCØS	DECK32 = DRAW
DECK7 = QIFIND	DECK20 = QUART	DECK33 = SKALE
DECK8 = TALLY	DECK21 = PHIFN	DECK34 = XINTRP
DECK9 = LØCUS	DECK22 = GFN	DECK35 = ACCEND
DECK10 = TEMPER	DECK23 = FØFX Y	DECK36 = HILØW
DECK11 = INIT	DECK24 = DELTA	DECK37 = TIDENT
DECK12 = FIND	DECK25 = ERRØR	DECK38 = FDTA

If Stromberg-Carlson 4020 (SC-4020) plots are not requested, LINK2 may be omitted entirely. This can be done by deleting card number DK033000 of subroutine LØØP (DECK3) and by removing all subroutines which make up LINK2 (DECK31 through DECK38). This procedure must be used if data-computation facilities do not have SC-4020 capabilities described in appendix H.

Data Deck Preparation

The data deck consists of three groups of data. They are the permanent-data cards, the material-properties cards, and the case-data cards. The permanent data consist of 147 permanent-data cards that include 144 cards containing the radiation configuration factors and three cards containing alphabetic data used in headings. For documentation purposes, the permanent-data cards are listed as part of the sample-case input data discussed in appendix G. Since the permanent data must be used exactly as listed, only the material-properties data and case data shall be of concern for data preparation.

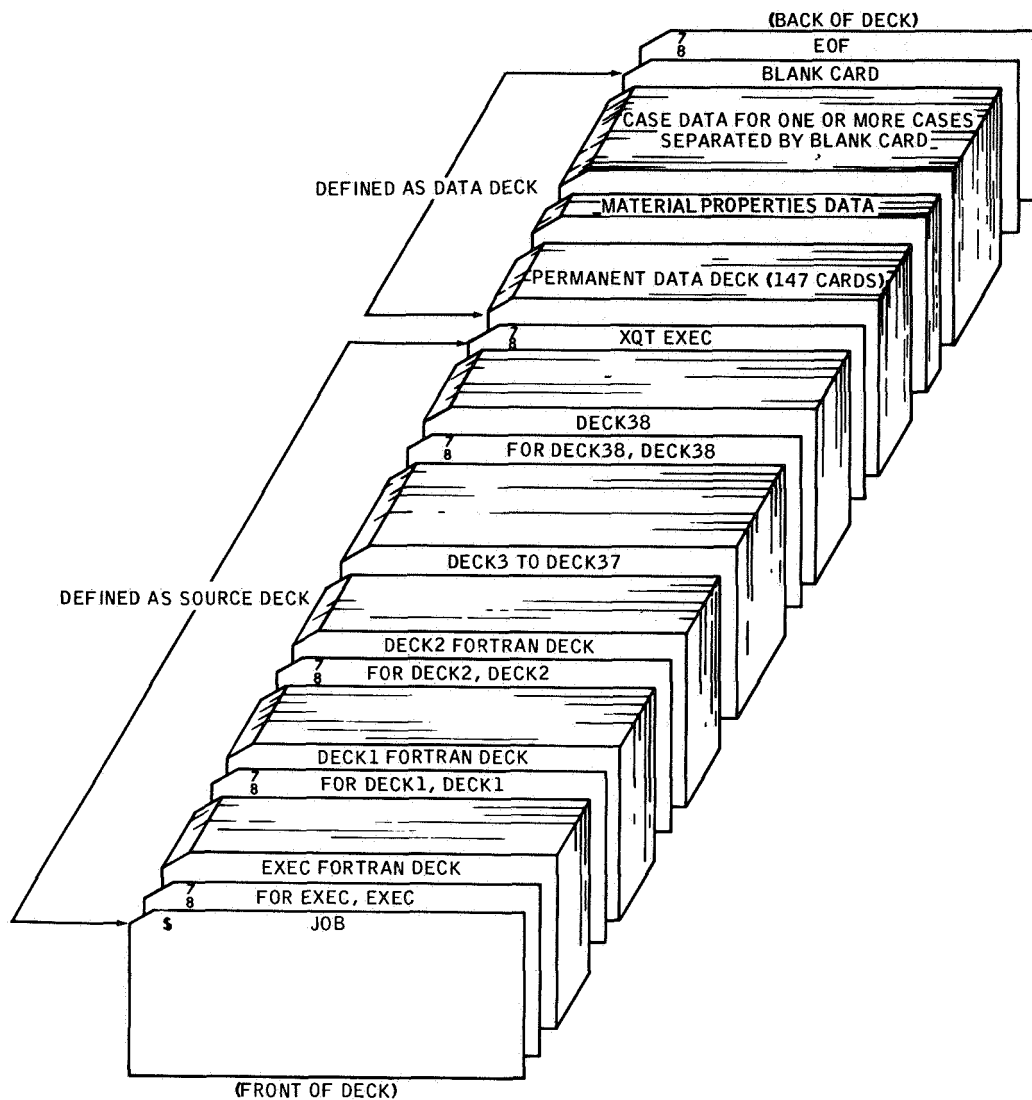


Figure 12. - Deck structure.

Appendix I is a guide for preparing program data. The guide is complete enough that the program user can use the appendix to prepare data decks. Additional information and the related theory are described in other sections of the report and in the other appendixes.

COMPUTER PROGRAM APPLICATION

A hypothetical lunar mission was run, using the described computer program. For the hypothetical lunar-orbital mission, the spacecraft was planet oriented, and a variable planet surface temperature was used. The orbit parameters were such that

pericyynthion occurred over the intended landing site, which was 13° south of the Moon equator and 3° east (as seen from Earth) of the prime meridian of the Moon. A detailed discussion of the use of the computer program in spacecraft design and orbit selection is presented in reference 3.

Pertinent orbit data consisted of the following:

Altitude H , n. mi.	10 to 190
Inclination i , deg	13
Right ascension of ascending node Ω , deg	87
Argument of perifocus ω , deg	270
True anomaly at initial time ϕ_0 , deg	0

The mission was flown on December 1, 1963. The position of the Sun in selenographic coordinates, as given by an ephemeris, was as follows:

Colongitude, deg	88.74
Declination DEC, deg	0.86
Right ascension (90° - colongitude) RA, deg	1.26

At a declination of 0.86° and a right ascension of 1.26° , the Sun is almost in the orbital plane.

The temperature-time history of differently oriented and differently painted surface elements reveals several interesting characteristics. In figure 13, it is shown that a white element facing away from the Moon cools initially, even though it may be facing almost directly into the Sun. However, a similarly oriented black element rises to a peak temperature at $\phi = 60^\circ$, then gradually drops in temperature as it turns away from the Sun. This difference is to be expected, since the white element reflects a considerable amount of solar energy; whereas, the black element will absorb most of the solar radiation that impinges upon its surface.

In figure 14, it is shown that the temperature curves of a black element and a white element facing the Moon are similar, since black and white elements absorb long-wavelength radiation in almost the same amount. Also, the hump in the black-element curve at about $\phi = 100^\circ$ and $\phi = 260^\circ$ occurs because both elements are briefly irradiated by the Sun immediately before entering the shadow of the Moon and immediately after leaving the shadow. However, the hump appears in the black-element curve only, since the black element absorbs the solar radiation more readily.

In figure 15, comparison of the black element facing the Moon with the black element facing away from the Moon and toward the Sun reveals that at $\phi = 0^\circ$, the element facing the Moon is almost as hot as the element facing away from the Moon and toward the Sun. The conclusion is that, at low lunar orbits, planet heat can be as significant as solar heat.

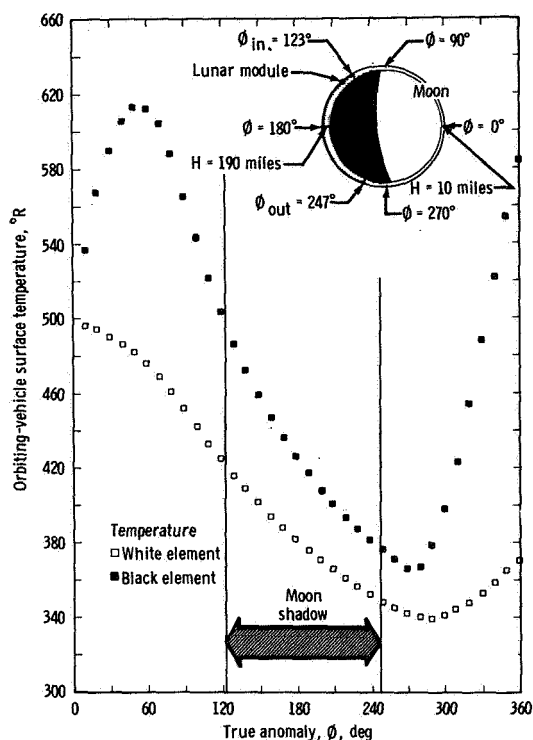


Figure 13. - Simulated lunar module, elements facing away from the Moon.

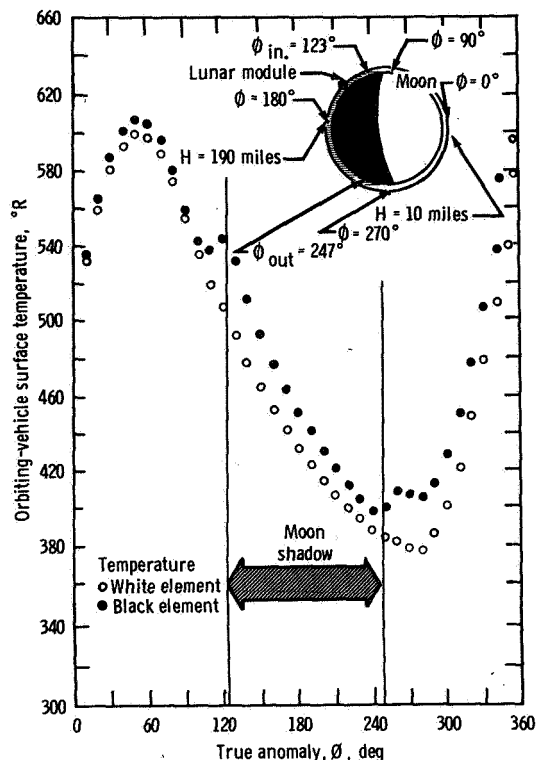


Figure 14. - Simulated lunar module, elements facing the Moon.

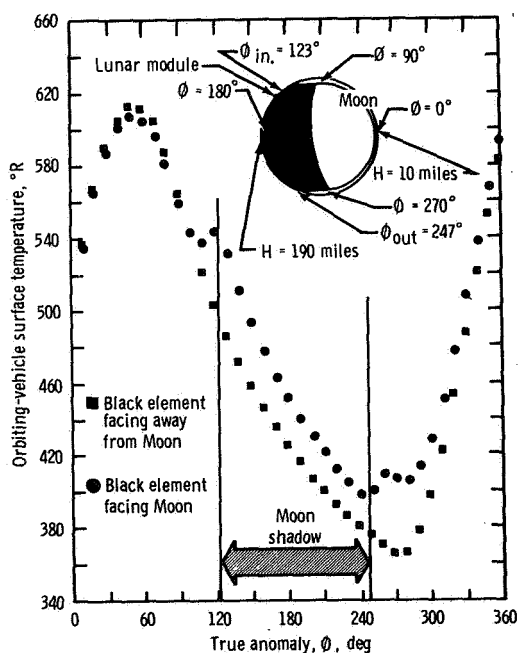


Figure 15. - Simulated lunar module.

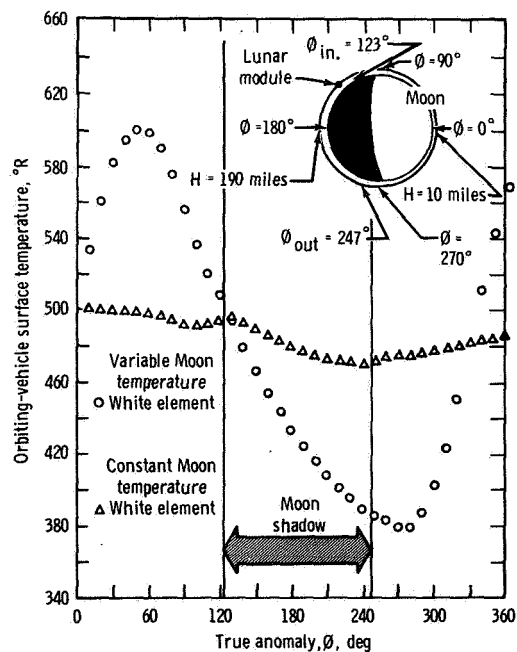


Figure 16. - Comparison of constant and variable Moon temperature.

To investigate the effect of the surface temperature gradients of the Moon not being considered, the same case was run, assuming a constant Moon temperature. The results plotted in figure 16 reveal that the constant-Moon-temperature curve is a flat curve that averages out the maximum and minimum peaks of the variable-Moon-temperature curve. It is clearly shown in figure 16 that a significant variation (about 100° R) is caused by neglecting the surface temperature gradient of the Moon. These results confirm the importance of the variable-planet-temperature method in analyzing lunar missions.

CONCLUDING REMARKS

An operational computer program to predict spacecraft heat loads and temperatures has been developed. The program is applicable to a wide variety of vehicle-mission combinations. Although no unduly restricted assumptions are incorporated into the celestial, thermal, and numerical theory, certain conditions were not considered in order to retain the desired versatility. However, the program was written so that it could be expanded readily to include additional features and details.

Manned Spacecraft Center
National Aeronautics and Space Administration
Houston, Texas, August 22, 1968
039-00-00-85-72

APPENDIX A

INTERSECTION OF ORBIT AND SHADOW

The intersection of the orbit and the shadow can be determined by simultaneously solving the equations that express the vehicle path and the trace of the shadow on the orbital plane.

If $\beta = 0^\circ$ or $\beta = 180^\circ$, the trace of the shadow is a circle of radius r_p that does not intersect the vehicle orbit. (The angle β is illustrated in appendix B.) If $0^\circ < \beta < 180^\circ$, but $\beta \neq 90^\circ$, the trace is a semiellipse, as shown in figure A-1. The equation of the orbit path, based on the notation shown, is

$$\frac{(X_p + c)^2}{a^2} + \frac{Y_p^2}{b^2} = 1 \quad (A1)$$

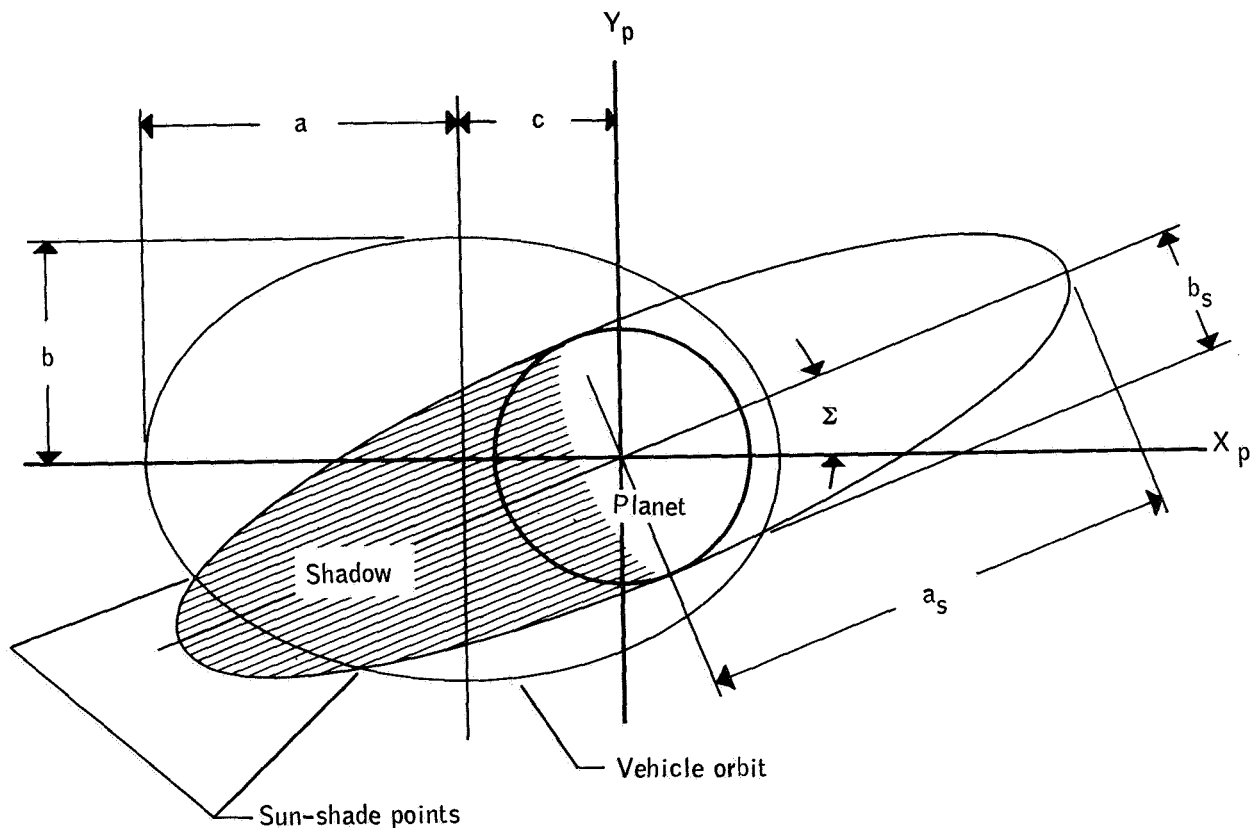


Figure A-1. - Intersection of the orbit plane and the shadow of the planet.

The corresponding expression for the intersection of the shadow with the orbital plane is

$$\frac{(X_p \cos \Sigma + Y_p \sin \Sigma)^2}{a_s^2} + \frac{(X_p \sin \Sigma - Y_p \cos \Sigma)^2}{b_s^2} = 1 \quad (\text{A2})$$

Equations (A1) and (A2) can be combined to yield a biquadratic (quartic), which is solved in this analysis by the Ferrari method. Other numerical approaches may be faster, but they become unstable in some cases. The Ferrari method, which can be used to solve any quartic equation, is explained in most theory-of-equation texts and in many mathematical handbooks (for example, ref. 4).

After the quartic equation has been solved for X_p , the corresponding Y_p values can be determined from either equation (A1) or (A2); however, care must be taken to avoid accepting an extraneous root. Extraneous roots do not occur if both equations are solved for Y_p^2 and the expressions equated. The result reduces to the following equation, which is single valued for all X_p .

$$\begin{aligned} \frac{Y_p}{a} = & \chi \left[(\cos \beta \cos \Sigma)^2 + \sin^2 \Sigma \right] - \left(\frac{b_s}{a} \right)^2 \\ & + \left(\frac{b}{a} \right)^2 \left[\left(1 - \chi + \frac{c}{a} \right)^2 \right] \left[(\cos \beta \sin \Sigma)^2 + \cos^2 \Sigma \right] \end{aligned} \quad (\text{A3})$$

where

$$\chi = \frac{X_p}{a} \quad (\text{A4})$$

The true anomaly of the Sun-shade points is calculated from

$$\tan \theta = \frac{Y_p}{X_p} \quad (\text{A5})$$

If the shadow and orbit ellipses do not intersect or if they cross in only two places, the quartic equation can yield spurious (that is, false) real roots. Since equation (A3) is valid only at the ellipse intersections, solutions of equation (A3) that correspond to spurious roots are meaningless. Therefore, the validity of each pair of coordinates must be determined, and this is accomplished by testing whether the X_p and Y_p values satisfy both equations (A1) and (A2).

If $\beta = 90^\circ$, the trace of the shadow ellipse degenerates to a pair of parallel lines that are expressed in polar form as

$$r_s \sin(\Sigma - \emptyset) = r_p \quad (\text{A6a})$$

and

$$r_s \sin(\Sigma - \emptyset) = -r_p \quad (\text{A6b})$$

where r_s is the distance from the lines to the center of mass of the planet. A similar expression of the orbit ellipse is

$$r_o = \frac{b^2}{a + c \cos \emptyset} \quad (\text{A7})$$

where r_o is the distance from the orbit ellipse to the center of mass of the planet.

Equating r_s and r_o in equations (A6a) and (A7) gives

$$\left(b^2 \sin \Sigma + cr_p\right) \cos \emptyset - b^2 \cos \Sigma \sin \emptyset - ar_p = 0 \quad (\text{A8a})$$

In general, equation (A8a) is transformed to a quadratic in $\cos \emptyset$ by substituting $\pm (1 - \cos^2 \emptyset)^{1/2}$ for $\sin \emptyset$. The roots of this quadratic are substituted into equation (A8a) to give $\sin \emptyset$. Once $\sin \emptyset$ and $\cos \emptyset$ are known, \emptyset is easily found. A similar procedure yields the other Sun-shade point if equations (A6b) and (A7) are equated.

$$\left(b^2 \sin \Sigma + cr_p\right) \cos \emptyset - b^2 \cos \Sigma \sin \emptyset + ar_p = 0 \quad (\text{A8b})$$

If the coefficient of $\sin \varnothing$ or $\cos \varnothing$ in equation (A8b) vanishes, the problem is less difficult, in that $\sin \varnothing$ or $\cos \varnothing$ can be solved for directly, and the corresponding function is determined from either $\sin \varnothing = \pm (1 - \cos^2 \varnothing)^{1/2}$ or $\cos \varnothing = \pm (1 - \sin^2 \varnothing)^{1/2}$

Equations (A5) and (A8b) may give zero, two, or four valid mathematical roots; however, two of the roots may not satisfy the condition that the Sun-shade points lie on the shaded side of the orbit. A root is discarded unless the angle between the line connecting the Sun-shade point and the planet and the line of the projection of the Sun on the orbital plane is obtuse.

APPENDIX B

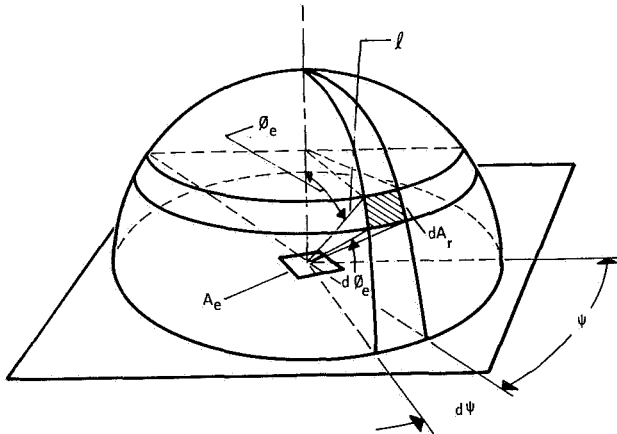
RADIATION HEAT-TRANSFER DERIVATIONS

A general expression for thermal radiation from one surface to another can be derived by considering the total radiation from the emitting surface. This derivation may be visualized by placing a hemisphere over the emitting-surface element A_e , as shown in figure B-1. This hemisphere will intercept all of the heat emitted from A_e , and the radiation received by an element dA_r is given by the equation

$$dq = I \frac{\epsilon_e \cos \vartheta_e A_e \cos \vartheta_r}{\ell^2} dA_r \quad (B1)$$

where ϑ_r (not shown) is the angle between a normal to dA_r and a line from dA_r to A_e , and ϵ_e is the emittance of A_e . The proportionality factor I is shown as the intensity of radiation. Assuming that the emitting surface is diffuse, the intensity is independent of direction and is constant. The apparent or intercepted intensity is proportional to A_e , as seen from dA_r on the hemisphere. Thus

$$I(\vartheta_e) = I \cos \vartheta_e \quad (B2)$$



which is Lambert's cosine law. If dA_r is on the hemisphere surface, ϑ_r is 0° ; therefore

$$dq = \frac{I \cos \vartheta_e A_e \epsilon_e dA_r}{\ell^2} \quad (B3)$$

Figure B-1. - Mathematical model for deriving the general thermal-radiation equation.

Writing dA_r in terms of θ_e and ψ and integrating, equation (B3) becomes

$$q = IA_e \epsilon_e \int_0^{2\pi} d\psi \int_0^{\pi/2} \cos \theta_e \sin \theta_e d\theta_e = IA_e \epsilon_e \pi \quad (B4)$$

The heat radiated by A_e is

$$q_{out} = \epsilon_e (\sigma T_e^4) A_e = \epsilon_e E_e A_e \quad (B5)$$

where E_e is the emissive power of A_e . Since all of the emitted heat is intercepted by the hemisphere, the heat emitted is equivalent to the heat received. Equating equations (B4) and (B5), simplifying the results, and solving yields

$$I = \frac{E_e}{\pi} \quad (B6)$$

Combining equations (B1) and (B6) and introducing the absorptance α_r of the receiving surface gives the general equation for radiant heat transfer between two surfaces

$$dq = \frac{E_e \cos \theta_e A_e \cos \theta_r \alpha_r \epsilon_e}{\pi \ell^2} dA_r \quad (B7)$$

In the following section of this appendix, equation (B7) will be used to derive specific equations that express the amount of planetary and albedo heat that is received, absorbed, or both, by either a spinning or an oriented vehicle. Derivations will be made for constant-temperature planets, as well as for variable-temperature planets.

Planet Heat Flux for a Spinning Vehicle and Constant Planet Temperature

Seavey (ref. 5) simplified the mathematics of the derivation with the theorem that the flux incident on a small surface element from an extended source of uniform brightness is independent of the shape of the element and the shape of the emitting surface. A modification of Seavey's solution is developed by first replacing the part of the planet that the vehicle receives radiation from with a concave disk of radius ℓ ,

as shown in the top portion of figure B-2. This geometrical substitution can be visualized by observing that the Sun, a uniform emitting surface with respect to surface position, appears to the observer on Earth as a disk. Similarly, the vehicle is replaced by a flat disk that has an infinitesimal area, compared to the area of the planet disk.

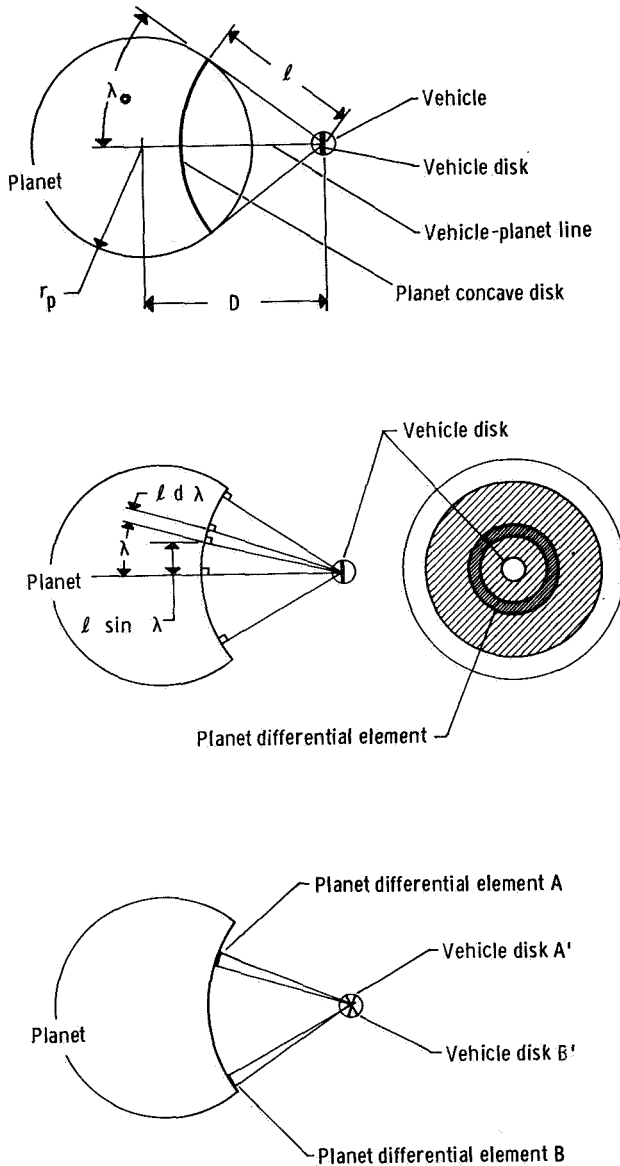


Figure B-2. - Seavey's model for determining radiation between two spheres of uniform brightness.

It can be seen in the center diagram of figure B-2 that every line between the vehicle and the planet intersects the planet concave disk perpendicularly. Likewise, the vehicle-planet lines can be made normal to the planet concave disk by letting the disk pivot about its center, allowing the planet differential elements to receive radiation from different orientations of the vehicle disk. Thus, in the bottom portion of figure B-2, element A receives radiation from element A', and element B receives radiation from element B'. The pivoting of the planet concave disk is justified by the previously mentioned theorem employed by Seavey.

Equation (B7) is modified to express the heat received by a vehicle area A_v from a ring-shaped element of area dA_p as

$$dq_p = \frac{E_p \cos \phi_p \cos \phi_v A_v \epsilon_p \alpha_p}{\pi l^2} dA_p \quad (B8)$$

Assuming that celestial bodies are perfect emitting surfaces (that is, that $\epsilon_p = 1$) and substituting $\phi_p = \phi_v = 0^\circ$, equation (B8) is reduced to

$$dq_p = \frac{E_p A_v \alpha_p}{\pi l^2} dA_p \quad (B9)$$

The planet differential area and the effective vehicle area are

$$dA_p = [2\pi(\ell \sin \lambda)](\ell d\lambda) = 2\pi\ell^2 \sin \lambda d\lambda \quad (B10)$$

and

$$A_v = \pi r_v^2 \quad (B11)$$

The emissive power is

$$E_p = \sigma T_p^4 \quad (B12)$$

where T_p is the average planet temperature. A heat balance on the planet yields

$$T_p = \left[\frac{S(1 - R)}{4\sigma} \right]^{1/4} \quad (B13)$$

Combining equations (B12) and (B13) gives

$$E_p = \sigma \left\{ \left[\frac{S(1 - R)}{4\sigma} \right]^{1/4} \right\}^4 \quad (B14)$$

which becomes

$$E_p = \frac{S(1 - R)}{4} \quad (B15)$$

Substituting equations (B10), (B11), and (B15) into equation (B9) and integrating yields the desired relationships

$$q_p = 2S(1 - R)\alpha_p F_1 \pi r_v^2 \quad T_p = \text{constant} \quad (B16a)$$

and

$$\frac{q_p}{4\pi r_v^2} = \frac{S(1-R)\alpha_p F_1}{2} \quad T_p = \text{constant} \quad (\text{B16b})$$

where

$$F_1 = \frac{1 - \left[1 - \left(\frac{r_p}{D} \right)^2 \right]^{1/2}}{4} \quad (\text{B16c})$$

The planet heat rate per unit area impinging on the spherical vehicle is

$$\frac{q_p}{A_v} = 2S(1-R)F_1 \quad T_p = \text{constant} \quad (\text{B16d})$$

A derivation not based on Seavey's theorem can also be made and will be presented in condensed form. The heat emitted from planet element dA_p (fig. B-3) and absorbed by a satellite of radius r_v is

$$dq_p = \frac{E_p \cos \theta_p \cos \theta_v dA_p \alpha_p \epsilon_p \pi r_v^2}{\pi l^2} \quad (\text{B17})$$

Again substituting $\epsilon_p = 1$, $\theta_v = 0$, and $E_p = S(1-R)/4$ and integrating, equation (B17) becomes

$$q_p = \frac{S(1-R)r_v^2 \alpha_p}{4} \iint \frac{\cos \theta_p}{l^2} dA_p \quad (\text{B18})$$

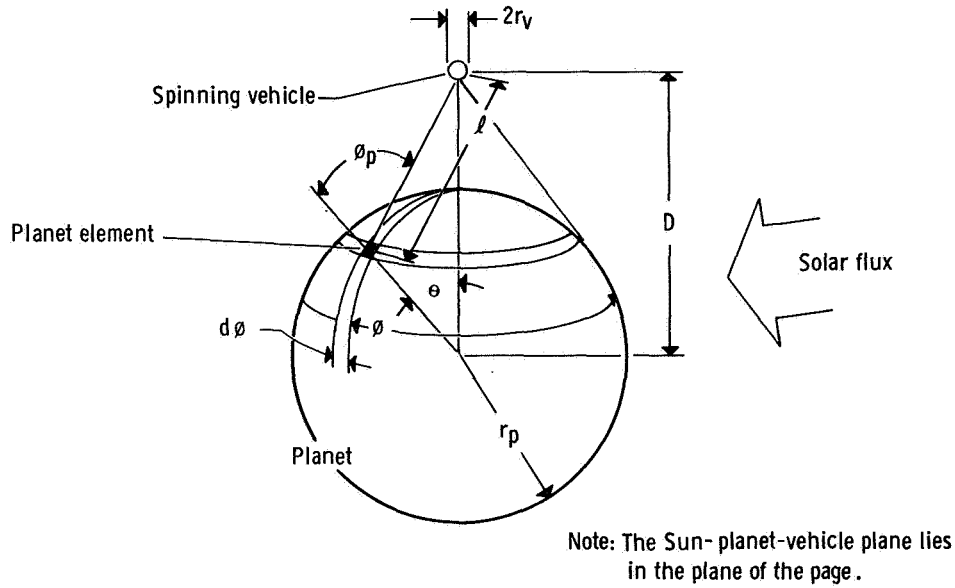


Figure B-3. - Geometry for determining heat exchange between planet and spinning vehicle.

It can be shown that

$$\cos \phi_p = \frac{D \cos \theta - r_p}{l} \quad (\text{B19})$$

$$l^2 = r_p^2 + D^2 - 2r_p D \cos \theta \quad (\text{B20})$$

and

$$dA_p = r_p^2 \sin \theta \, d\theta \, d\phi \quad (\text{B21})$$

where the variables are illustrated in figure B-3. Equations (B19) to (B21) were used in reference to evaluate an expression similar to equation (B18), and from this work

the following equality can be developed

$$\iint \frac{\cos \theta_p}{\ell^2} dA_p = 2\pi \left\{ 1 - \left[1 - \left(\frac{r_p}{D} \right)^2 \right]^{1/2} \right\} \quad (B22)$$

Combining equation (B22) with equation (B18) gives

$$q_p = \frac{S(1 - R)\alpha_p}{2} \left\{ 1 - \left[1 - \left(\frac{r_p}{D} \right)^2 \right]^{1/2} \right\} \pi r_v^2 \quad T_p = \text{constant} \quad (B23)$$

which is equivalent to equation (B16a).

Planet Heat Flux for a Spinning Vehicle and Variable Moon Temperature

Equation (B17) for radiant heat exchange between a surface differential element and a spherical satellite is applicable also to a variable Moon temperature. If the Moon has a negligible atmosphere and is a good thermal insulator, the element dA_p is in thermal equilibrium. That is, if mechanisms for heat transfer do exist (for example, conduction and convection in a planet atmosphere) so that dA_p is not in equilibrium, the surface temperatures tend to approach an average value. Also, the presence of an atmosphere, which may change considerably from one orbit to the next, makes it impractical to prepare meaningful albedo input data. Thus, average albedo values are used for the planets, and the method for constant (that is, average) planet temperature is employed. Based on the correlation of independently obtained experimental data, the Moon, as "seen" by a spacecraft in lunar orbit, for all practical purposes, can be considered a smooth sphere when determining the lunar planetary heat received by a lunar-orbital vehicle. Therefore, the heat emitted is equivalent to the solar heat absorbed, as shown in figure B-4, and

$$E_p = q_{in} = q_{out} = S(1 - R)\cos \beta \quad (B24)$$

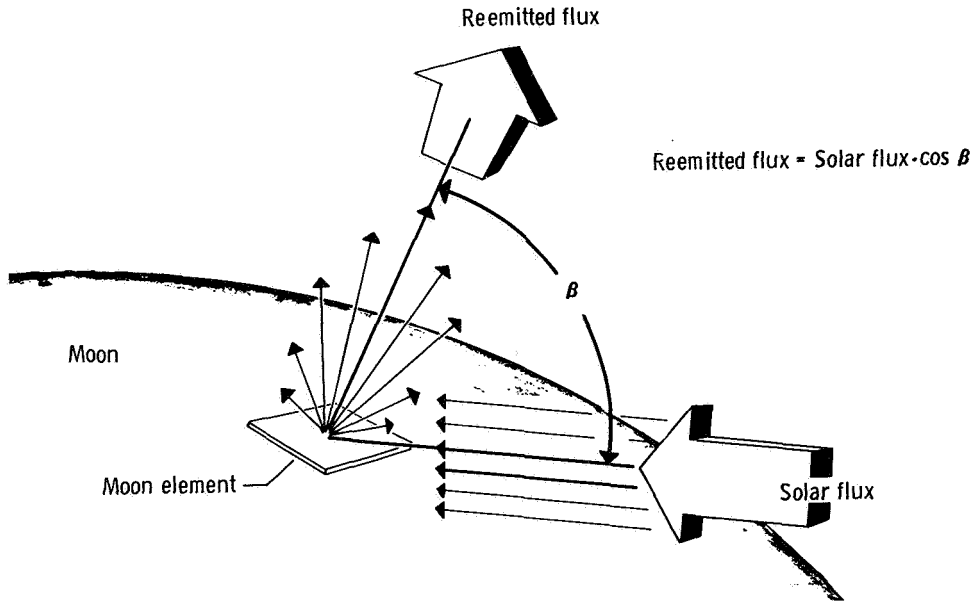


Figure B-4. - Heat emitted by an element of a variable-temperature Moon.

Combining equations (B17) and (B24), letting $\epsilon_p = 1$ and $\theta_v = 0$, integrating, and rearranging yields

$$q_p = 4 \left[\frac{S(1-R)r_v^2 \alpha_p}{4} \right] \iint \cos \beta \frac{\cos \theta_p}{\ell^2} dA_p \quad (B25)$$

The function $\cos \beta$ is shown to be

$$\cos \beta = \cos \theta \cos \theta_s + \sin \theta \sin \theta_s \cos \theta \quad (B26)$$

which can be approximated by

$$\cos \beta \approx \cos \theta_s \quad (B27)$$

Equation (B27) can be mathematically substantiated for small values of θ since

$$\lim_{\theta \rightarrow 0} \cos \beta = \cos \theta_s \quad (\text{B28})$$

For larger values of θ , the altitude is large and F is small; consequently, the error introduced by the approximation is relatively insignificant. Combining equations (B25) and (B27) gives

$$q_p \approx 4 \cos \theta_s \left[\frac{S(1-R)r_v^2 \alpha_p}{4} \iint \frac{\cos \theta_p}{\ell^2} dA_p \right] \quad (\text{B29})$$

Combining equations (B22) and (B29) gives

$$q_p = 8S(1-R)\alpha_p F_2 \pi r_v^2 \quad T_p \neq \text{constant} \quad (\text{B30a})$$

and

$$\frac{q_p}{4\pi r_v^2} = 2S(1-R)\alpha_p F_2 \quad T_p \neq \text{constant} \quad (\text{B30b})$$

where

$$F_2 \approx \left\{ \frac{1 - \left[1 - \left(\frac{r_p}{D} \right)^2 \right]^{1/2}}{4} \right\} \cos \theta_s \quad (\text{B30c})$$

The variable-temperature planet heat rate per unit area impinging on the spherical vehicle is

$$\frac{q_p}{A_v} = 8S(1 - R)F_2 \quad T_p \neq \text{constant} \quad (\text{B30d})$$

An expression similar to equation (B25) has been numerically evaluated by Ballinger (ref. 6) on a digital computer. The resulting F-factors were tabulated as a function of altitude H and Sun-planet-vehicle angle θ_s . (The table of F-factors was duplicated at MRI as part of a study to determine the effect of surface topology on planet thermal emission.) The approximate radiation configuration factor F_2 is related to the tabulated factor $F(H, \theta_s)$ by

$$F_2 \approx \frac{F(H, \theta_s)}{8} \quad (\text{B31})$$

Random values of $F(H, \theta_s)/8$ and the approximate function F_2 are plotted in figure B-5. As expected, the percent deviation is small for large values of F , and the

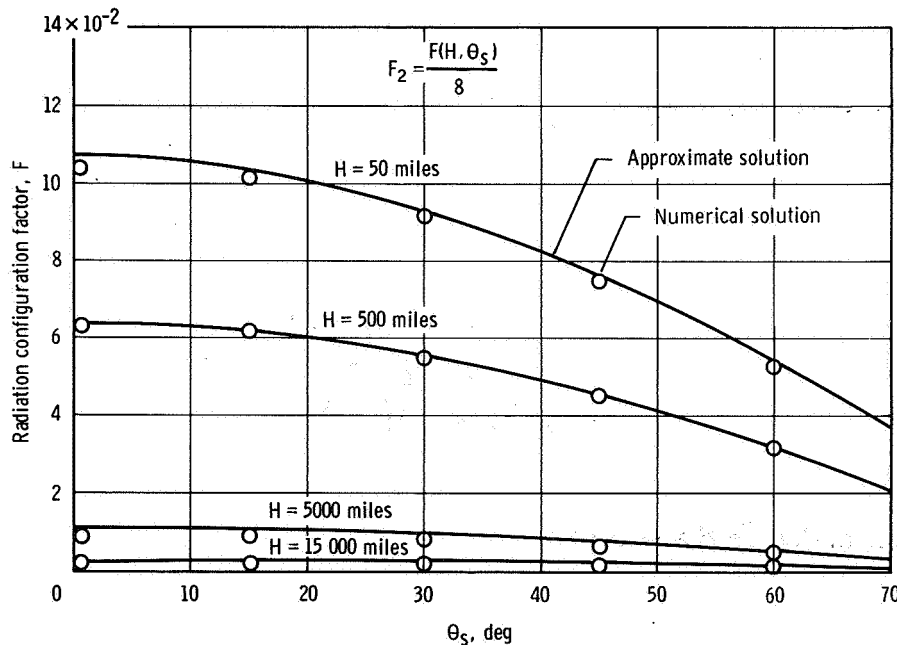


Figure B-5. - Radiation configuration factors, approximate and numerical solutions compared.

magnitude of deviation is negligible for low values of F ; therefore, the approximation $\cos \beta \approx \cos \theta_s$ introduces insignificant errors into the derivation of F_2 . Consequently, the F_2 function has been incorporated into the computer program in lieu of the slower table-reference method to obtain $F(H, \theta_s)/8$.

The dark side of a variable-temperature Moon is essentially in thermal equilibrium; therefore, the heat emitted can be derived from the relations that are applicable to a constant-temperature planet.

The emissive power is

$$E_p = \sigma T_m^4 \quad (B32)$$

where T_m is the dark-side or minimum temperature of the Moon. Combining equations (B32), (B9), (B10), and (B11) and integrating gives

$$q_p = 8\sigma T_m^4 \alpha_p F_1 \pi r_v^2 \quad T_p \neq \text{constant (dark side of planet)} \quad (B33a)$$

and

$$\frac{q_p}{4\pi r_v^2} = 2\sigma T_m^4 \alpha_p F_1 \quad T_p \neq \text{constant (dark side of planet)} \quad (B33b)$$

where

$$F_1 = \frac{1 - \left[1 - \left(\frac{r_p}{D} \right)^2 \right]^{1/2}}{4} \quad (B33c)$$

The heat impinging on the sphere is

$$\frac{q_p}{A_v} = 8\sigma T_m^4 F_1 \quad T_p \neq \text{constant (dark side of planet)} \quad (B33d)$$

Albedo Heat Flux for a Spinning Vehicle

Solar radiation reflects diffusely from celestial bodies; therefore, the equations derived to express planetary diffuse thermal emission can be modified to be applicable to albedo calculations. Albedo includes solar energy bounced or scattered from the atmosphere above dA_p , as well as solar energy reflected from the planet surface element. The heat reflected from differential element dA_p (fig. B-3) and absorbed by A_v is given by the familiar relationship

$$dq_{\text{albedo}} = \frac{E_p \cos \theta_p dA_p \alpha_s \epsilon_s \pi r_v^2}{\pi \ell^2} \quad (\text{B34})$$

where E_p is the effective emissive power of the element

$$E_p = SR \cos \beta \quad (\text{B35})$$

Combining equations (B34) and (B35), letting $\epsilon_s = 1$, simplifying, and integrating gives

$$q_{\text{albedo}} = SR r_v^2 \alpha_s \iint (\cos \beta) \left(\frac{\cos \theta_p}{\ell^2} dA_p \right) \quad (\text{B36})$$

Introducing the approximation $\cos \beta \approx \cos \theta_s$, equation (B36) becomes

$$q_{\text{albedo}} = SR r_v^2 \alpha_s \cos \theta_s \iint \frac{\cos \theta_p}{\ell^2} dA_p \quad (\text{B37})$$

Substituting equation (B22) into equation (B37) gives

$$q_{\text{albedo}} = 8SR \alpha_s F_2 \pi r_v^2 \quad (\text{B38a})$$

and

$$\frac{q}{4\pi r_v^2} = 2SR\alpha_s F_2 \quad (B38b)$$

where

$$F_2 \approx \left\{ \frac{1 - \left[1 - \left(\frac{r_p}{D} \right)^2 \right]^{1/2}}{4} \right\} \cos \theta_s \quad (B38c)$$

The albedo heat rate per unit area impinging on the spherical vehicle is

$$\frac{q_{\text{albedo}}}{A_v} = 8SRF_2 \quad (B38d)$$

Planet Heat Flux for an Oriented Vehicle and Variable Moon Temperature

Equations that express the amount of thermal radiation received by a flat element in space can be derived from equation (B7). The derivations for a flat element are similar to the derivations for a sphere; however, the plate is not necessarily symmetrical with respect to the vehicle-planet line; consequently, the orientation of the plate is difficult to define. The heat absorbed by the flat-plate element (fig. B-6) from the variable surface temperature of the Moon is

$$dq_p = \frac{E_p \cos \phi_p \cos \phi_v dA_p \alpha_p \epsilon_p A_v}{\pi \ell^2} \quad (B39)$$

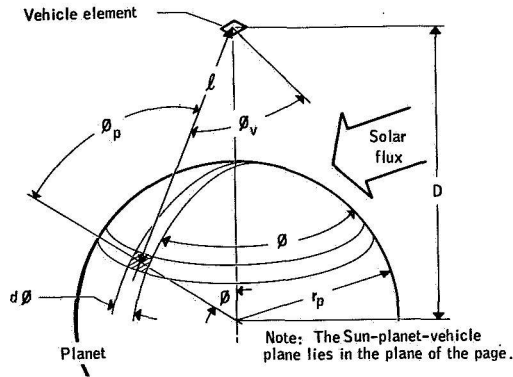


Figure B-6. - Geometry for determining heat between planet and vehicle element.

Substituting $\epsilon_p = 1$ and $E_p = S(1 - R)\cos \beta$ (eq. (B24)) into equation (B39) and integrating yields

$$q_p = S(1 - R)\alpha_p A_v \iint \frac{\cos \beta \cos \theta_p \cos \theta_v}{\pi l^2} dA_p \quad (\text{B40})$$

An equation similar to equation (B40) was numerically integrated in reference 6, and the following equality was developed.

$$\iint \frac{\cos \beta \cos \theta_p \cos \theta_v}{\pi l^2} dA_p = F_3(H, \theta_s, \theta_c, \epsilon) \quad (\text{B41})$$

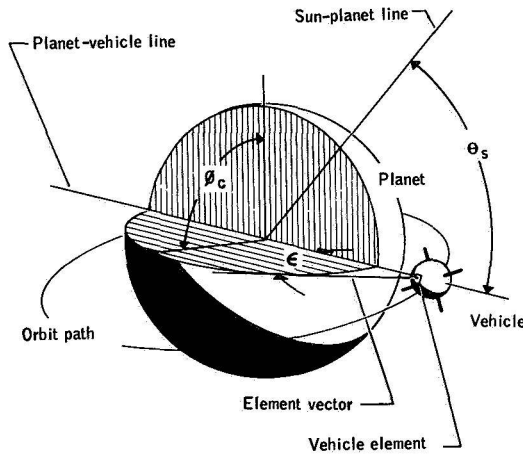


Figure B-7. - Angular variables for determining radiation configuration factors of oriented vehicles.

where H is vehicle altitude, and the parameters θ_s , θ_c , and ϵ are illustrated in figure B-7. Values of F_3 as a function of the four independent variables have been calculated and tabulated. The validity of the radiation-configuration-factor table was investigated and established by hand calculations for numerous special cases and for two general cases.

Combining equations (B40) and (B41) gives

$$q_p = S(1 - R)\alpha_p F_3 A_v \quad T_p \neq \text{constant} \quad (\text{B42a})$$

and

$$\frac{q_p}{A_v} = S(1 - R)\alpha_p F_3 \quad T_p \neq \text{constant} \quad (\text{B42b})$$

where values of F_3 are found by reference to the radiation-configuration-factor table. The planet heat rate per unit area impinging on the flat plate is

$$\frac{q_p}{A_v} = S(1 - R)F_3 \quad T_p \neq \text{constant} \quad (\text{B42c})$$

Planet Heat Flux for an Oriented Vehicle and Constant Planet Temperature

The emissive power of a planet of uniform temperature is independent of β ; $E_p = S(1 - R)/4$, as in equation (B15). Substituting the expression for E_p into equation (B39) and integrating gives

$$q_p = \frac{S(1 - R)\alpha_p A_v}{4} \iint \frac{\cos \theta_p \cos \theta_v}{\pi \ell^2} dA_p \quad (\text{B43})$$

Comparing equations (B41) and (B43), it is seen that

$$\iint \frac{\cos \theta_p \cos \theta_v}{\pi \ell^2} dA_p = \iint \frac{\cos(\beta = 0) \cos \theta_p \cos \theta_v}{\pi \ell^2} dA_p \quad (\text{B44})$$

Introducing the approximation $\cos \beta \approx \cos \theta_s$, which was previously shown to have a negligible effect on accuracy, equation (B44) becomes

$$\iint \frac{\cos \theta_p \cos \theta_v}{\pi \ell^2} dA_p \approx F_3(H, \theta_s = 0, \theta_c, \epsilon) \quad (\text{B45})$$

Combining equations (B45) and (B43) gives

$$q_p = \frac{S(1 - R)\alpha_p F_4 A_v}{4} \quad T_p = \text{constant} \quad (\text{B46a})$$

and

$$\frac{q_p}{A_v} = \frac{S(1-R)\alpha_p F_4}{4} \quad T_p = \text{constant} \quad (\text{B46b})$$

where values of F_4 are found by reference to the radiation-configuration-factor table and are the same as F_3 , except for one of the independent variables θ_s that is treated as a constant zero. The impinging heat rate is

$$\frac{q_p}{A_v} = \frac{S(1-R)F_4}{4} \quad T_p = \text{constant} \quad (\text{B46c})$$

The dark side of the variable-temperature Moon is essentially in thermal equilibrium; therefore, the heat emitted can be derived from the equations that are applicable to a planet of uniform temperature.

Substituting the emissive power ($E_p = \sigma T_m^4$) of the element shown in figure B-6 into equation (B39), integrating, and introducing the equality given in equation (B45) gives

$$q_p = \sigma T_m^4 \alpha_p F_4 A_v \quad T_p \neq \text{constant (dark side of Moon)} \quad (\text{B47a})$$

and

$$\frac{q_p}{A_v} = \sigma T_m^4 \alpha_p F_4 \quad T_p \neq \text{constant (dark side of Moon)} \quad (\text{B47b})$$

where F_4 is found by reference to the radiation-configuration-factor table. The incident heat rate is

$$\frac{q_p}{A_v} = \sigma T_m^4 F_4 \quad T_p \neq \text{constant (dark side of Moon)} \quad (\text{B47c})$$

Albedo Heat Flux for an Oriented Vehicle

The solar heat reflected from dA_p and absorbed by A_v (fig. B-6) is

$$dq_{\text{albedo}} = \frac{E_p \cos \theta_p \cos \theta_v dA_p \alpha_s \epsilon_s A_v}{\pi l^2} \quad (\text{B48})$$

Substituting the effective emissive power of the planet element (eq. (B35)), letting $\epsilon_s = 1$, simplifying, and integrating gives

$$q_{\text{albedo}} = SR \alpha_s A_v \iint \frac{\cos \beta \cos \theta_p \cos \theta_v}{\pi l^2} dA_p \quad (\text{B49})$$

The desired relations are found by combining equations (B41) and (B49) to obtain

$$q_{\text{albedo}} = SR \alpha_s F_3 A_v \quad (\text{B50a})$$

and

$$\frac{q_{\text{albedo}}}{A_v} = SR \alpha_s F_3 \quad (\text{B50b})$$

where F_3 is found by reference to the radiation-configuration-factor table (eq. (B41)). The impinging albedo is

$$\frac{q_{\text{albedo}}}{A_v} = SR F_3 \quad (\text{B50c})$$

APPENDIX C

DISCUSSION OF PRACTICAL INTEGRATION STEP-SIZE

The only feasible method to increment time in the numerical integration of the governing temperature equation is to increment the true anomaly and compute the corresponding change in time. (Refer to the section of this report entitled "Transient Temperatures.") This procedure is expressed mathematically as

$$\phi_{n+1} = \phi_n + \Delta\phi \quad (C1)$$

and

$$\Delta t_{n+1} = t(\phi_{n+1}) - t(\phi_n) \quad (C2)$$

where $\Delta\phi$ is a constant.

Since the analysis is applicable to all planet orbits, the magnitude of Δt_{n+1} depends not only on $\Delta\phi$, but also upon variables such as orbit eccentricity, semi-major axis, vehicle position in orbit, and planet being orbited. This dependence is illustrated by comparing times that correspond to a fixed true anomaly for three Earth satellites. For $\Delta\phi = 1^\circ$, Δt is 0.3 minute for Explorer, which was put into a low circular orbit; Δt is 4.0 minutes for Syncom, which is in a large circular orbit; and Δt varies from about 0.3 to 10.0 minutes for Eccentric Geophysical Observatory (EGO), which is in a large, eccentric orbit. These widely varying times are one reason why it is necessary to refer to the length of the integration interval in terms of practicality rather than as a fixed $\Delta\phi$.

There are three conditions that can make the integration interval impractical. Two of these are controlled by the computer program. The conditions and the control on them are described, as follows:

1. The magnitude of Δt_{n+1} can be so large that the one-step truncation error temporarily influences the accuracy of the integration. To minimize computation time, the integration interval should be large as d^2T/dt^2 approaches zero; and to insure an accurate solution, the integration interval should be small as d^2T/dt^2

becomes large. This philosophy is expressed mathematically as

$$\Delta t_1' \propto \left(\frac{d^2 T_n}{dt_n^2} \right)^{-1} \quad (C3)$$

The increment $\Delta t_1'$ is calculated from this equation.

2. The magnitude of Δt_{n+1} can be so large that the equilibrium temperature is reached and vastly exceeded within one integration interval, thus causing the solution to be unstable. This condition occurs if $dT/dt \gg 1$, which could be caused by a vehicle skin with negligible heat capacity ($\rho C_p h \rightarrow 0$) or by an unrealistic initial temperature T_0 . The time increment $\Delta t_2'$ required to reach equilibrium temperature in one interval is calculated.

The intervals $\Delta t_1'$, $\Delta t_2'$, and Δt_{n+1} are compared, and the smallest value is selected and used to integrate the appropriate temperature equation numerically. If the computer program subdivides the time increment Δt_{n+1} into two or more values of $\Delta t'$, the entire procedure is repeated until $\sum \Delta t' = \Delta t_{n+1}$.

3. An impractical interval exists if $\Delta \theta$ is so large that the assumption of constant heat fluxes within the interval is absurd. The computer program will not use time increments that are larger than those computed from the value of $\Delta \theta$ in the input data; therefore, the program user must exercise judgment in choosing $\Delta \theta$ so that it is small enough to be meaningful but large enough to allow fast computation.

APPENDIX D

SUPPLEMENTARY INFORMATION FOR DATA PREPARATION

The following supplementary information is provided to the program user to aid him in preparing data in accordance with appendix I.

Summary of Coordinate Systems

A summary of coordinate systems (described in detail in the section of this report entitled "Celestial Mechanics Theory: Coordinate Systems") is provided in table D-I as a ready reference in preparing data in accordance with appendix I entitled "Program User's Guide for Data Preparation."

Planet Data Used by the Computer Program

Planet data stored in the computer program are listed in table D-II.

Sample Data Preparation for Right Ascension and Declination of the Sun from an Ephemeris

In preparing data, the user has an option in appendix I on the card type 05 to express directly the position of the Sun with respect to the planet X_p , Y_p , and Z_p coordinate axes in terms of α , β , and γ , as shown in figure 9; or they can be calculated by the computer program from five input variables: Ω , ω , i , RA, and DEC (figs. 5 and 8).

The angles RA and DEC, which are illustrated in figure 8, can be obtained for each day of the year from an ephemeris. Sample calculations for Earth, a planet (Mars), and the Moon are illustrated below for July 2, 1963, using the American Ephemeris and Nautical Almanac (ref. 1).

For Earth, RA and DEC are determined from apparent right-ascension and apparent declination data. These data are tabulated at the top of figure D-1 in hours, minutes, and seconds (24 hours equivalent to 360°). Accordingly

$$RA = \frac{6 \text{ hr } 40 \text{ min } 58.88 \text{ sec}}{24 \text{ hr}} \times 360^\circ \approx 100.2^\circ$$

and $DEC \approx 23.1^\circ$.

For a planet other than Earth, longitude and latitude data are used. The data for Mars, given in the middle of figure D-1, are converted as follows:

$$RA = 200^{\circ}20'43.5'' + 180^{\circ} \approx 380^{\circ}21' \approx 20^{\circ}21'$$

$$DEC = -(+0^{\circ}53'42.5'') \approx -0^{\circ}54'$$

For the Moon, colongitude and latitude are used. The data at the bottom of figure D-1 are transformed as follows:

$$RA = 90^{\circ} - \text{colongitude} \approx 90^{\circ} - 34.71^{\circ} \approx 55^{\circ}29'$$

$$DEC = \text{latitude} \approx -0.35^{\circ}$$

Possible User Modifications

For some studies, it may be desirable to modify the internally stored data. The following are means of affecting such modifications for typical changes.

<u>Possible user change</u>	<u>Affected-card location</u>
Tolerance for temperature stabilization check, computer program now uses $0.5^{\circ} R$	Subroutine LOOP, card number DK032490
Planet albedos, computer program now uses values in table E-II	Subroutine FREAD, card numbers DK060870 to DK060950; the subscript for the variable RR corresponds to planet code on 03 card in data input
Moon cold-side temperatures used for variable temperature case, computer program now uses $186^{\circ} R$	Subroutine TINPUT, card number DK020770
Solar constant = $443.0 \text{ Btu/ft}^2\text{-hr}$, used for 1 A. U. from the Sun and as a reference value for other distances	Subroutine TINPUT, card number DK022560

TABLE D-1. - SUMMARY OF COORDINATE SYSTEMS

Vehicle coordinate system					
Vehicle orientation	Origin of axis	Location of X_v	Location of Y_v	Location of Z_v	Location of surface element
Planet oriented	Vehicle center of mass	Line connecting centers of vehicle and planet, positive direction toward planet	In the orbital plane at right angles to X_v -axis, positive direction opposite velocity vector	Normal to orbital plane to form a right-handed coordinate system	Λ' measured from X_v toward Y_v and Ω' measured from Z_v
Sun oriented	Vehicle center of mass	Line connecting centers of vehicle and Sun, positive direction toward Sun	In the orbital plane at right angles to X_v -axis ^a	To form a right-handed coordinate system, with Z_v in same hemisphere as Z_p	Λ' measured from X_v toward Y_v and Ω' measured from Z_v
Spinning ^b					
Planet coordinate system					
Vehicle orientation	Origin of axis	Location of X_p	Location of Y_p	Location of Z_p	Location of vehicle
All	At center of mass of celestial body being orbited	Along major axis of orbit, positive direction toward perigee	Along minor axis of orbit	To form a right-handed coordinate system	Cartesian coordinates X_p and Y_p (may be positive or negative) Polar coordinates True anomaly θ and distance D from principal focus to vehicle
Celestial coordinate system					
Vehicle orientation	Coordinate basis	Location of X_c	Location of Y_c	Location of Z_c	Location of Sun
About Earth	Geocentric reference, center at Earth	Positive X_c toward vernal equinox γ	Perpendicular to X_c in plane of Earth equator	To form a right-handed coordinate system	From ephemeris, geocentric coordinate system, apparent right ascension and apparent declination
About the Moon	Selenographic, reference prime meridian of Moon, center at Moon	Positive along line connecting Moon and Earth	Perpendicular to X_c in plane of Moon equator	To form a right-handed coordinate system	From ephemeris, in selenographic coordinate system, colongitude and latitude of Sun
About a planet	Modified heliocentric, reference, center at planet	Parallel to heliocentric X -axis, positive toward vernal equinox γ	Perpendicular to X_c in plane parallel to ecliptic	To form a right-handed coordinate system	Ephemeris gives position of planet with respect to Sun, 180° added to longitude and sign of latitude changed to obtain position of Sun with respect to planet

^aFor the special case in which the rays of the Sun are normal to the orbital plane, in addition to the stated requirements, the positive Y_v -axis must be in the same direction as the positive Y_p -axis.

^bThe X_v -, Y_v -, and Z_v -axes are not applicable to spinning vehicles.

TABLE D-II. - PLANET DATA USED BY COMPUTER PROGRAM^a

Planet code	Planet	Distance from Sun, ft	Radius, ft	Albedo	GM _p , ft ³ /sec ²
1	Earth	48.89×10^{10}	20.9×10^6	0.35	141×10^{14}
2	Moon	48.89	5.702	.047	1.73
3	Jupiter	255.3	229.3	.51	44 900
4	Mars	74.81	10.87	.148	15.20
5	Mercury	19.03	8.151	.058	7.66
6	Neptune	1475	81.51	.62	2435
7	Saturn	467.9	188.7	.50	13 450
8	Uranus	941.3	83.6	.66	2058
9	Venus	35.43	20.34	.76	114.8

^aThe distance from Sun and the mass data for Moon are taken from reference 7. The radius, mass, and albedo data, except for the Earth and Moon albedo that are in accordance with Apollo specifications, are taken from reference 8.

SUN, 1963						27
FOR 0 ^h EPHEMERIS TIME						
Date	Apparent Right Ascension	Apparent Declination	Radius Vector	Semi- diameter	Equation of Time Apparent - Mean	
July 1	6 36 50.66	+23 10 22.5	1.016 6913	15 45.40	-3 32.12	-11.66
2	6 40 58.88	23 06 31.8	1.016 7036	15 45.39	3 43.78	11.40
3	6 45 00.83	23 02 10.8	1.016 7115	15 45.38	3 55.18	11.10
4	6 49 14.49	22 57 37.7	1.016 7154	15 45.38	3 55.18	11.10
5	6 53 27.84	22 52 22.2	1.016 7154	15 45.38	3 55.18	11.10

MARS, 1963								173
HELIOCENTRIC POSITIONS FOR 0 ^h EPHEMERIS TIME								
MEAN EQUINOX AND ECLIPTIC OF DATE								
Date	Julian Date	Longitude	Latitude	Radius Vector	Orbital Longitude	Daily Motion	Orb. Lat.	
July 2	243 8212.5	200 20 43.5	+0 53 42.5	1.617 224	200.33278	0.463 189	+0.02	
6	8216.5	202 12 00.9	0 50 32.0	1.613 476	202.18981	0.465 342	0.01	
10	8220.5	204 04 01.7	0 47 17.3	1.609 611	204.05563	0.467 579	+0.01	
14	8224.5	205 56 29.1	0 43 58.7	1.605 635	205.93044	0.469 896	0.00	
		0 30.3		1.601 540			-0.01	

MOON, 1963											314
EPHEMERIS FOR PHYSICAL OBSERVATIONS											
FOR 0 ^h UNIVERSAL TIME											
Date	Age	The Earth's Selenographic		Physical Libration	The Sun's Selenographic		Position Angle of		Frac- tion Illumi- nated		
		Longitude	Latitude	Lg. Lt. P.A.	Colong.	Lat.	Axis	Bright Limb			
July 1	9.5	+1.28	-6.71	0 -4 -1	22.50	-0.26	19.57	292.1	0.70		
2	10.5	+0.02	6.29	0 4 -1	34.71	0.35	16.38	290.3	0.78		
3	11.5	-1.20	5.60	0 4 0	40.92	0.32	12.47	287.7	0.86		
4	12.5	2.32	4.65	0 4 0	59.12	0.30	7.91	284.4	0.92		
5	13.5	0.99	3.49	0 4 0	71.32	0.27		280.7	0.96		

Figure D-1. - Ephemeris data required for RA and DEC sample calculations.

APPENDIX E

COMPUTER PROGRAM ORGANIZATION

The FORTRAN V computer program is divided into two sections called LINK1 and LINK2. Both sections reside in core at the same time; although, LINK2 is not entered unless Stromberg-Carlson 4020 (SC-4020) plots are requested. (Refer to appendix H.) The links are not to be confused with a FORTRAN II technique in which two sections of a program share core alternately. It is important that the main program of LINK1 is named PILOT and the main program of LINK2 is named MAIN2. The MSC SC-4020 programming package, which is not included in the deck, is used by LINK2. On the Univac 1108 computers at NASA MSC, the SC-4020 routines are available from a system tape.

Simplified flow charts of LINK1 and LINK2 are given in figures E-1 and E-2. The hierarchy of subroutines in each link is shown in figures E-3 and E-4. The source deck contains comment cards that may also help the user if he should desire greater detail.

LINK1

Main routine and subroutines. - PILOT is the main routine for LINK1. The function of PILOT is to define necessary constants and call two major subroutines, TINPUT and LOOP. If plots are not requested, program control passes directly from LOOP to PILOT. When plots are requested, control goes from LOOP to MAIN2 to PILOT.

All input to the program is controlled by TINPUT. The output of headings, comments, and information used to explain and define each case are also controlled by TINPUT. Subroutine SIGBET is also called by TINPUT so that the values of Σ , β , ϕ_{in} , and ϕ_{out} will be available for printing with the case headings.

Caption pages are printed by HEAD, and FREAD is called by HEAD to read in permanent data. Permanent data such as radiation-configuration-factor tables, physical constants for the Moon and eight planets, alphabetical heading information, and frequently used constants are read in by FREAD. Also, general comments are printed by FREAD, and TABLE is called by FREAD to read in material property tables. Tables of material properties are read by TABLE and rewritten on the output tape.

New tables of internal heat loads are set up by QIN whenever the tables are to be read in. Continuation cards are read until the table contains the specified number of entries. The table is also written on a scratch tape to be copied later on the output tape.

The angular position of the Sun relative to the orbital plane is computed by SIGBET. Two types of input data are accommodated: α , β , γ or Ω , ω , i , RA, and DEC. This routine also calls FIND to locate the Sun-shade points and SUNOR to transform element angles for the Sun-oriented cases.

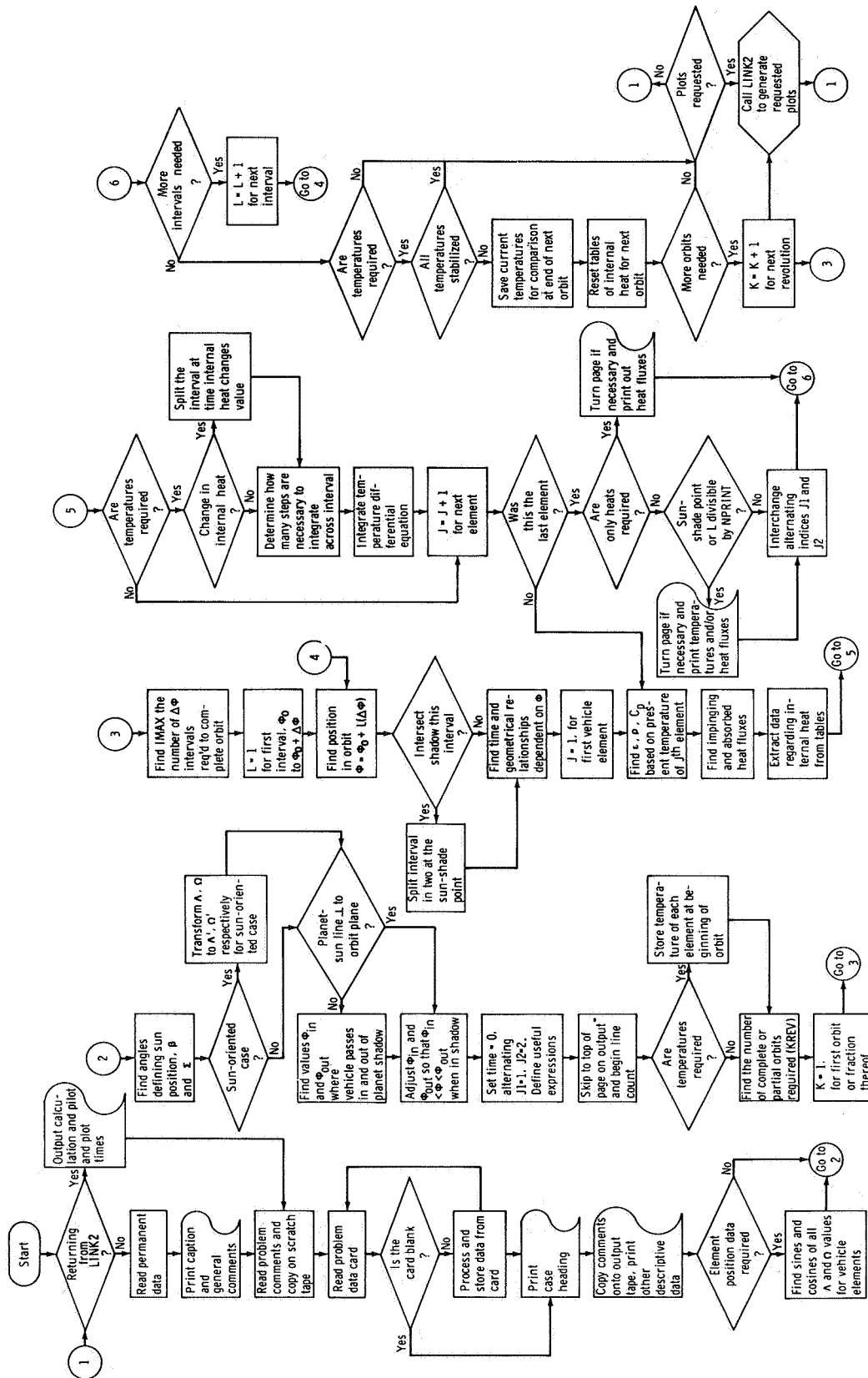


Figure E-1. - Simplified flow chart of LINK1.

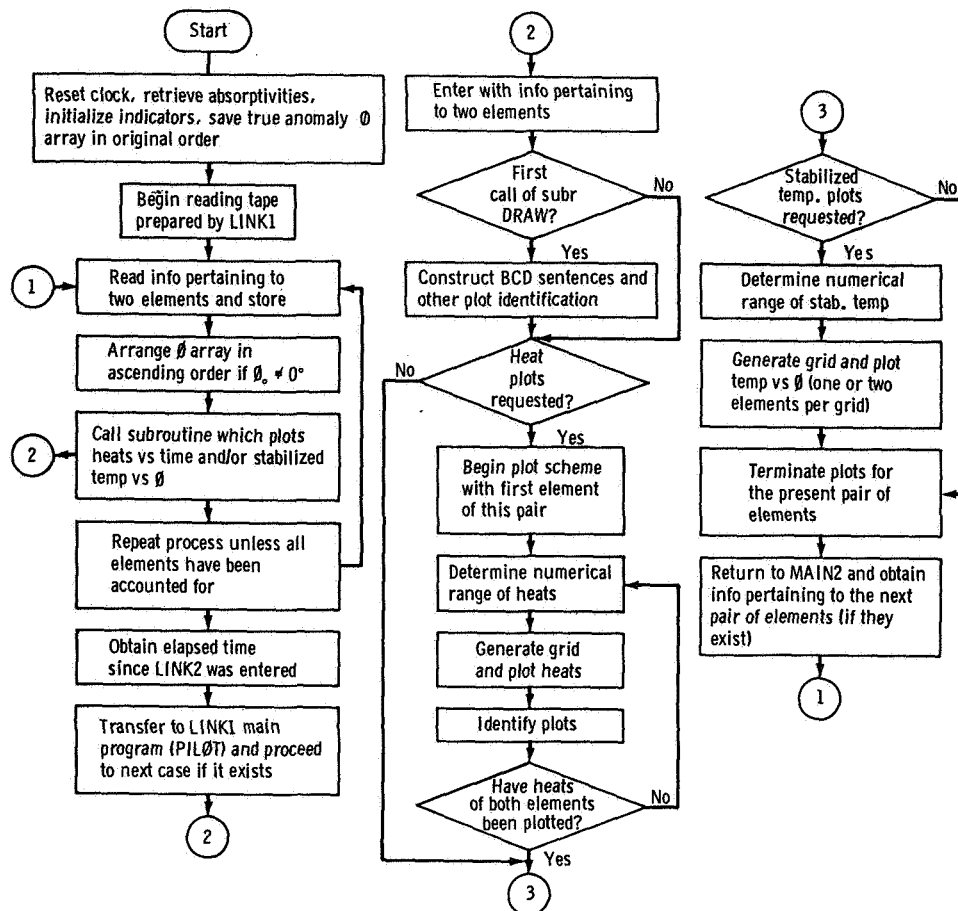


Figure E-2. - Simplified flow chart of LINK2.

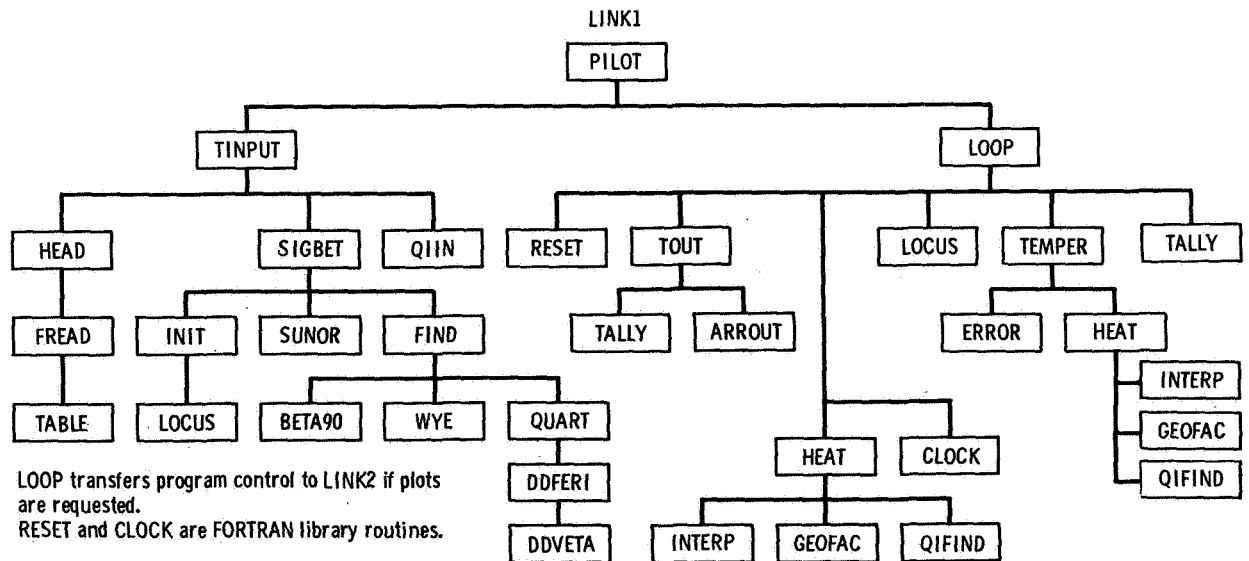
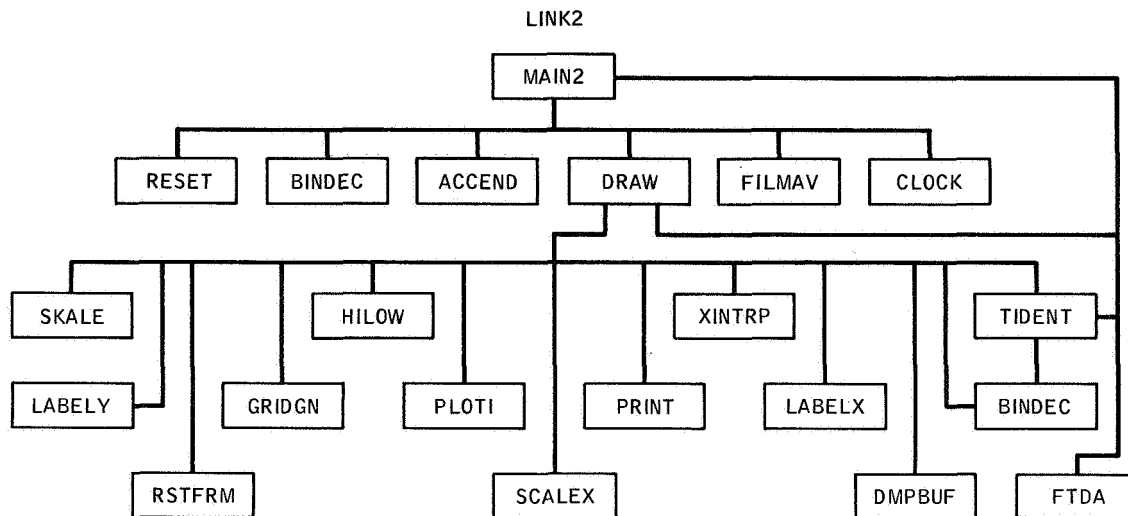


Figure E-3. - Chart of subroutine hierarchy for LINK1.



When LINK2 has completed its task, MAIN2 transfers program control to PILOT in LINK1.

RESET and CLOCK are FORTRAN library routines.

BINDEC, FILMAV, LABELY, LABELX, GRIDGN, PLOTI, PRINT, RSTFRM, SCALEX, and DMPBUF are contained in MSC SC-4020 programming package.

Figure E-4. - Chart of subroutine hierarchy for LINK2.

The quadrants of $\phi_o, \phi_{in}, \phi_{out}$ are adjusted by INIT so that $\phi_{in} < \phi < \phi_{out}$ whenever the vehicle is in the shadow. The determination whether the vehicle is initially in the Sun or in the shadow is also made by INIT.

Time, altitude, and the angle θ_s are calculated by LOCUS from ϕ , which represents the vehicle position. For oriented vehicles, ϕ_c and δ are found. The angle ϵ is also found for Sun-oriented cases. The symbols ϕ_c, θ_s , and ϵ are arguments of the radiation-configuration-factor tables, and δ is necessary to determine solar heat flux. If planet temperature T_p is variable, it is also found.

The angular coordinates of the satellite elements are converted by SUNOR from the input values to values acceptable to the program logic for Sun-oriented satellites.

The Sun-shade points ϕ_{in} and ϕ_{out} are located by FIND. If the Sun is in the orbital plane, BETA90 is called. Otherwise, QUART is called to determine the X_p value for the coordinate and WYE is called to find the Y_p value for the coordinate. Each point is examined to insure that it is not on the sunward side of the planet. The Sun-shade points for cases in which the Sun is in the orbital plane are computed by BETA90.

The Y_p values of the Sun-shade points for which the X_p values were calculated in QUART are found by WYE. Pairs of X_p and Y_p values are then examined to insure that they satisfy the equations of both the orbit ellipse and the shadow ellipse.

The roots of a quartic equation are solved for by QUART. The quartic arises when the equations for the orbit ellipse and the shadow ellipse are solved simultaneously for X_p .

DDFERI is a double-precision quartic-factorization program from the SHARE library. DDVETA is a double-precision cubic-factorization program from the SHARE library. DDVETA is required by DDFERI.

A heading to be printed at the top of an output page, and an output line count is begun by LOOP. Unless temperature calculations are not required, the maximum time interval is found for possible use in subroutine ERROR. The initial temperature of each surface element is stored for comparison with the corresponding temperatures one full orbit later.

In LINC, the number of intervals of width $\Delta\theta$ necessary to make one orbit is found, and in the routine KREV, the number of complete or fractional orbits that will be required, provided that all temperatures do not stabilize earlier, is found.

The vehicle is then allowed to move counterclockwise around its orbit in steps $\Delta\theta$. For each step, LOCUS is called to find the time and other information depending on θ , and heat fluxes and temperatures are determined for each surface element. Output may be written after each interval $\Delta\theta$, or it may be deferred for two or more intervals under the control of the input parameter NPRINT.

If a Sun-shade point is reached, output occurs automatically. If temperatures are being found, the integration of the differential equation over the interval is broken into two steps at the Sun-shade point.

At the end of each complete orbit, temperatures are compared with corresponding temperatures one orbit earlier. If these agree to within 0.5°R , the case is considered finished. If not, the new temperatures are stored, the tables of internal heat loads are reset for a new orbit, and the program loops back unless the specified number of orbits read in has been exceeded. When the case is finished, control passes to PILOT or to LINK2, depending on whether or not plots were requested.

Lines of output, aside from comments, headings, and case-identifying information, are counted by TALLY. The printer carriage is restored when an output page is nearly filled. All output, except for comments, headings, and case-identifying information, are printed by TOUT. Output includes θ , time, and either heat loads or temperatures, or both. With the normal option, data are printed in columnar format. However, an option to allow output to be printed in block form is also provided. ARROUT is an auxiliary subroutine for output of arrays of data in block form.

The subroutine HEAT is called by the subroutine TEMPER to obtain the net heat flux into each vehicle surface element and the internal heat load. The differential equation for temperature is then integrated over a given interval $\Delta\theta$ of the orbit. If the interval is too long or if internal heat loads are discontinuous within the interval, the interval is subdivided into two or more smaller steps.

A theoretical approximation to the truncation error resulting from the numerical integration is computed by ERROR. This routine is normally suppressed.

Impinging heat fluxes for a specified satellite surface element are found by HEAT. In cases for which temperature calculations are required, absorbed heat fluxes are also found, and QIFIND is called to determine the internal heat load.

Values of emittance, density, or specific heat are found by INTERP for a specified temperature by means of linear interpolation in material-property tables.

Tables of radiation configuration factors are interpolated for by GEOFAC to find a value corresponding to given ϵ , θ_c , θ_s , and altitude. Values of these parameters outside the range of tabulated entries are extrapolated for by the computer program. This feature is useful for θ_s greater than 90° . However, if the factor found results in a negative heat flux, subroutine HEAT will replace it by zero.

Determination of whether the internal heat for the surface element in question is varied during the interval of integration is made by the routine QIFIND. If the element is varied, initial and final heat loads and the time at which switching occurs are provided by QIFIND.

Function subroutines. - The arccosine is found in degrees of the argument by ARCOS. ARCOS always extends from 0 to 180, inclusive. PHIFN is the ratio of temperature change to time increment $\Delta T/\Delta t$, as calculated from the numerical-integration algorithm. FOFXY is the derivative of temperature with respect to time for a given temperature and location in orbit. DELTA and GFN are auxiliary functions used by ERROR.

LINK2

MAIN2 is the main routine for LINK2 and is reached whenever SC-4020 plots are requested. The primary functions of MAIN2 are to control LINK2 and to retrieve data from a scratch tape constructed by LINK1.

Identified plots of heats compared with time, stabilized temperature compared with θ , or both are constructed by subroutine DRAW. The θ array is arranged in ascending order for plot purposes by ASCEND.

A linear interpolation for θ , corresponding to a chosen time, is performed by XINTERP.

Minimum and maximum values of heats and temperatures are determined by HILOW for plot scaling purposes. A suitable scale and numeric labels are established by SKALE for use on plot axes. Sentences (arrays) describing temperature plots are constructed by TIDENT. IBM binary-coded-decimal data are converted to Univac field data by FTDA.

APPENDIX F

OPERATING INSTRUCTIONS

The following operating instructions are pertinent when submitting computer runs.

Tape Usage

The computer tapes used by the program are as follows:

<u>Logical no.</u>	<u>Use</u>
5	Input
6	Output
4	Scratch tape
9	Scratch tape
11	Scratch tape

Run Time on Computer

Running time varies greatly with the number and types of cases to be run. Until experience with the program enables the user to make more accurate estimates, the following suggestions are offered as a guide for running time: 0.5 minute per case for up to 10 elements and 0.5 minute per case for each additional 20 elements.

Job Submittal Sheet

When submitting the computer program to be run on one of the Univac 1108 computers at NASA MSC, the operation instructions should be as shown in figure F-1 (MSC form 588).

INSTRUCTIONS FOR SCIENTIFIC COMPUTER RUNS

(DO NOT FILL IN SHADED AREAS)

[illegible]

MSC FORM 588 (REV APR 67) PREVIOUS EDITION MAY BE USED.

Figure F-1. - MSC form 588.

APPENDIX G

SAMPLE CASES

The following three sample cases are given to demonstrate to the program user the input and output format of the computer program. The sample cases were selected to show the various options available to the program user and are not necessarily realistic cases.

Problem Definition of Sample Cases

Case 501. - The problem is to determine the incident heats, absorbed heats, and temperatures of a Moon-oriented vehicle orbiting the Moon. Plots of absorbed heats versus ϕ and stabilized temperatures versus time are requested. Four surface elements are to be analyzed. Output data are to be printed every 10° for one orbit. Further specifications are as follows:

Moon surface temperature — variable

Sun position — given in an ephemeris for July 2, 1963

$$i = 10^\circ$$

$$\omega = 130^\circ$$

$$\Omega = 70^\circ$$

Maximum orbit altitude = 165.0 n. mi.

Minimum orbit altitude = 8.73 n. mi.

$$\phi_o = 0^\circ$$

$$\Delta\phi = 10^\circ$$

Element 1:

$$\Lambda' = 30^\circ, \Omega' = 90^\circ$$

$$T_o = 530^\circ \text{ R}$$

$$h = 0.01 \text{ ft}$$

$$Q_g = 0$$

$$\text{Surface area} = 1.0 \text{ ft}^2$$

Material — uncoated aluminum

Element 2:

$$\Omega' = 180^\circ (\Lambda' \text{ is undefined})$$

$$T_o = 375^\circ \text{ R}$$

$$h = 0.005 \text{ ft}$$

$$\text{Surface area} = 2.0 \text{ ft}^2$$

All other variables same as element 1

Element 3:

$$\Lambda' = 210^\circ, \quad \Omega' = 90^\circ$$

$$T_o = 540^\circ \text{ R}$$

All other variables same as element 1

Element 4:

$$\Omega' = 0^\circ (\Lambda' \text{ is undefined})$$

$$T_o = 405^\circ \text{ R}$$

$$h = 0.005 \text{ ft}$$

$$\text{Node number} = 44$$

All other variables same as element 1

Case 502. - The problem is to determine the incident heats, absorbed heats, and temperatures of a spinning vehicle orbiting the Earth. Plots are requested. Output data are to be printed every 30° for 7.25 orbits unless the temperatures stabilize first. Two different internal heat loads are to be compared. (This will be accomplished by running a single case using two "elements" that refer to different internal heat tables.) The following are further specifications:

Sun positions — specified in terms of α , β , and γ ; date not needed

$$\alpha = 80^\circ, \quad \beta = 90^\circ, \quad \gamma = 170^\circ$$

Maximum orbit altitude = minimum orbit altitude = 181.0 n. mi.

$$\phi_o = 280^\circ$$

$$\Delta\phi = 6^\circ$$

Element 1:

$$\Lambda' = 30^\circ, \quad \Omega' = 90^\circ$$

$$T_o = 530^\circ \text{ R}$$

$$h = 0.01 \text{ ft}$$

$$Q_g = 0 \text{ for 15 min, then } 20 \text{ Btu/ft}^2\text{-hr for 15 min}$$

Cycle continuing throughout each orbit

Material — uncoated aluminum

$$\text{Surface area} = 1.0 \text{ ft}^2$$

Element 2:

$$\Omega' = 180^\circ (\Lambda' \text{ is undefined})$$

$$T_o = 375^\circ \text{ R}$$

$$Q_g = 10 \text{ Btu/ft}^2\text{-hr continuously}$$

$$\text{Surface area} = 2.0 \text{ ft}^2$$

All other variables same as element 1

Case 503. - The problem is to determine the temperatures of a Sun-oriented vehicle orbiting the Earth. Output data are to be printed in block form every 30° for one-half of an orbit. Four surface elements are to be analyzed. The following are further specifications:

α, β, γ , minimum orbit altitude, maximum orbit altitude — same as in case 502

$$\phi_o = 190^\circ$$

$$\Delta\phi = 2.5^\circ$$

Element 1:

$$\Lambda' = 0^\circ, \quad \Omega' = 90^\circ$$

$$T_o = 530^\circ \text{ R}$$

$$h = 0.01 \text{ ft}$$

$$Q_g = 0$$

Material — aluminum, painted black

$$\text{Surface area} = 1.0 \text{ ft}^2$$

Element 2:

Material — aluminum, painted white

$$T_o = 375^\circ$$

$$\text{Surface area} = 2.0 \text{ ft}^2$$

All other variables same as element 1

Element 3:

$$\Lambda' = 180^\circ$$

$$T_o = 540^\circ \text{ R}$$

All other variables same as element 1

Element 4:

$$\Lambda' = 180^\circ$$

Material — aluminum, painted white

$$T_o = 405^\circ \text{ R}$$

Node number = 44

All other variables same as element 1

Listings of Input Data

Permanent data. - The permanent data consist of 147 permanent cards that must always be present when running the computer program. They are described in the section of this report entitled "Data Deck Preparation." A complete listing is given in table G-I.

Sample-case data. - A listing of the sample-case data is defined in table G-II. The data format is described in appendix I. Several material-properties tables were prepared; however, not all the tables were used in the analysis.

Output

Reproductions of printed and plotted output from the sample cases are presented in table G-III and in figures G-1 and G-2.

TABLE G-I. - LISTING OF PERMANENT INPUT DATA

* DATA
95318956825473016126476532591645082301128422791472946452541442112874145907770253
71216691616754554577355724231271076003675792544350164437372328821976110207280429
25492395220719441625127809340619048003591102103609500837070805730440031602600209
05140482044203910335027602180163013801140158014801360121010500890072005600480041
82437795721164075409424629551620097001787241689263995712485138422750150908900351
61675896548949154192333723701358087604675016480944844022343827331959117808190511
22072119197717661503121109120625049403760954091308460754064705320416030402520205
04450423039103500302025202020153012901080137012901190107009400790065005000430037
82437788720163955394423029371554084101677241688063825690482438132680147608610334
61675883546948894161330323341323084704475016479644653998340927021925114807940492
22072113196817551490119708980613048303670954091108430750064205270411030002490201
04450422039003480301025102000151012801070137012901190107009300790064005000430037
82437770717563615354418428871455079001377241684863355629475237312595138007780285
61675846541548194078320922341229076703925016476144143932333126151835106607250441
22072097194417251455115908600581045503430954090508340740063005140398028902390193
04450420038703440296024601950147012401030137012901190106009200780063004900420036
82437746713863145298412128191430072100907241680462715547465436202475126606710217
61675795534047243964308120981101065803185016471443443843322424961711095406310371
22072074191216831407110708090536041603100954089708230725061304960381027402250181
04450417038303390290023901890141011900990137012801180105009100770062004800410035
82437721710262685243405827511359065200637241676062075464455635092355114505650153
6167574226646283650295219620973054902445016466642743753311723781588084205360301
22072052187916421359105507570491037602780954088908110710059604780363025802110169
04450414037803330284023301830136011400940137012701170104009000750061004700400034
8243770370766234520240132701130706030039724167286160540448434282267105604880107
61675707521245583766285818630879047001895016463142233687303922911497076104670250
22072036185516121324101607190458034802540954088308020699058304650350024702010160
04450412037503290279022801780132011100910137012701160103008900750060004600400034
8243769670666221518739962683128805840030724167176143538244583398223510240450090
61675693519245333736282418260844044001695016461842053663301022591464073104420232
22072030184616011311100307050446033702450954088007990695057904600346024301970157
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60045708529647234006316822341220067901964905472744203979341727513000117007600355
3889379835773248282123341675105007580441301729792823258226218851390093806910456
12601259120111160982082006460464037502920551054605160470041303490281021101770146
02570251023602140189019001310101008600720079007600710064005700480040003100270023
60045699528347063986314622091206068601774905471143973949338127111962113807210325
38893778354932122778226516641027070704063017296027952545221818301365088706470423
12601249118610860955079506220443035702770551054205100463040503400272020401710140
02570250023402120186015701280098008400700079007500700064005600480039003100270023
60045675524846613933308521431136061601344905466843343868328526021837101506060252
38893726347231142660212415180886058003253017290727182446210017001224074505290345
12601222114610350895072705560387030702360551053204950444038303170250018401530125
02570246022802050178014901200091007700650079007500690062005400460038002900250022
60045642520045993659300220531042052500784905460942483757315324521675085204590148
38893654336829792499194313250700042001983017283526132311193915031035058003810222
12601184109109640012063704640306023501750551051804750418035402860220015801300104
02570241022101950167013801090081006900570079007400680060005200440036002800240020
60045609515245383786291919630948043500524905455041623647302123031514069303200074
38893582326328442339176211320521027101053017276325072176177813230847041402430123
12601146103608940729054803760230017001210551050304540393032402550189013101060083
02570235021301860157012700980072006000490079007300660059005000420034002600220019
60045585511644923732285818970880037000124905450740993566292421941396057702220026
38893530318627452221162909920391016600423017271024302077166011910710029301450065
12601119099608420668048203110174012200800551049304390374030202320167011200880068
02570232020801790149011800900065005400430079007200650057004900400032002400210017

TABLE G-I. - LISTING OF PERMANENT INPUT DATA - Continued.

60045576510444763712283618730855034700034905449140763536288921541353053401870010
38893511315827092178158109400344012800203017269124022040161711430660024901090032
12601108098108230646045802870154010400660551048904340367029502230159010500820062
02570230020601770146011500870062005100410079007200650057004800400031002400200017
35523393315728252407191613670890059501612538247523302114183414981206077905530264
17701765168115471365120409400644048702971198122111771097098308640696050704060278
03060332033003240304027402360200016401300084009500980100009800950086007600650048
00260031003200320031002900250022002000170004000500050005000500050004000400030002
35523385314628112390189713460777044301422538246323122091180714671105066304430233
17701751166115211335112608510553040602621198120811581073095508160633043803430245
03060326032203100284025502060163014001120084009300920090008400760065005500490042
00260030003200300028002500230019001700150004000500050005000500040004000300030002
35523365311627732345184612900690038200982538243022642029173213830992056403520161
17701714160714511251101507380449031001811198117311071007087707150532034602550169
03060310029802750243020501650122010000790084008700840078007000600051004000340029
00260027002700250023002000170014001200100004000400040004000400030003000200020001
3552337307627212283177612140608030500512538238521981944163112690865043802380084
17701663153213561137088305990316019100941198112510370917077005930406022701500088
03060288026502330194015201110073005600410084007800720063005300430033002300190015
0026002400220020001700140011000800060005000400040003000300020002000100010000
35523310303526692221170611380529023000172538233921321859153011540742031701350028
17701612145812601023075504640193009200321198107809670828066304750284012300660030
03060265023201910146010100620032002200140084007000600048003600240016000900070005
0026002200180014001100080005000300020001000400030003000200010000000000000000000
3552329030062631217516541083047101770022538230620831797145610700651023100680003
17701575140411900940066103650109003200041198104309160762058503890197005500180004
03060249020901610111006500300009000400010084006400510037002700160006000200010000
0026001900140010000600020001000
35523282299526172159163510620450015700002538229420651774142810390618019900440000
17701561138411640909062703290079001300001198103008980738055603570165003100040000
030602430200015000980052001800020000000000840062004800340020001000030000000000000
002600180014000900050002000
1351129912141091093607520545032102500880746074007030645056604710362024101780113
03690379036703430310026602150157012600940157016801660159014701300110008600730059
0
0
13511295120810840927074105340310021000720746073506950635055504580347022601650098
03690375036103350300025502030144011700810157016501620154014101230102007800650050
0
0
13511284119210630902071305030295018800350746072106750608052304220308020501480054
03690363034303130273022501800135010500500157015801510140012401040930082007000031
0
0
13511269117010340868067504620265016000250746070106460572047903730280018501280048
03690347032002830237018401450112009200350157014801360120010100850073005400450020
0
0
13511254114710060834063704200189007600150746068206180535043603240220007700300008
03690331029602520201014300810028001100030157013801210101007800520025000900450001
0
0
13511243113109850809060903900158004900010746066805970508040402880160004200070000
03690319027902300174011300510007000100000157013001100087006100340011000000000000
0
0
13511239112509780800059803780130002400050746066205890499039302750145002400050001

TABLE G-I. - LISTING OF PERMANENT INPUT DATA - Concluded.

```

036903140272022201640100004000020000000000157012701060082005500200000000000000000
0
0
01440141013201200104008500640040002800150015001600150014001300110009000600050004
0
0
0
01440140013101190103008300620038002500130015001600150014001200110008000600050003
0
0
0
01440138012901150098007900560032001500080015001500140013001100090007000400030002
0
0
0
01440135012501100092007200490020000700020015001400130012001000070005000300010000
0
0
0
01440133012101050086006500420017000400000015001400120010000800060004000000000000
0
0
0
01440131011801020082006000360011000100000015001300110009000700040002000000000000
0
0
0
01440130011701000001005800340007000100000015001300110009000600040001000000000000
0
0
0
PLANE SATELLITE 1 PLANET TEMPERATURE IS CONSTANT      E IS VARIABLE      S SUN
ORIENTED      S SPINNING      S PLANET ORIENTED T EARTH      T MOON      T JUPI
TER      T MARS      T MERCURY      T NEPTUNE      T SATURN      T URANUS      T VENUS

```

TABLE G-II. - LISTING OF SAMPLE CASE INPUT DATA

10MISCELLANEOUS COMPUTATIONS FOR SPACECRAFT IN ORBIT AROUND EARTH AND MOON
10MATERIAL PROPERTIES

```
0602
0.      .055      9000.      .055      10000.      .3      KEAL1
0.      .970      10000.     .970      10000.      .3      EBP1
0.      .930      9000.      .93      10000.     .180     EWP1
0.      .36      9000.      .36      10000.     .49      EEX1
0.      .14      9000.      .14      10000.     .183     EVN1
+0      +00+22    +00+1    +05+22    +00      ET31
0.      172.8     10000.     172.8     10000.     .22      KRAL1
0.      .22      10000.     .22      10000.     .22      KCAL1
+0      +00+212   +00+56   +03+212   +00+66   +03+215   +00      CT31
+86     +03+228   +00+106   +04+24    +00+1    +05+24    +00      CT32
+0      +00+1728  +03+1     +05+1728  +03      RT31
```

10COMPUTATION OF HEATS AND TEMPERATURES FOR FOUR ELEMENTS ON THE SURFACE OF AN
10ALUMINUM VEHICLE ORBITING THE MOON

10TEST CASE 501

```
01 501 4 1
02      0.      10.      1.
03 2 1 1      165.      8.73
04 1 62130.    90.      530.    .01      1.
04 4 6210.      0.      405.    .005     1.      44.
04 3 621210.   90.      540.    .01      1.
04 2 6210.     180.     375.    .005     2.
05      10.     130.     70.     55.29   -0.35
07 4
```

10 CASE TO STUDY EFFECT OF SWITCHING FREQUENCY OF INTERNAL HEAT LOADS

10TEST CASE 502

```
01 502 4 1
02 5      280.     6.      7.25
03 1      181.      181.
04 1 2
04 2 3      .01
05 3      30.      90.      170.
06 2 8      0.      15.      20.      30.      0.      45.      20.
0660.      0.      75.      20.      90.      0.      105.      20.
06120.      0.
06 3 1      10.      90000.    10.
07 2
```

10TEST CASE 503

```
01 503 1
0212 359 190.    2.5     .5
03 1-1      131.      181.
04 1 2110.
04 2 3110.      90.
04 3 211180.
04 4 311180.     90.      .01
07 4
```

TABLE G-III. - PRINTED OUTPUT FROM SAMPLE CASES

MISCELLANEOUS COMPUTATIONS FOR SPACECRAFT IN ORBIT AROUND EARTH AND MOON									
MATERIAL PROPERTIES									
COATING MATERIAL	1 IS	KEAL1							
0.0000000	5.5000000-02	9.0000000+03	5.5000000-02	1.0000000+04	3.0000000-01				
0.0000000									
COATING MATERIAL	2 IS	EBP1							
0.0000000	9.7000000-01	1.0000000+04	9.7000000-01	-0.0000000					
COATING MATERIAL	3 IS	EMP1							
0.0000000	9.3000000-01	9.0000000+03	9.3000000-01	1.0000000+04	1.8000000-01				
0.0000000									
COATING MATERIAL	4 IS	EEX1							
0.0000000	3.6000000-01	9.0000000+03	3.6000000-01	1.0000000+04	4.9999999-01				
0.0000000									
COATING MATERIAL	5 IS	EVN1							
0.0000000	1.3999999-01	9.0000000+03	1.3999999-01	1.0000000+04	1.8300000-01				
0.0000000									
COATING MATERIAL	6 IS	ET31							
0.0000000	2.2000000-01	1.0000000+04	2.2000000-01	-0.0000000					
SUBSTRATE MATERIAL	1 IS	KRAL1							
0.0000000	1.7280000+02	1.0000000+04	1.7280000+02	-0.0000000					
SUBSTRATE MATERIAL	1 IS	KCAL1							
0.0000000	2.2000000-01	1.0000000+04	2.2000000-01	-0.0000000					
SUBSTRATE MATERIAL	2 IS	CT32							
0.0000000	2.1200000-01	5.5999999+02	2.1200000-01	6.6000000+02	2.1500000-01				
8.6000000+02	2.2799999-01	1.0600000+03	2.3999999-01	1.0000000+04	2.3999999-01				
0.0000000									
SUBSTRATE MATERIAL	2 IS	RT31							
0.0000000	1.7280000+02	1.0000000+04	1.7280000+02	-0.0000000					

TABLE G-III. - PRINTED OUTPUT FROM SAMPLE CASES - Continued

CASE NO. 501
 COMPUTATION OF HEATS AND TEMPERATURES FOR FOUR ELEMENTS ON THE SURFACE OF AN
 ALUMINUM VEHICLE ORBITING THE MOON

TEST CASE 501

PLANET MOON SATELLITE IS PLANET ORIENTED PLANET TEMPERATURE IS VARIABLE

MAX. # ORBIT ALT. (NM) # MIN. PHIO DPHI SIGMA BETA PHIN PHOUT

165.00 8.73 .00000 10.00000 215.44510 87.81780 318.71350 100.61337

INCLINATION= 10.00000 ARGUMENT OF PERIFOCUS= 130.00000 LONG. OF ASC. NODES= 70.00000

RIGHT ASCENSION= 55.29000 DECLINATION= -.35000

ELEMENT	COATING	SUBSTRATE	DUTY	CYCLE	LAMDA	OMEGA	T (G)	THICKNESS	AREA	NODE NO.
1	6	2	1	30.	90.	530.	.01	1.		
4	6	2	1	0.	0.	405.	.005	1.		44.
3	6	2	1	210.	90.	540.	.01	1.		
2	6	2	1	0.	180.	375.	.005	2.		

MAXIMUM NO. OF ORBITS REQUESTED= 1.000

TABLE G-III.- PRINTED OUTPUT FROM SAMPLE CASES - Continued

PHI	TIME	TEMPERATURE	INCIDENT (BTU/HR-FT**2)				ABSORBED (BTU/HR)			
			QSOLAR	QALBEDO	QPLANET		QSOLAR	QALBEDO	QPLANET	QTOTAL
10.0	2.94	1 526.12	.00	.00	1.76		.00	.00	.39	.39
		2 373.07	.00	.00	.80		.00	.00	.35	.35
		3 535.77	.00	.00	.04		.00	.00	.01	.01
		44 402.36	.00	.00	.80		.00	.00	.18	.18
20.0	5.90	1 522.34	.00	.00	1.75		.00	.00	.38	.38
		2 371.17	.00	.00	.78		.00	.00	.34	.34
		3 531.66	.00	.00	.04		.00	.00	.01	.01
		44 399.79	.00	.00	.78		.00	.00	.17	.17
30.0	8.88	1 518.64	.00	.00	1.73		.00	.00	.38	.38
		2 369.29	.00	.00	.76		.00	.00	.34	.34
		3 527.63	.00	.00	.03		.00	.00	.01	.01
		44 397.25	.00	.00	.76		.00	.00	.17	.17
40.0	11.90	1 515.00	.00	.00	1.70		.00	.00	.37	.37
		2 367.43	.00	.00	.74		.00	.00	.32	.32
		3 523.68	.00	.00	.03		.00	.00	.01	.01
		44 394.75	.00	.00	.74		.00	.00	.16	.16
50.0	14.97	1 511.40	.00	.00	1.67		.00	.00	.37	.37
		2 365.57	.00	.00	.71		.00	.00	.31	.31
		3 519.78	.00	.00	.03		.00	.00	.01	.01
		44 392.27	.00	.00	.71		.00	.00	.16	.16
60.0	18.10	1 507.83	.00	.00	1.63		.00	.00	.36	.36
		2 363.72	.00	.00	.67		.00	.00	.29	.29
		3 515.92	.00	.00	.02		.00	.00	.00	.00
		44 389.80	.00	.00	.67		.00	.00	.15	.15
70.0	21.30	1 504.28	.00	.00	1.59		.00	.00	.35	.35
		2 361.86	.00	.00	.62		.00	.00	.27	.27
		3 512.09	.00	.00	.02		.00	.00	.00	.00
		44 387.34	.00	.00	.62		.00	.00	.14	.14
80.0	24.58	1 500.75	.00	.00	1.54		.00	.00	.34	.34
		2 359.99	.00	.00	.57		.00	.00	.25	.25
		3 508.28	.00	.00	.01		.00	.00	.00	.00
		44 384.87	.00	.00	.57		.00	.00	.13	.13
90.0	27.94	1 497.22	.00	.00	1.49		.00	.00	.33	.33
		2 358.10	.00	.00	.52		.00	.00	.23	.23
		3 504.49	.00	.00	.00		.00	.00	.00	.00
		44 382.41	.00	.00	.52		.00	.00	.12	.12
100.0	31.40	1 493.70	.00	.00	1.45		.00	.00	.32	.32
		2 356.21	.00	.00	.50		.00	.00	.22	.22
		3 500.71	.00	.00	.00		.00	.00	.00	.00
		44 379.94	.00	.00	.50		.00	.00	.11	.11
100.5	31.58	1 493.52	.00	.00	1.45		.00	.00	.32	.32
		2 356.11	.00	.00	.49		.00	.00	.22	.22
		3 500.51	.00	.00	.00		.00	.00	.00	.00
		44 379.82	.00	.00	.49		.00	.00	.11	.11

TABLE G-III. - PRINTED OUTPUT FROM SAMPLE CASES - Continued

PHI	TIME	TEMPERATURE	INCIDENT (BTU/HR-FT ²)				ABSORBED (BTU/HR)			
			QSOLAR	QALBEDO	QPLANET	QPLANET	QSOLAR	QALBEDO	QPLANET	QTOTAL
100.0	51.61	1 493.48	.00	.00	1.45	.00	.00	.00	.32	.32
		2 356.09	.00	.00	.49	.00	.00	.00	.22	.22
		3 500.49	39.89	.00	.00	8.78	.00	.00	.00	8.78
		44 379.80	16.87	.00	.49	.00	3.71	.00	.11	3.82
110.0	54.94	1 490.19	.00	.00	1.41	.00	.00	.00	.31	.31
		2 354.31	.00	.00	.47	.00	.00	.00	.21	.21
		3 500.61	111.23	.00	.02	24.47	.00	.00	.00	24.46
		44 378.60	16.87	.00	.47	.00	3.71	.00	.10	3.81
120.0	58.58	1 486.68	.00	.00	1.38	.00	.00	.00	.30	.30
		2 352.39	.00	.00	.44	.00	.00	.00	.20	.20
		3 503.35	183.96	.00	.06	40.47	.00	.00	.01	40.49
		44 377.31	16.87	.00	.44	.00	3.71	.00	.10	3.81
130.0	42.31	1 483.84	.00	.00	1.40	.00	.00	.00	.32	.32
		2 351.07	.00	.00	.03	.00	.00	.00	.00	.00
		3 508.55	231.09	.00	.06	59.24	.00	.00	.02	59.26
		44 376.67	16.87	.43	.07	.00	3.71	.09	1.91	5.72
140.0	46.11	1 482.57	.00	2.83	37.44	.00	.00	.62	12.64	13.26
		2 350.71	.00	1.00	20.26	.00	.00	.44	8.91	9.35
		3 515.87	310.38	.00	.07	68.33	.00	.00	.02	68.34
		44 377.00	16.87	1.04	21.08	.00	3.71	.23	4.64	8.58
150.0	49.99	1 482.99	.00	4.88	38.99	.00	.00	1.07	21.78	22.85
		2 351.30	.00	1.58	32.08	.00	.00	.70	14.10	14.80
		3 524.98	380.84	.00	.03	79.34	.00	.00	.01	79.35
		44 378.29	16.87	1.62	32.93	.00	3.71	.36	7.23	11.32
160.0	53.92	1 484.99	.00	6.81	138.12	.00	.00	1.90	30.39	31.89
		2 352.74	.00	2.10	48.60	.00	.00	.92	18.74	19.67
		3 535.38	598.74	.00	.03	87.34	.00	.00	.01	87.35
		44 380.42	16.87	2.15	43.91	.00	3.71	.47	9.57	13.76
170.0	57.89	1 488.58	.00	8.53	173.04	.00	.00	1.88	38.07	39.95
		2 354.92	.00	2.35	51.72	.00	.00	1.12	22.76	23.88
		3 546.90	426.89	.00	.00	93.67	.00	.00	.00	93.67
		44 383.23	16.87	2.60	52.63	.00	3.71	.57	11.98	15.66
180.0	61.88	1 492.94	.00	10.04	203.32	.00	.00	2.21	44.77	46.98
		2 357.73	.00	2.95	59.89	.00	.00	1.30	26.35	27.65
		3 557.73	440.68	.00	.00	96.95	.00	.00	.00	96.95
		44 386.69	16.87	3.00	60.80	.00	3.71	.66	13.38	17.75
190.0	65.87	1 498.40	.00	11.28	228.78	.00	.00	2.48	50.33	52.81
		2 361.04	.00	3.29	66.70	.00	.00	1.45	29.35	30.79
		3 568.47	441.28	.00	.00	97.08	.00	.00	.00	97.08
		44 390.58	16.87	3.33	67.61	.00	3.71	.73	14.87	19.32
200.0	69.84	1 504.39	.00	12.18	247.06	.00	.00	2.68	54.35	57.03
		2 364.71	.00	3.57	72.30	.00	.00	1.57	31.81	33.38
		3 578.12	428.47	.00	.03	94.26	.00	.00	.01	94.27
		44 394.78	16.87	3.61	73.20	.00	3.71	.79	16.10	20.61

TABLE G-III. - PRINTED OUTPUT FROM SAMPLE CASES - Continued

PHI	TIME	TEMPERATURE	INCIDENT (BTU/HR-FT ²)			ABSORBED (BTU/HR)			
			QSOLAR	QALBEDO	QPLANET	QSOLAR	QALBEDO	QPLANET	QTOTAL
210.0	73.77	1 510.67	.00	12.85	260.46	.00	2.83	57.30	60.13
		2 368.63	.00	3.80	77.11	.00	1.67	33.93	35.60
		3 586.20	402.64	.00	.08	98.39	.00	.02	98.60
		44 339.20	16.87	3.85	78.02	3.71	.85	17.16	21.72
220.0	77.65	1 516.35	.00	13.22	267.38	.00	2.91	58.35	61.86
		2 372.70	.00	3.88	80.75	.00	1.75	35.53	37.28
		3 592.31	364.58	.01	.15	90.21	.00	.03	90.24
		44 403.70	16.87	4.03	81.66	3.71	.89	17.36	22.56
230.0	81.46	1 522.87	.00	13.20	267.48	.00	2.90	58.89	61.79
		2 376.71	.00	4.06	82.41	.00	1.79	36.26	38.05
		3 596.17	313.44	.01	.21	69.40	.00	.05	69.45
		44 408.10	16.87	4.11	83.28	3.71	.90	18.32	22.94
240.0	85.18	1 528.24	.00	12.31	261.71	.00	2.84	57.38	60.42
		2 380.34	.00	4.06	82.42	.00	1.79	36.26	38.05
		3 597.63	456.72	.01	.26	96.48	.00	.06	96.54
		44 412.26	16.87	4.11	83.28	3.71	.90	18.32	22.94
250.0	89.82	1 532.66	.00	12.08	264.38	.00	2.66	53.30	56.55
		2 383.90	.00	3.89	78.85	.00	1.71	34.69	36.41
		3 598.64	130.20	.01	.29	41.84	.00	.06	41.91
		44 413.31	16.87	3.93	79.69	3.71	.86	17.53	22.11
260.0	92.37	1 535.87	.00	10.83	212.50	.00	2.38	48.23	50.67
		2 386.36	.00	3.39	72.00	.00	1.36	31.66	33.24
		3 593.23	117.89	.01	.27	23.94	.00	.06	26.00
		44 418.64	16.87	3.39	72.62	3.71	.79	16.02	20.52
270.0	95.82	1 537.68	.00	9.19	189.31	.00	2.01	40.81	42.82
		2 389.40	.00	3.09	61.66	.00	1.34	27.22	28.56
		3 597.52	42.01	.01	.28	9.24	.00	.06	9.30
		44 420.87	16.87	3.09	62.66	3.71	.68	13.79	18.18
280.0	99.18	1 539.12	35.13	7.32	144.42	7.73	1.97	31.77	41.07
		2 389.30	.00	2.48	50.19	.00	1.09	22.09	23.17
		3 590.89	.00	.03	.80	.00	.01	.13	.14
		44 421.36	16.87	2.31	50.32	3.71	.59	11.20	13.47
290.0	102.46	1 541.28	111.25	4.70	95.39	29.47	1.03	20.39	46.30
		2 389.03	.00	1.83	33.31	.00	.73	14.75	15.47
		3 574.82	.00	.02	.47	.00	.01	.10	.11
		44 421.89	16.87	1.88	34.16	3.71	.57	7.31	11.60
300.0	105.88	1 543.81	183.36	2.07	42.04	40.47	.46	9.23	30.18
		2 387.40	.00	.83	12.79	.00	.28	5.63	5.91
		3 568.77	.00	.00	.03	.00	.00	.01	.01
		44 420.38	16.87	.66	13.32	3.71	.14	2.93	6.79
310.0	108.79	1 546.97	251.09	.00	1.87	59.24	.00	.37	35.61
		2 385.09	.00	.00	.71	.00	.00	.31	.31
		3 563.27	.00	.00	.03	.00	.00	.01	.01
		44 418.16	16.87	.00	.71	3.71	.00	.16	3.87

TABLE G-III. - PRINTED OUTPUT FROM SAMPLE CASES - Continued

				INCIDENT (BTU/HR-FT**2)			ABSORBED (BTU/HR)			
PHI	TIME	TEMPERATURE		QSOLAR	QALBEDO	QPLANET	QSOLAR	QALBEDO	QPLANET	QTOTAL
310.0	7.58	1	533.68	443.00	184.71	99.02	97.46	40.64	21.79	159.88
		2	389.76	443.00	184.71	194.92	81.27	43.57	319.76	
340.0	15.17	1	536.13	443.00	106.64	99.02	97.46	23.46	21.79	142.71
		2	402.95	443.00	106.64	194.92	46.92	43.57	285.41	
10.0	22.75	1	543.10	443.00	.00	99.02	97.46	.00	21.79	119.25
		2	413.89	443.00	.00	99.02	194.92	.00	43.57	238.49
28.2	27.35	1	546.57	443.00	.00	99.02	97.46	.00	21.79	119.25
		2	419.84	443.00	.00	99.02	194.92	.00	43.57	238.49
28.3	27.37	1	546.56	.00	.00	99.02	.00	.00	21.79	21.79
		2	419.84	.00	.00	99.02	.00	.00	43.57	43.57
40.0	30.33	1	545.17	.00	.00	99.02	.00	.00	21.79	21.79
		2	420.34	.00	.00	99.02	.00	.00	43.57	43.57
70.0	37.92	1	535.95	.00	.00	99.02	.00	.00	21.79	21.79
		2	421.56	.00	.00	99.02	.00	.00	43.57	43.57
100.0	45.50	1	527.89	.00	.00	99.02	.00	.00	21.79	21.79
		2	422.78	.00	.00	99.02	.00	.00	43.57	43.57
130.0	53.08	1	526.61	.00	.00	99.02	.00	.00	21.79	21.79
		2	423.93	.00	.00	99.02	.00	.00	43.57	43.57
160.0	60.67	1	524.83	.00	.00	99.02	.00	.00	21.79	21.79
		2	425.03	.00	.00	99.02	.00	.00	43.57	43.57
171.7	63.63	1	521.76	.00	.00	99.02	.00	.00	21.79	21.79
		2	425.45	.00	.00	99.02	.00	.00	43.57	43.57
171.8	63.65	1	521.76	443.00	.00	99.02	97.46	.00	21.79	119.25
		2	425.48	443.00	.00	99.02	194.92	.00	43.57	238.49
190.0	68.25	1	522.14	443.00	.00	99.02	97.46	.00	21.79	119.25
		2	431.16	443.00	.00	99.02	194.92	.00	43.57	238.49
220.0	75.83	1	524.69	443.00	106.64	99.02	97.46	23.46	21.79	142.71
		2	441.42	443.00	106.64	99.02	194.92	46.92	43.57	285.41
250.0	83.42	1	534.61	443.00	184.71	99.02	97.46	40.64	21.79	159.88
		2	432.93	443.00	184.71	99.02	194.92	81.27	43.57	319.76
280.0	91.00	1	543.77	443.00	213.28	99.02	97.46	46.92	21.79	166.17
		2	464.81	443.00	213.28	99.02	194.92	93.84	43.57	332.33
THE OUTPUT AT TIME= 91.00 ENDS ORBIT NUMBER 1										
THE TEMPERATURES OF THESE NODES STABILIZED DURING THE LAST ORBIT...										
310.0	98.58	1	546.37	443.00	184.71	99.02	97.46	40.64	21.79	159.88
		2								

TABLE G-III. - PRINTED OUTPUT FROM SAMPLE CASES - Continued

CALCULATION TIME= .07 , PLOT TIME= .06 , TOTAL TIME FOR THIS CASE= .15...ALL TIMES ARE IN MINUTES...

CASE NO. 502
CASE TO STUDY EFFECT OF SWITCHING FREQUENCY OF INTERNAL HEAT LOADS
TEST CASE 502
PLANET EARTH
MAX. # ORBIT ALT. (NM) # MIN. PHIO DPH1 PLANET TEMPERATURE IS CONSTANT PHOUT
181.00 181.00 279.99999 6.00000 279.99999 90.00000 29.19306 171.80689
ALPHA= 90.00000 GAMMA= 169.99999
NEW DUTY CYCLES READ IN
INDEX 2
81N= .0000 T= 19.00 81N= 20.0000 T= 30.00 81N= .0000 T= 45.00 81N= 20.0000 T= 60.00
81N= .0000 T= 75.00 81N= 20.0000 T= 90.00 81N= .0000 T= 105.00 81N= 20.0000 T= 120.00
81N= .0000 T=
INDEX 3
81N= 10.0000 T=90000.00 81N= 10.0000 T=
ELEMENT CONTING SUBSTRATE DUTY CYCLE LAMDA OMEGA T(G) THICKNESS AREA NODE NO.
1 2 3
2
.01
MAXIMUM NO. OF ORBITS REQUESTED= 7.250

TABLE G-III. - PRINTED OUTPUT FROM SAMPLE CASES - Continued

INCIDENT (BTU/HR-FT**2)				ABSORBED (BTU/HR)						
PHI	TIME	TEMPERATURE	Q SOLAR	Q ALBEDO	Q PLANET	Q SOLAR	Q ALBEDO	Q PLANET	Q TOTAL	
310.0	7.58	1	533.68	443.00	184.71	99.02	97.46	40.64	21.79	159.88
		2	389.76	443.00	184.71	99.02	194.32	81.27	43.57	319.76
340.0	15.17	1	536.13	443.00	106.64	99.02	97.46	23.46	21.79	142.71
		2	402.95	443.00	106.64	99.02	194.32	46.92	43.57	285.41
10.0	22.75	1	543.10	443.00	.00	99.02	97.46	.00	21.79	119.25
		2	413.89	443.00	.00	99.02	194.32	.00	43.57	238.49
28.2	27.35	1	546.57	443.00	.00	99.02	97.46	.00	21.79	119.25
		2	419.84	443.00	.00	99.02	194.32	.00	43.57	238.49
28.3	27.37	1	546.56	.00	.00	99.02	.00	.00	21.79	21.79
		2	419.84	.00	.00	99.02	.00	.00	43.57	43.57
40.0	30.33	1	545.17	.00	.00	99.02	.00	.00	21.79	21.79
		2	420.34	.00	.00	99.02	.00	.00	43.57	43.57
70.0	37.92	1	535.95	.00	.00	99.02	.00	.00	21.79	21.79
		2	421.58	.00	.00	99.02	.00	.00	43.57	43.57
100.0	45.50	1	527.89	.00	.00	99.02	.00	.00	21.79	21.79
		2	422.78	.00	.00	99.02	.00	.00	43.57	43.57
130.0	53.08	1	526.61	.00	.00	99.02	.00	.00	21.79	21.79
		2	423.93	.00	.00	99.02	.00	.00	43.57	43.57
160.0	60.67	1	524.83	.00	.00	99.02	.00	.00	21.79	21.79
		2	425.03	.00	.00	99.02	.00	.00	43.57	43.57
171.7	63.63	1	521.76	.00	.00	99.02	.00	.00	21.79	21.79
		2	425.45	.00	.00	99.02	.00	.00	43.57	43.57
171.8	63.65	1	521.76	443.00	.00	99.02	97.46	.00	21.79	119.25
		2	425.48	443.00	.00	99.02	194.32	.00	43.57	238.49
190.0	68.25	1	522.14	443.00	.00	99.02	97.46	.00	21.79	119.25
		2	431.16	443.00	.00	99.02	194.32	.00	43.57	238.49
220.0	75.83	1	524.69	443.00	106.64	99.02	97.46	23.46	21.79	142.71
		2	441.42	443.00	106.64	99.02	194.32	46.92	43.57	285.41
250.0	83.42	1	534.61	443.00	184.71	99.02	97.46	40.64	21.79	159.88
		2	452.93	443.00	184.71	99.02	194.32	81.27	43.57	319.76
280.0	91.00	1	543.77	443.00	213.28	99.02	97.46	46.92	21.79	166.17
		2	464.81	443.00	213.28	99.02	194.32	93.84	43.57	332.33
THE OUTPUT AT TIME= 91.00 ENDS ORBIT NUMBER 1										
THE TEMPERATURES OF THESE NODES STABILIZED DURING THE LAST ORBIT...										
310.0	98.98	1	546.37	443.00	184.71	99.02	97.46	40.64	21.79	159.88

TABLE G-III. - PRINTED OUTPUT FROM SAMPLE CASES - Continued

			INCIDENT (BTU/HR-FT**2)				ABSORBED (BTU/HR)			
PHI	TIME	TEMPERATURE	QSOLAR	QALBEDO	QPLANET		QSOLAR	QALBEDO	QPLANET	QTOTAL
		2 475.97	443.00	184.71	99.02		194.92	81.27	43.57	319.76
340.0	106.17	1 547.82	443.00	106.64	99.02		97.46	23.46	21.79	142.71
		2 489.43	443.00	106.64	99.02		194.92	46.92	43.57	285.41
10.0	113.75	1 553.84	443.00	.00	99.02		97.46	.00	21.79	119.25
		2 492.54	443.00	.00	99.02		194.92	.00	43.57	238.49
28.2	118.35	1 556.76	443.00	.00	99.02		97.46	.00	21.79	119.25
		2 496.16	443.00	.00	99.02		194.92	.00	43.57	238.49
28.3	118.38	1 556.75	.00	.00	99.02		.00	.00	21.79	21.79
		2 496.15	.00	.00	99.02		.00	.00	43.57	43.57
40.0	121.33	1 555.02	.00	.00	99.02		.00	.00	21.79	21.79
		2 495.17	.00	.00	99.02		.00	.00	43.57	43.57
70.0	128.92	1 545.00	.00	.00	99.02		.00	.00	21.79	21.79
		2 492.75	.00	.00	99.02		.00	.00	43.57	43.57
100.0	136.50	1 536.24	.00	.00	99.02		.00	.00	21.79	21.79
		2 490.48	.00	.00	99.02		.00	.00	43.57	43.57
130.0	144.08	1 534.34	.00	.00	99.02		.00	.00	21.79	21.79
		2 486.35	.00	.00	99.02		.00	.00	43.57	43.57
160.0	151.67	1 531.98	.00	.00	99.02		.00	.00	21.79	21.79
		2 486.34	.00	.00	99.02		.00	.00	43.57	43.57
171.7	154.63	1 528.70	.00	.00	99.02		.00	.00	21.79	21.79
		2 485.59	.00	.00	99.02		.00	.00	43.57	43.57
171.8	154.65	1 528.70	443.00	.00	99.02		97.46	.00	21.79	119.25
		2 485.61	443.00	.00	99.02		194.92	.00	43.57	238.49
190.0	159.25	1 528.77	443.00	.00	99.02		97.46	.00	21.79	119.25
		2 489.46	443.00	.00	99.02		194.92	.00	43.57	238.49
220.0	166.84	1 530.84	443.00	106.64	99.02		97.46	23.46	21.79	142.71
		2 496.77	443.00	106.64	99.02		194.92	46.92	43.57	285.41
250.0	174.42	1 540.29	443.00	184.71	99.02		97.46	40.64	21.79	159.88
		2 505.28	443.00	184.71	99.02		194.92	81.27	43.57	319.76
280.0	182.00	1 549.00	443.00	213.28	99.02		97.46	46.92	21.79	166.17
		2 514.15	443.00	213.28	99.02		194.92	93.84	43.57	332.33
THE OUTPUT AT TIME= 182.00 ENDS ORBIT NUMBER 2										
THE TEMPERATURES OF THESE NODES STABILIZED DURING THE LAST ORBIT...										
310.0	189.59	1 551.17	443.00	184.71	99.02		97.46	40.64	21.79	159.88
		2 522.30	443.00	184.71	99.02		194.92	81.27	43.57	319.76
340.0	197.17	1 552.23	443.00	106.64	99.02		97.46	23.46	21.79	142.71

TABLE G-III. - PRINTED OUTPUT FROM SAMPLE CASES - Continued

PHI	TIME	TEMPERATURE	INCIDENT (BTU/HR-FT ²)			ABSORBED (BTU/HR)			
			QSOLAR	QALBEDO	QPLANET	QSOLAR	QALBEDO	QPLANET	GTOTAL
10.0	204.75	1 557.87	443.00	.00	99.02	194.92	46.32	43.57	285.41
		2 532.99	443.00	.00	99.02	97.46	.00	21.79	119.25
28.2	209.35	1 560.58	443.00	.00	99.02	194.92	.00	43.57	238.49
		2 534.90	443.00	.00	99.02	97.46	.00	21.79	119.25
28.3	209.38	1 560.56	.00	.00	99.02	194.92	.00	43.57	238.49
		2 534.89	.00	.00	99.02	.00	.00	21.79	21.79
40.0	212.34	1 559.70	.00	.00	99.02	.00	.00	43.57	43.57
		2 532.84	.00	.00	99.02	.00	.00	21.79	21.79
70.0	219.92	1 548.37	.00	.00	99.02	.00	.00	43.57	43.57
		2 527.88	.00	.00	99.02	.00	.00	21.79	21.79
100.0	227.50	1 539.35	.00	.00	99.02	.00	.00	43.57	43.57
		2 523.29	.00	.00	99.02	.00	.00	21.79	21.79
130.0	235.09	1 537.21	.00	.00	99.02	.00	.00	43.57	43.57
		2 519.02	.00	.00	99.02	.00	.00	21.79	21.79
160.0	242.67	1 534.63	.00	.00	99.02	.00	.00	43.57	43.57
		2 515.06	.00	.00	99.02	.00	.00	21.79	21.79
171.7	245.63	1 531.27	.00	.00	99.02	.00	.00	43.57	43.57
		2 513.58	.00	.00	99.02	.00	.00	21.79	21.79
171.8	245.65	1 531.27	443.00	.00	99.02	97.46	.00	21.79	119.25
		2 513.60	443.00	.00	99.02	194.92	.00	43.57	238.49
190.0	250.25	1 531.22	443.00	.00	99.02	97.46	.00	21.79	119.25
		2 516.38	443.00	.00	99.02	194.92	.00	43.57	238.49
220.0	257.84	1 533.11	443.00	106.64	99.02	97.46	23.46	21.79	142.71
		2 521.91	443.00	106.64	99.02	194.92	46.32	43.57	285.41
250.0	265.42	1 542.39	443.00	184.71	99.02	97.46	40.64	21.79	159.88
		2 528.71	443.00	184.71	99.02	194.92	81.27	43.57	319.76
280.0	273.00	1 550.92	443.00	213.28	99.02	97.46	46.32	21.79	166.17
		2 535.91	443.00	213.28	99.02	194.92	93.84	43.57	332.33
THE OUTPUT AT TIME= 273.00 ENDS ORBIT NUMBER 3									
THE TEMPERATURES OF THESE NODES STABILIZED DURING THE LAST ORBIT...									
310.0	280.59	1 552.93	443.00	184.71	99.02	97.46	40.64	21.79	159.88
		2 542.45	443.00	184.71	99.02	194.92	81.27	43.57	319.76
340.0	288.17	1 553.84	443.00	106.64	99.02	97.46	23.46	21.79	142.71
		2 547.38	443.00	106.64	99.02	194.92	46.32	43.57	285.41
10.0	299.79	1 559.35	443.00	.00	99.02	97.46	.00	21.79	119.25

TABLE G-III. - PRINTED OUTPUT FROM SAMPLE CASES - Continued

PHI	TIME	TEMPERATURE	INCIDENT (BTU/HR-FT**2)				ABSORBED (BTU/HR)			
			QSOLAR	QALBEDO	QPLANET		QSOLAR	QALBEDO	QPLANET	GTOTAL
28.2	300.35	1	561.97	443.00	99.02		194.92	.00	43.57	238.43
		2	551.20	443.00	99.02		194.92	.00	43.57	238.49
28.3	300.38	1	561.96	.00	99.02		.00	.00	21.79	21.79
		2	551.18	.00	99.02		.00	.00	43.57	43.57
40.0	303.34	1	560.05	.00	99.02		.00	.00	21.79	21.79
		2	548.62	.00	99.02		.00	.00	43.57	43.57
70.0	310.92	1	549.61	.00	99.02		.00	.00	21.79	21.79
		2	542.43	.00	99.02		.00	.00	43.57	43.57
100.0	318.50	1	540.48	.00	99.02		.00	.00	21.79	21.79
		2	536.74	.00	99.02		.00	.00	43.57	43.57
130.0	326.09	1	538.25	.00	99.02		.00	.00	21.79	21.79
		2	531.48	.00	99.02		.00	.00	43.57	43.57
160.0	333.67	1	535.59	.00	99.02		.00	.00	21.79	21.79
		2	526.82	.00	99.02		.00	.00	43.57	43.57
171.7	336.63	1	532.20	.00	99.02		.00	.00	21.79	21.79
		2	524.82	.00	99.02		.00	.00	43.57	43.57
171.8	336.66	1	532.20	443.00	99.02		97.46	.00	21.79	119.25
		2	524.84	443.00	99.02		194.92	.00	43.57	238.49
190.0	341.25	1	532.11	443.00	99.02		97.46	.00	21.79	119.25
		2	527.13	443.00	99.02		194.92	.00	43.57	238.49
220.0	348.84	1	533.93	443.00	99.02		97.46	23.46	21.79	142.71
		2	531.68	443.00	99.02		194.92	46.92	43.57	285.41
250.0	356.42	1	543.15	443.00	99.02		97.46	40.64	21.79	159.88
		2	537.94	443.00	99.02		194.92	81.27	43.57	319.76
280.0	364.00	1	551.62	443.00	99.02		97.46	46.92	21.79	166.17
		2	544.43	443.00	99.02		194.92	93.84	43.57	332.33
THE OUTPUT AT TIME= 364.00 ENDS ORBIT NUMBER 4										
THE TEMPERATURES OF THESE NODES STABILIZED DURING THE LAST ORBIT...										
310.0	371.59	1	553.57	443.00	99.02		97.46	40.64	21.79	159.88
		2	550.28	443.00	99.02		194.92	81.27	43.57	319.76
340.0	379.17	1	554.43	443.00	99.02		97.46	23.46	21.79	142.71
		2	554.57	443.00	99.02		194.92	46.92	43.57	285.41
10.0	386.75	1	559.89	443.00	99.02		97.46	.00	21.79	119.25
		2	556.71	443.00	99.02		194.92	.00	43.57	238.49
28.2	391.35	1	562.48	443.00	99.02		97.46	.00	21.79	119.25

TABLE G-III. - PRINTED OUTPUT FROM SAMPLE CASES - Continued

PHI	TIME	TEMPERATURE	INCIDENT (BTU/HR-FT ²)			ABSORBED (BTU/HR)		
			QSOLAR	QALBEDO	QPLANET	QSOLAR	QALBEDO	QPLANET
28.3	331.38	1 562.47	443.00	.00	99.02	194.92	.00	43.57
		2 557.42	.00	.00	99.02	.00	.00	21.79
			.00	.00	99.02	.00	.00	43.57
40.0	334.34	1 560.54	.00	.00	99.02	.00	.00	21.79
		2 554.65	.00	.00	99.02	.00	.00	43.57
			.00	.00	99.02	.00	.00	21.79
70.0	401.92	1 550.05	.00	.00	99.02	.00	.00	43.57
		2 547.96	.00	.00	99.02	.00	.00	21.79
			.00	.00	99.02	.00	.00	43.57
100.0	409.50	1 540.89	.00	.00	99.02	.00	.00	21.79
		2 541.83	.00	.00	99.02	.00	.00	43.57
			.00	.00	99.02	.00	.00	21.79
130.0	417.09	1 538.63	.00	.00	99.02	.00	.00	43.57
		2 536.18	.00	.00	99.02	.00	.00	21.79
			.00	.00	99.02	.00	.00	43.57
160.0	424.67	1 535.94	.00	.00	99.02	.00	.00	21.79
		2 530.97	.00	.00	99.02	.00	.00	43.57
			.00	.00	99.02	.00	.00	21.79
171.7	427.63	1 532.54	.00	.00	99.02	.00	.00	43.57
		2 529.04	.00	.00	99.02	.00	.00	21.79
			.00	.00	99.02	.00	.00	43.57
171.8	427.66	1 532.54	443.00	.00	99.02	97.46	.00	21.79
		2 529.05	443.00	.00	99.02	194.92	.00	43.57
			.00	.00	99.02	.00	.00	21.79
190.0	432.23	1 532.44	443.00	.00	99.02	97.46	.00	21.79
		2 531.15	443.00	.00	99.02	194.92	.00	43.57
			.00	.00	99.02	.00	.00	21.79
220.0	439.84	1 534.23	443.00	106.64	99.02	97.46	23.46	21.79
		2 535.60	443.00	106.64	99.02	194.92	46.92	43.57
			.00	.00	99.02	.00	.00	21.79
250.0	447.42	1 543.42	443.00	184.71	99.02	97.46	40.64	21.79
		2 541.38	443.00	184.71	99.02	194.92	81.27	43.57
			.00	.00	99.02	.00	.00	21.79
280.0	455.01	1 551.87	443.00	213.28	99.02	97.46	46.92	21.79
		2 547.59	443.00	213.28	99.02	194.92	93.84	43.57
			.00	.00	99.02	.00	.00	21.79
THE OUTPUT AT TIME= 455.01 ENDS ORBIT NUMBER 5								
THE TEMPERATURES OF THESE MODES STABILIZED DURING THE LAST ORBIT...								
310.0	462.59	1 553.57	443.00	184.71	99.02	97.46	40.64	21.79
		2 553.18	443.00	184.71	99.02	194.92	81.27	43.57
			.00	.00	99.02	.00	.00	21.79
340.0	470.17	1 554.43	443.00	106.64	99.02	97.46	23.46	21.79
		2 557.22	443.00	106.64	99.02	194.92	46.92	43.57
			.00	.00	99.02	.00	.00	21.79
10.0	477.76	1 559.89	443.00	.00	99.02	97.46	.00	21.79
		2 559.14	443.00	.00	99.02	194.92	.00	43.57
			.00	.00	99.02	.00	.00	21.79
28.2	482.33	1 562.48	443.00	.00	99.02	97.46	.00	21.79
		2 559.74	443.00	.00	99.02	194.92	.00	43.57
			.00	.00	99.02	.00	.00	21.79
28.3	482.38	1 562.47	.00	.00	99.02	.00	.00	21.79

TABLE G-III.- PRINTED OUTPUT FROM SAMPLE CASES - Continued

		INCIDENT (BTU/HR-FT ²)				ABSORBED (BTU/HR)			
PHI	TIME	TEMPERATURE	QSOLAR	QALBEDO	QPLANET	QSOLAR	QALBEDO	QPLANET	QTOTAL
		2 559.72	.00	.00	99.02	.00	.00	43.57	43.57
40.0	485.34	1 560.54	.00	.00	99.02	.00	.00	21.79	21.79
		2 556.86	.00	.00	99.02	.00	.00	43.57	43.57
70.0	492.92	1 550.05	.00	.00	99.02	.00	.00	21.79	21.79
		2 549.99	.00	.00	99.02	.00	.00	43.57	43.57
100.0	500.51	1 540.89	.00	.00	99.02	.00	.00	21.79	21.79
		2 543.69	.00	.00	99.02	.00	.00	43.57	43.57
130.0	508.09	1 538.63	.00	.00	99.02	.00	.00	21.79	21.79
		2 537.90	.00	.00	99.02	.00	.00	43.57	43.57
160.0	515.67	1 535.94	.00	.00	99.02	.00	.00	21.79	21.79
		2 532.56	.00	.00	99.02	.00	.00	43.57	43.57
171.7	518.63	1 532.54	.00	.00	99.02	.00	.00	21.79	21.79
		2 530.58	.00	.00	99.02	.00	.00	43.57	43.57
171.8	518.66	1 532.54	443.00	.00	99.02	97.46	.00	21.79	119.25
		2 530.59	443.00	.00	99.02	194.92	.00	43.57	238.49
190.0	523.26	1 532.44	443.00	.00	99.02	97.46	.00	21.79	119.25
		2 532.62	443.00	.00	99.02	194.92	.00	43.57	238.49
220.0	530.84	1 534.23	443.00	106.64	99.02	97.46	23.46	21.79	142.71
		2 536.96	443.00	106.64	99.02	194.92	46.92	43.57	285.41
250.0	538.42	1 543.42	443.00	184.71	99.02	97.46	40.64	21.79	159.88
		2 542.63	443.00	184.71	99.02	194.92	81.27	43.57	319.76
280.0	546.01	1 551.87	443.00	213.28	99.02	97.46	46.92	21.79	166.17
		2 548.74	443.00	213.28	99.02	194.92	93.84	43.57	332.33
THE OUTPUT AT TIME= 546.01 ENDS ORBIT NUMBER 6									
THE TEMPERATURES OF THESE NODES STABILIZED DURING THE LAST ORBIT...									
310.0	553.59	1 553.57	443.00	184.71	99.02	97.46	40.64	21.79	159.88
		2 554.24	443.00	184.71	99.02	194.92	81.27	43.57	319.76
340.0	561.17	1 554.43	443.00	106.64	99.02	97.46	23.46	21.79	142.71
		2 558.18	443.00	106.64	99.02	194.92	46.92	43.57	285.41
10.0	568.76	1 559.89	443.00	.00	99.02	97.46	.00	21.79	119.25
		2 560.02	443.00	.00	99.02	194.92	.00	43.57	238.49
28.2	573.36	1 562.48	443.00	.00	99.02	97.46	.00	21.79	119.25
		2 560.57	443.00	.00	99.02	194.92	.00	43.57	238.49
28.3	573.38	1 562.47	.00	.00	99.02	.00	.00	21.79	21.79
		2 560.55	.00	.00	99.02	.00	.00	43.57	43.57
40.0	576.34	1 560.54	.00	.00	99.02	.00	.00	21.79	21.79

TABLE G-III. - PRINTED OUTPUT FROM SAMPLE CASES - Continued

PHI	TIME	TEMPERATURE	INCIDENT (BTU/HR-FT**2)			ABSORBED (BTU/HR)			
			QSOLAR	GALBEDO	GPLANET	QSOLAR	GALBEDO	GPLANET	GTOTAL
70.0	583.32	1	557.67	.00	99.02	.00	.00	43.57	43.57
		2	550.05	.00	99.02	.00	.00	21.79	21.79
100.0	591.51	1	550.73	.00	99.02	.00	.00	43.57	43.57
		2	540.89	.00	99.02	.00	.00	21.79	21.79
130.0	599.09	1	544.37	.00	99.02	.00	.00	43.57	43.57
		2	538.63	.00	99.02	.00	.00	21.79	21.79
160.0	606.67	1	538.52	.00	99.02	.00	.00	43.57	43.57
		2	535.94	.00	99.02	.00	.00	21.79	21.79
171.7	609.63	1	533.13	.00	99.02	.00	.00	43.57	43.57
		2	532.54	.00	99.02	.00	.00	21.79	21.79
171.8	609.66	1	531.14	.00	99.02	.00	.00	43.57	43.57
		2	532.54	.00	99.02	.00	.00	21.79	21.79
190.0	614.26	1	531.15	.00	99.02	.00	.00	43.57	43.57
		2	532.44	.00	99.02	.00	.00	21.79	21.79
220.0	621.84	1	533.15	.00	99.02	.00	.00	43.57	43.57
		2	534.23	.00	99.02	.00	.00	21.79	21.79
250.0	629.42	1	537.45	.00	99.02	.00	.00	43.57	43.57
		2	543.42	.00	99.02	.00	.00	21.79	21.79
280.0	637.01	1	543.08	.00	99.02	.00	.00	43.57	43.57
		2	551.87	.00	99.02	.00	.00	21.79	21.79
THE OUTPUT AT TIME= 637.01 ENDS ORBIT NUMBER 7			549.16	213.28	99.02	97.46	46.92	21.79	166.17
THE TEMPERATURES OF THESE NODES STABILIZED DURING THE LAST ORBIT...			443.00	213.28	99.02	194.92	93.84	43.57	332.35

S-C 4020 PLOTS HAVE BEEN REQUESTED AND SHALL BE PROVIDED BY LINK 2

CALCULATION TIME= .67 , PLOT TIME= .05 , TOTAL TIME FOR THIS CASE= .72...ALL TIMES ARE IN MINUTES...

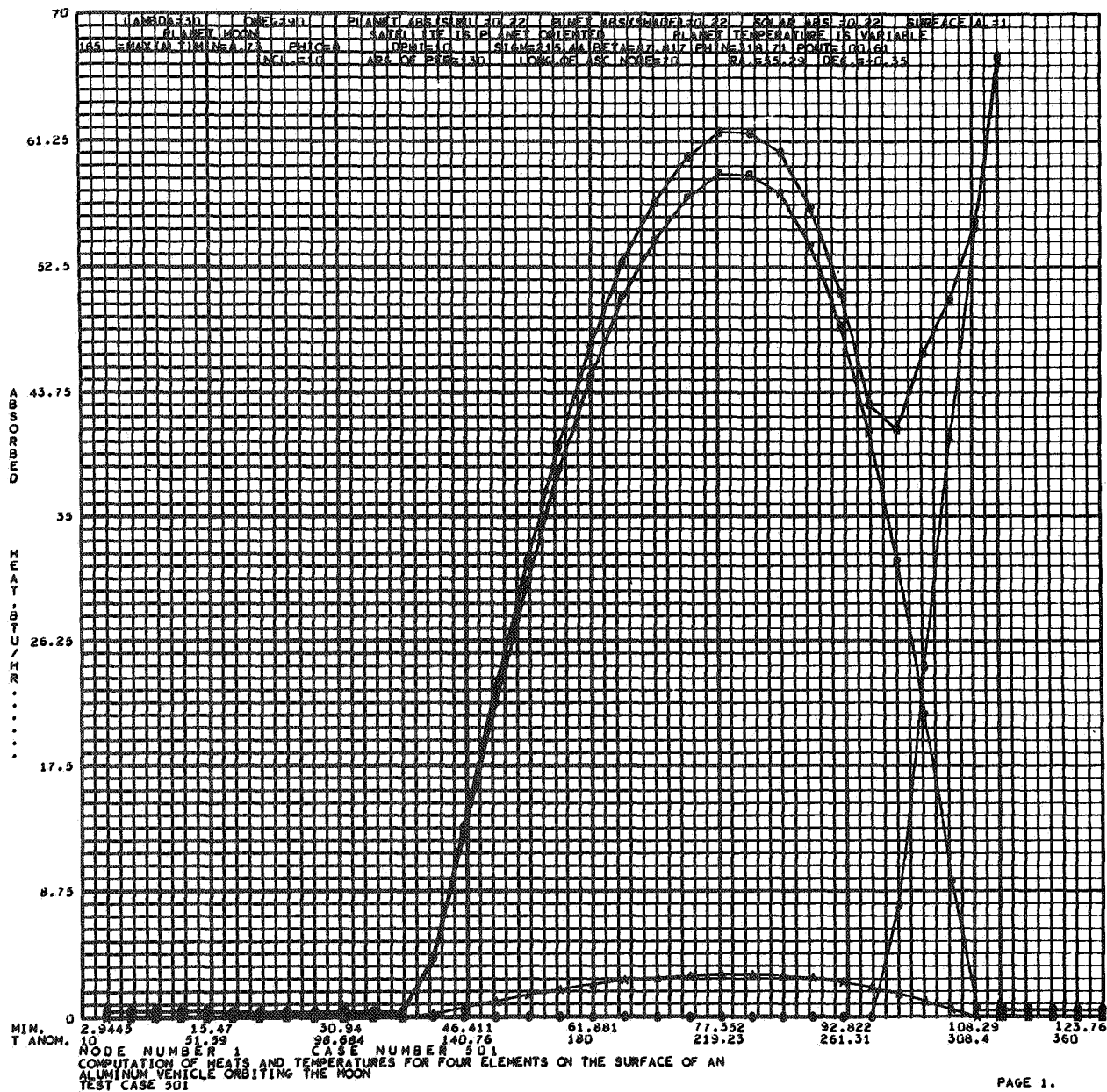
CASE NO.	SUN	PLANET EARTH	SATELLITE IS SUN ORIENTED	PLANET TEMPERATURE IS CONSTANT	PROOUT			
181.00	181.00	181.00	181.00	279.99999	26.19307			
MAX. & ORBIT ALT. (MM) & MIN. P.M.O.	181.00	180.00000	2.50000	90.00000	171.90469			
ALPHA= 80.00000	GAMMA= 160.99999							
ELEMENT	COATING	SUBSTRATE	DUTY CYCLE	OMEGA	T (D)	THICKNESS	AREA	MODE NO.
1	2	1	1					
2	1	1	0.					
2	3	1	1	90.				
3	2	1	180.					
4	3	1	180.	90.		.01		

MAXIMUM NO. OF ORBITS REQUESTED= 500

TABLE G-III. - PRINTED OUTPUT FROM SAMPLE CASES - Concluded

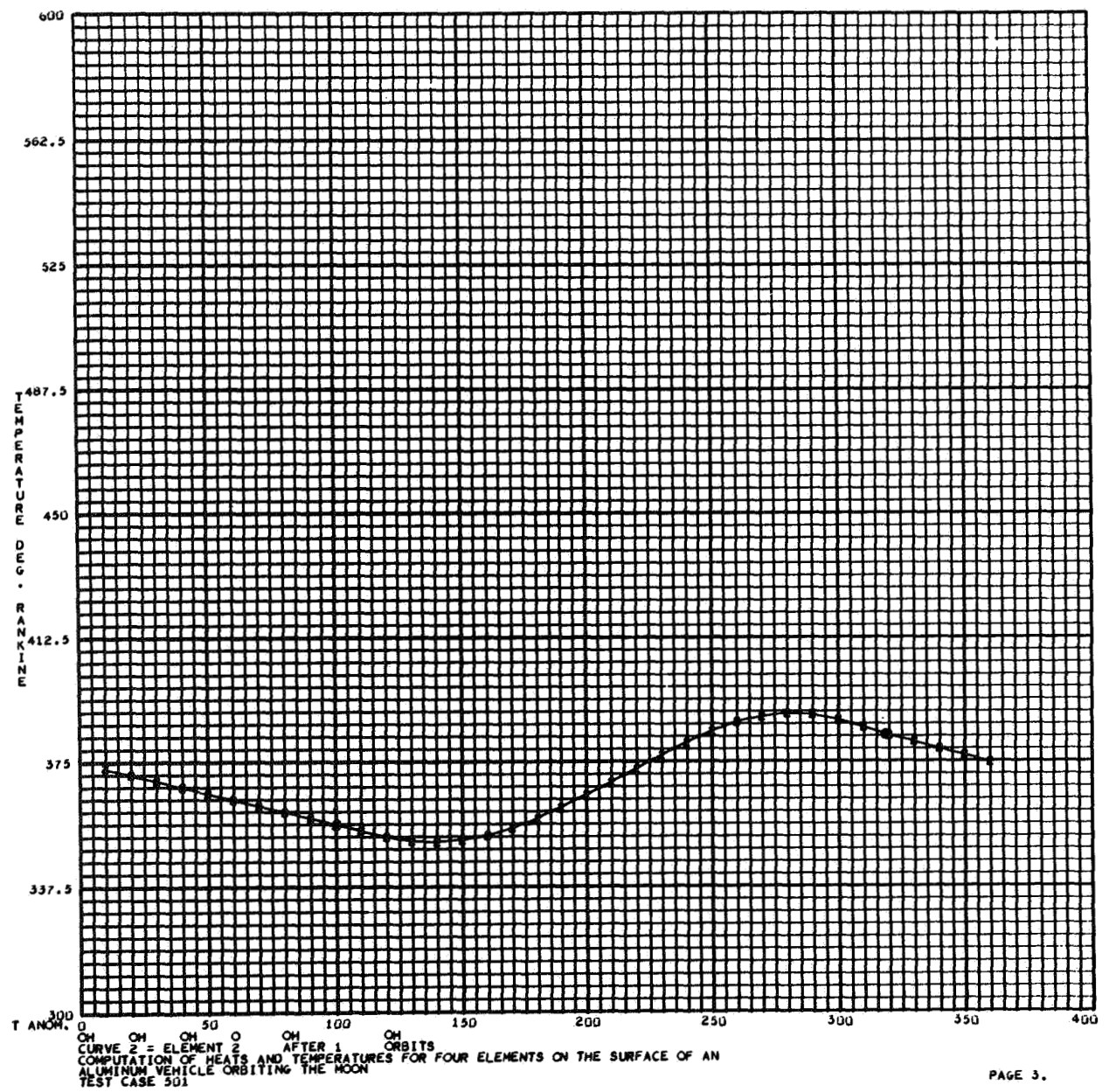
PHI	TIME	TEMPERATURE AND/OR HEAT FLUX
220.0	7.58	TEMPERATURES
	1	618.56 394.33 514.02 402.00
250.0	15.17	TEMPERATURES
	1	668.06 409.14 515.41 407.41
280.0	22.75	TEMPERATURES
	1	697.87 422.70 533.95 418.49
310.0	30.33	TEMPERATURES
	1	706.37 431.70 545.49 427.47
340.0	37.92	TEMPERATURES
	1	712.80 440.86 555.12 428.37
10.0	45.50	TEMPERATURES
	1	718.64 451.85 508.30 421.32

CALCULATION TIME FOR THIS CASE = .18 MINUTES...



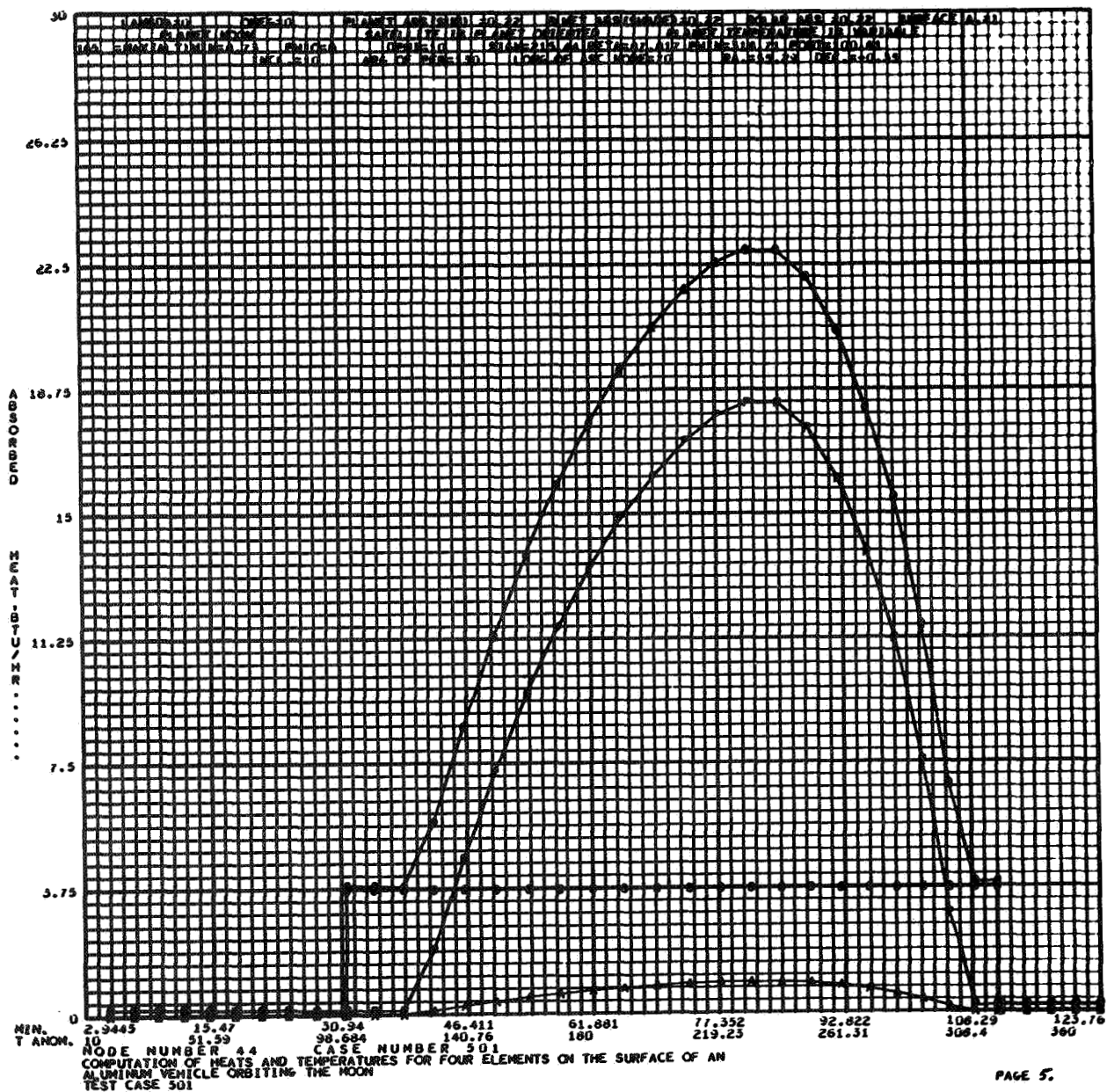
(a) Absorbed heat for node 1.

Figure G-1. - Plotted output from sample case 501.



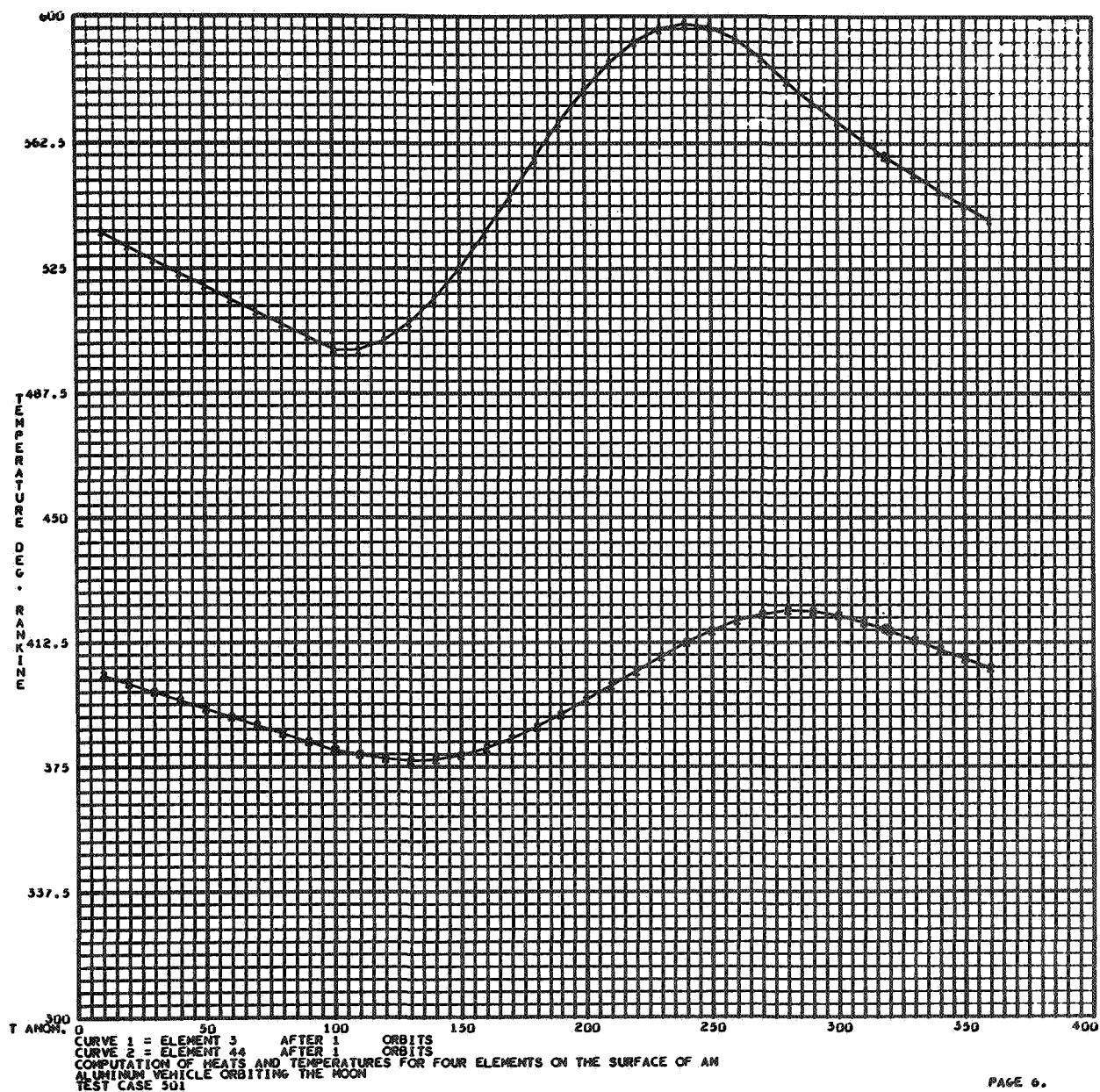
(c) Temperature for node 2.

Figure G-1. - Continued.



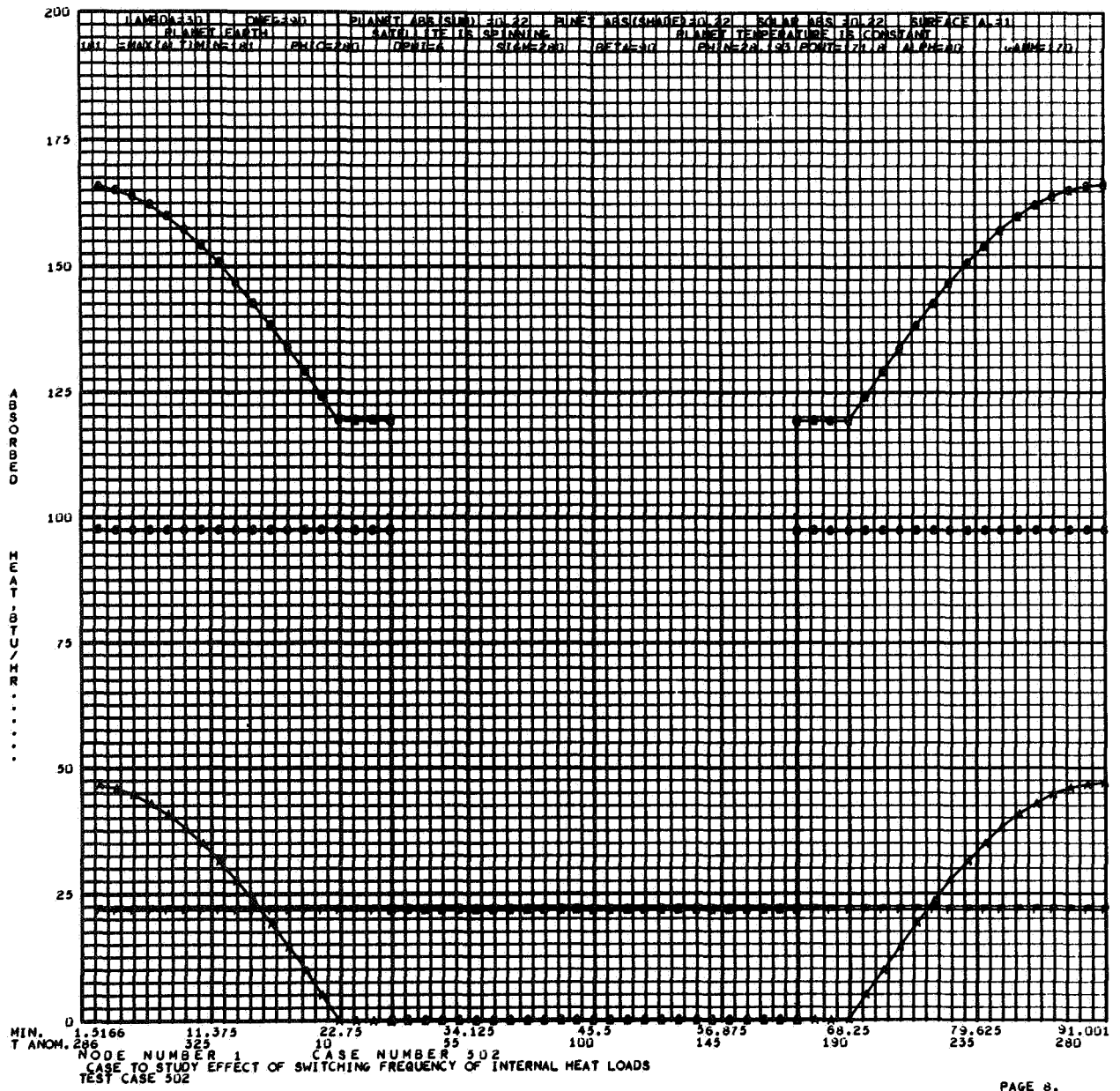
(e) Absorbed heat for node 4.

Figure G-1. - Continued.



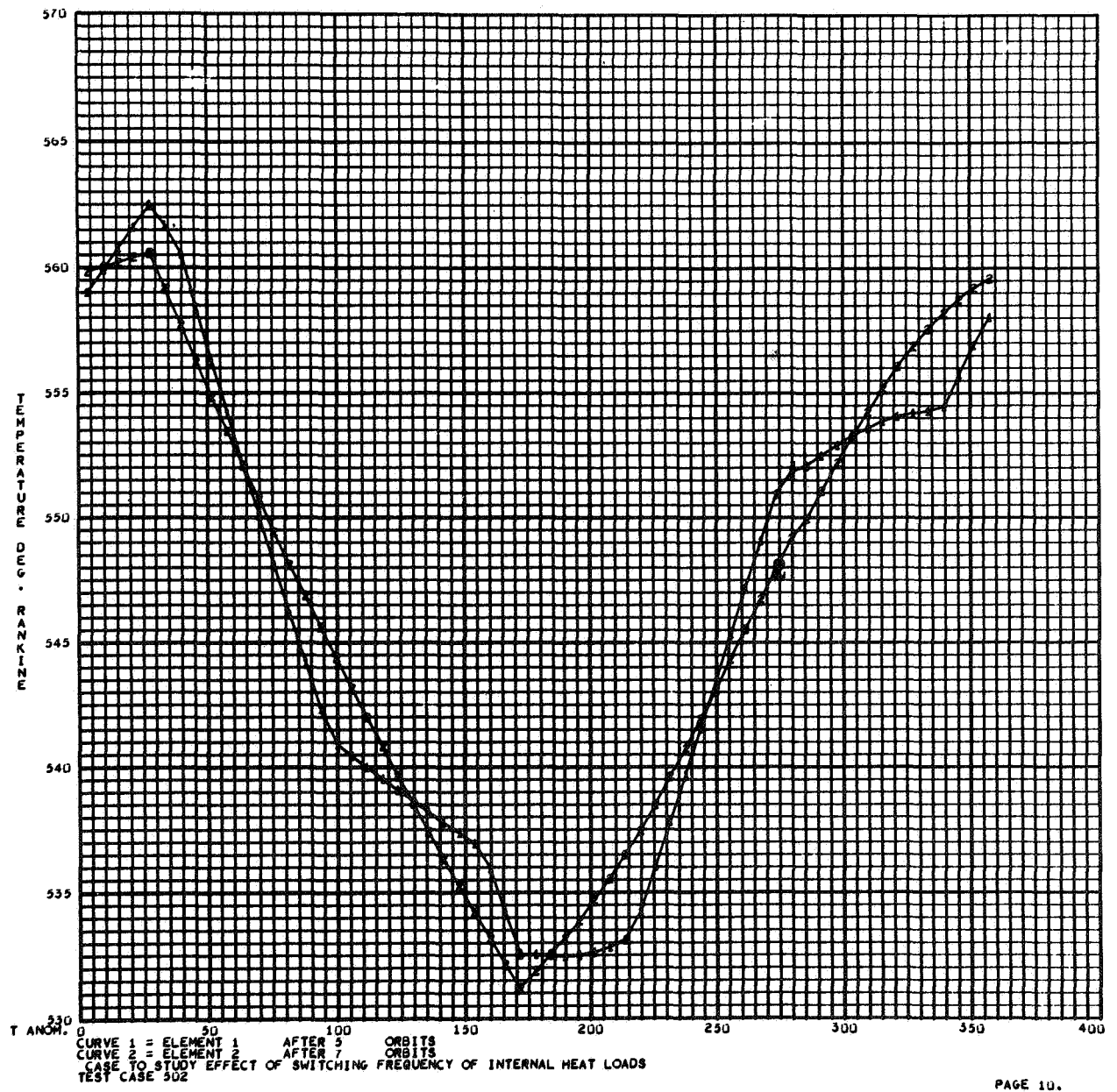
(f) Temperature for nodes 3 and 4.

Figure G-1. - Concluded.



(a) Absorbed heat for node 1.

Figure G-2. - Plotted output from sample case 502.



(c) Temperature for nodes 1 and 2.

Figure G-2. - Concluded.

APPENDIX H

OUTPUT PLOT OPTION

The SC-4020 data plotter can be instructed to produce two-dimensional plots and alphanumeric identification of the plots.

Upon request, plots of heat versus time, stabilized temperatures versus true anomaly, or both, are provided. The range for the independent variable is up to and including one complete orbit. The heat plots have a double abscissa scale, with the second scale being true anomaly. Heats can be either incident or absorbed, depending on the input code, with absorbed heat being also a function of input. Generous data identification is provided.

Description of Output

Information is taken from a magnetic tape produced by the program and reproduced visually on 35-mm film by the SC-4020. This film can be used to make enlarged reproductions on sensitized paper. The two types of plots obtainable (heat and stabilized temperature) are discussed separately.

Heat plots. - Plots of incident or absorbed heat versus time are obtained, one element per grid, unless the card type 01 requests temperatures only. The type of heat (albedo, planet, solar, or total) portrayed by each curve can be determined by examining the plotting symbols that correspond to each output point. The following are the symbol conventions:

<u>Plot symbol</u>	<u>Type of heat</u>	<u>Corresponding printout identification</u>
A	Albedo	QALBEDO
P	Planet	QPLAN
S	Solar	QSOLAR
Q	Total (present only on absorbed-heat plots)	QTOTAL

The curves are formed by connecting the points with straight lines. Thus, the smoothness of a curve is a function of the calculation interval.

Some of the parameters used by the program, plus other information, including vehicle-material absorptance with respect to planet-emitted radiation (PLANET ABS) and with respect to solar radiation (SOLAR ABS) are found in the upper portion of the grid.

The planet absorptance has two values, one for the period in which the vehicle is exposed to the Sun and one for the shaded portion of the orbit. When planet temperature is constant, both planet absorptance values, PLANET ABS (SUN) and PLANET ABS (SHADE), are constant throughout the orbit. With variable planet temperature, the PLANET ABS (SHADE) value is constant, but the PLANET ABS (SUN) value is not necessarily constant. If the emittance table used by a particular element is not constant over the entire sunlit effective-planet-temperature range, the PLANET ABS (SUN) value is an average.

Each axis of each plot is numerically labeled at nine locations that do not necessarily correspond to calculation points. The abscissa has two sets of labels, one for elapsed time (minutes) and another for true anomaly (degrees). The true-anomaly values correspond to the time values immediately above. Heat units (Btu/hr or Btu/hr-ft²) will be printed vertically to the left of the ordinate. Also present will be a notation of whether the heat is absorbed or incident. The element and case numbers are found below the true-anomaly scale.

The remaining portion of the frame will contain as many as three horizontal lines of information. These lines correspond to the first three comment cards (card type 10) physically input in the present case data. These comment cards should be used to identify the plot completely and tie it to the normal output.

Temperature plots. - When temperature calculations are requested by card type 01, plots of stabilized temperature versus true anomaly are obtained. If the temperature of an element has not been stabilized by the end of the last requested orbit, no temperature plot for the element will be given. The program logic determines whether one or two element temperatures will be plotted on a single grid. Each calculation point during the stabilization orbit will be represented on the curve by a numeral one or two. The curves are identified below the grid. As with the heat plots, the smoothness of the curve is a function of the calculation interval.

The vertical axis is numerically labeled with nine values. A vertical alphabetic label is also present. Immediately below the grid is a true-anomaly scale.

The case number, element number, and orbit during which the temperature became stabilized are given for each curve on the grid. Below these items will be the comment-card contents, as on the regular output and heat plots.

Stromberg-Carlson 4020 Data Plotter Requirements

The SC-4020 data plotter is a peripheral system designed to read magnetic-tape output from a digital computer program and to produce graphic and alphanumeric output. As the tape is read, the desired lines and characters are displayed on a cathode-ray tube and exposed to sensitized paper, film, or both. At MSC, the program user receives a strip of developed 35-millimeter negative film that contains the SC-4020 output. This film can then be put on a film viewer, and an enlarged, positive paper copy can be obtained.

When the computer program is run on a Univac 1108 computer at MSC, the routines necessary to generate SC-4020 control instructions are included in the system library. Program users, other than those at MSC, who wish to utilize the plot option must either obtain the routines from MSC or supply a compatible version of their own routines.

APPENDIX I

PROGRAM USER'S GUIDE FOR DATA PREPARATION

The table in this appendix contains detailed information on the preparation of data for the computer program discussed in this report. Throughout this appendix, the DP used in the format listings denote that a decimal point should be used within the specified column; RJ denotes that the data values must be justified on the right side within the specified columns.

MATERIAL-PROPERTIES GROUP

General-Comments Cards

Data-card description:

- . The use of the general comments cards is optional. When the card is used, it follows the permanent-data cards.
- . Any number of general-comments cards may be used in this location.
- . The comments are printed just above the material-properties "echo" check.

Format or columns	Contents	Data comments
1 to 2	10	. The 10 in columns 1 and 2 acts as a flag to the program to signify that general comments follow in columns 3 to 80.
3 to 80	General comments	. Any alphanumeric characters may be used. These columns are usually used for general comments common to the first and any succeeding case.

Material-Properties Table Count Card

Data-card description:

- . If comment cards are used, the material-properties table count card will follow. Otherwise, the card follows the permanent-data cards.
- . The material-properties table count card is common to the first and any succeeding case.

Material-Properties Table Count Card - Concluded

<u>Format or columns</u>	<u>Contents</u>	<u>Data Comments</u>
2	Number of optical-properties tables	. The total number of optical-properties tables to be loaded is entered in this column. Tables for all cases and not just for the first case must be loaded here, since this is the only place where the program accepts this type of data.
4	Number of substrate materials	. The maximum number of tables allowable is eight. . The total number of substrate materials to be loaded is entered in this column. Tables for all cases and not just for the first case must be loaded here, since this is the only place where the program accepts this type of data. . The maximum number of tables allowable is eight.

Optical-Properties Tables Cards

Data-card description:

- . All optical-properties tables are loaded next, one after the other.
- . Each table can have no more than seven cards. The desired optical-properties tables for the first and any succeeding cases must be given.
- . There is no requirement that all tables loaded must be used.
- . Columns 73 to 80 may be used to identify each table.

<u>Format or columns</u>	<u>Contents</u>	<u>Data comments</u>
(6E12. 8)DP	Emittance (infrared absorptance) as a	. Each table consists of alternating temperatures and emittance (infrared absorptance).

Optical-Properties Tables Cards - Concluded

<u>Format or columns</u>	<u>Contents</u>	<u>Data comments</u>
	function of source temperature	. Temperatures are in degrees Rankine.
	Solar absorptance as a constant value	. Each table terminates with the value that follows a temperature of 10 000° R. . The 10 000° R temperature acts as a flag within the program to signal the last pair of data in each table and is therefore required. . The value which follows and corresponds to the 10 000° R temperature, representing the temperature of the Sun, is the place at which the solar absorptance is loaded. Solar absorptance is a single value and cannot vary as a function of temperature.

Substrate-Properties Table Cards

Data-card description:

- . The substrate properties must follow the optical-properties tables.
- . For each substrate material, a pair of tables must be loaded, one containing specific heat versus temperature, the other density versus temperature.
- . Each table can have no more than seven cards.
- . Columns 73 to 80 may be used to identify each table.

Substrate-Properties Table Cards - Concluded

<u>Format or columns</u>	<u>Contents</u>	<u>Data comments</u>
(6E12. 8)DP	Specific heat as a function of element temperature	. The substrate-properties table consists of alternating temperatures and specific heat. . Temperatures are in degrees Rankine, and specific heat is in Btu/lb.
(6E12. 8)DP	Density as a function of element temperature	. As in the optical-properties tables, the last pair of data of each table must be flagged with the 10 000° R temperature. . This table consists of alternating temperatures and density. . Density should be given in lb/ft ³ . . The table is loaded in the same manner as the specific-heat table. . For each succeeding substrate material, a specific heat versus temperature table and a density versus temperature table are loaded.

CASE-DATA GROUP

Case-Comments Cards

Data-card description:

- . Case-comment cards are optional for each case. If used, they must go at the beginning of the case data.
- . Any number of case comment cards may be used in this location.
- . All comments are printed on the printout just above the respective case data "echo" check. If a plot routine is used, the first three case comment cards will be printed at the bottom of the respective frames for identification.

Case-Comments Cards - Concluded

<u>Format or columns</u>	<u>Contents</u>	<u>Data comments</u>
1 to 2	10	. The 10 in columns 1 and 2 acts as a flag to the program to signify general comments following in columns 3 to 80.
3 to 80	Case comments	. Any alphanumeric characters may be used. Case comments are usually used for case identification.

GENERAL NOTE PERTAINING TO THE FOLLOWING CASE-DATA CARDS

Except for the last card, the remainder of the data deck is made up of case-data cards, which are differentiated by the card code number appearing in columns 1 and 2. The card code number signifies the card type to the program, which may be numbered from 01 to 07. The card types are shown in numeric order below for convenience; however, there is no required order for card-type groups.

The input requirements have been minimized for parametric type analysis. In running back-to-back cases, the user does not have to load a particular card type, if it is desired to use all of the corresponding data from the preceding case. Therefore, one data card and a blank card to flag end of case may be sufficient data for any case except the first.

Except for a card type 04, the presence of a particular type of case-data card implies that all values normally defined on this card are now to be redefined, using values that follow. However, individual values may be changed or unchanged on the card type 04. If the card type 04, except for the first case, contains a blank in a particular data field, the program will use the corresponding data from the preceding case.

Case-Number and Output-Control Card

Data-card description:

Case-number and output-control card is referred to as card type 01.

<u>Format or columns</u>	<u>Contents</u>	<u>Data comments</u>												
1 to 2	01	. The 01 in columns 1 and 2 acts as a flag to the program to signify card type.												
4 to 8RJ	Case number	. The case number identifies the case. It is printed out on the printout at the beginning of each case, and if plots are requested it will appear on each corresponding frame. . The case number must be less than 32 768.												
10	Output-control code	. The output control applies to the printout and SC-4020 plots.												
	<table><tr><th><u>Output</u></th><th><u>Code</u></th></tr><tr><td>Incident heat flux only</td><td>0</td></tr><tr><td>Temperatures only</td><td>1</td></tr><tr><td>Incident heat flux and temperature</td><td>2</td></tr><tr><td>Incident and absorbed heats</td><td>3</td></tr><tr><td>Incident heat, absorbed heat, and temperature</td><td>4</td></tr></table>	<u>Output</u>	<u>Code</u>	Incident heat flux only	0	Temperatures only	1	Incident heat flux and temperature	2	Incident and absorbed heats	3	Incident heat, absorbed heat, and temperature	4	. When SC-4020 plots are requested (nonzero value in column 12 on this card), the only heats plotted will be absorbed heats.
<u>Output</u>	<u>Code</u>													
Incident heat flux only	0													
Temperatures only	1													
Incident heat flux and temperature	2													
Incident and absorbed heats	3													
Incident heat, absorbed heat, and temperature	4													
12	SC-4020 plot-control code	See appendix H for SC-4020 system requirements.												

Case-Number and Output-Control Card - Concluded

<u>Format or columns</u>	<u>Contents</u>	<u>Data comments</u>
	<u>Output</u>	<u>Code</u>
	Plots requested	Nonzero value
	No plots	Blank or zero

Print-Control and Angular-Interval Card

Data-card description:

. The print-control and angular-interval card is referred to as card type 02.

<u>Format or columns</u>	<u>Contents</u>	<u>Data comments</u>
1 to 2	02	. The 02 in columns 1 and 2 acts as a flag to the program as to signify card type.
3 to 4RJ	Print-control code (NPRINT)	. Blanks or a 1 cause printout after each interval. Other- wise, the program prints only after NPRINT intervals.
	<u>Print frequency</u>	<u>Code</u>
	Every compute interval	Blank, zero, or 1
	Every NPRINT compute interval	Value of NPRINT

Print-Control and Angular-Interval Card - Concluded

<u>Format or columns</u>	<u>Contents</u>	<u>Data comments</u>
6 to 8RJ	Printout format control	. With columnar format, the results are all printed out in columns.
	<u>Type</u>	
	Columnar	. The block format prints the results in rows; this generally takes less printout paper, but makes results more difficult to analyze.
	Block	. Block output cannot be requested when absorbed heats are called for; that is, when column 10 of the card type 01 has a 3 or 4.
11 to 18DP	Initial true anomaly θ_0	. Vehicle position at time zero is specified in degrees of arc.
19 to 26DP	Compute interval $\Delta\theta$. This is the increment in true anomaly which specifies the frequency of calculations. The basic angular interval $\Delta\theta$ should be a submultiple of 360; that is, 2, 5, 6, 7.5, 8, 9, 10 are suitable values but 7 is not, since 360/7 is not an integer. . If SC-4020 plots are requested (card type 01 column 12), $\Delta\theta$ must be $\geq 2^\circ$.
		. Care should be given to the choice of $\Delta\theta$, since all computed values are assumed constant over the interval unless the vehicle enters or leaves the planet shadow. From experience, a $\Delta\theta$ of 5° is generally acceptable.
27 to 34DP	Number of orbits	. The number of orbits or fractions of orbits that the vehicle is to make around the planet are specified.

Planet, Orbit, and Vehicle Attitude Card

Data-card description:

. The planet, orbit, and vehicle-attitude card is referred to as card type 03.

<u>Format or columns</u>	<u>Contents</u>	<u>Data comments</u>
1 to 2	03	. The 03 in columns 1 and 2 acts as a flag to the program to signify card type.
4	Planet code	. The planet to be orbited is called out by the corresponding code.

<u>Planet</u>	<u>Code</u>
Earth	1
Moon	2
Jupiter	3
Mars	4
Mercury	5
Neptune	6
Saturn	7
Uranus	8
Venus	9

Planet, Orbit, and Vehicle Attitude Card - Concluded

<u>Format or columns</u>	<u>Contents</u>	<u>Data comments</u>
5 to 6RJ	Vehicle-attitude code	. For definition of a spinning, planet-oriented, or Sun-oriented vehicle, see the section of this report entitled "Heat-Transfer Theory." The method and equations used internally for each of these cases are also given in this section.
	<u>Attitude</u>	<u>Code</u>
	Spinning	Blank or zero
	Planet oriented	1
	Sun oriented	-1
8	Constant or variable planet-temperature code	. The method and equations for handling a constant or variable planet temperature are given in the section of this report entitled "Heat-Transfer Theory."
	<u>Temperature</u>	<u>Code</u>
	Constant	Blank or zero
	Variable	1
51 to 65DP	Maximum orbit altitude (n. mi.)	. Constant planet temperature must be used for all celestial bodies except the Moon. The program sets the planet temperature code to zero (constant temperature) whenever any planet code except 2 is used in column 4.
66 to 80DP	Minimum orbit altitude (n. mi.)	

Element-Data Card

Data-card description:

- . The element-data card is referred to as card type 04.
- . If it is desired to change information pertaining to a particular element (or elements) in a succeeding case, cards type 04 need not be reinput for all elements. For example, if in case 1 there are three elements and if in case 2 it is desired to change only the skin thickness of element 2, it is necessary only to input a card type 04 containing a 2 in column 6 and the new thickness in columns 35 to 42. A blank card would also have to be present to signify end of case input.

<u>Format or columns</u>	<u>Contents</u>	<u>Data comments</u>
1 to 2	04	. The 04 in columns 1 and 2 acts as a flag to the program to signify card type.
4 to 6RJ	Element-number identification	. The element numbers for a particular case must be consecutive from one to NSATP (maximum = 200). NSATP is defined by card type 07 as the total number of elements present (that is, 1, 2, 3, 4, . . . NSATP). Although the elements must be numbered consecutively, they can be arranged in any order (that is, 1, 3, 2, NSATP, . . . 4).
8	Optical-properties table number	. This is the optical-properties table number for the element being described. The first optical-properties table loaded is optical-properties table number 1, et cetera. . The number may be from 1 to 8.
9	Substrate-properties table number	. This is the number of the substrate-properties table for the element being described. The first substrate material loaded is substrate-properties table number 1, and so forth. . The number may be from 1 to 8. . This input is not required when temperature calculations are not requested.

Element Data Card - Continued

<u>Format or columns</u>	<u>Contents</u>	<u>Data comments</u>
10	Internal-heat table number	<p>. This is the number of the internal-heat table, loaded on card type 06, for the element being described.</p> <p>. The number may be from 1 to 8.</p> <p>. This input is not required when temperature calculations are not requested.</p>
11 to 18DP	Λ'	<p>. The symbol Λ' is defined as the angle measured from the X_v-axis (toward Y_v) to the projection of the line connecting the center of the vehicle coordinate system and the element (referred to as the vehicle-vehicle element line) on the X_v-Y_v plane.</p> <p>. The symbol Λ' is given in degrees of arc, which may vary from 0° to 360°.</p> <p>. Another angle is needed to locate the element; this is Ω' and is defined in the next data field.</p> <p>. For a description of the vehicle coordinate system, refer to table D-I and the section of this report entitled "Celestial Mechanics Theory: Coordinate Systems."</p>
19 to 26DP	Ω'	<p>. The symbol Ω' is defined as the angle measured from the Z_v-axis to the vehicle-vehicle element line.</p> <p>. The symbol Ω' is given in degrees of arc, which may vary from 0° to 180°.</p> <p>. Note that if vehicle orientation or attitude is changed from the preceding case, it is necessary that Λ' and Ω' be determined again, since the vehicle axes change.</p>

Element Data Card - Concluded

<u>Format or columns</u>	<u>Contents</u>	<u>Data comments</u>
27 to 34DP	Initial element temperature ($^{\circ}\text{R}$)	. This value is not required when temperature calculations are not requested.
35 to 42DP	Element skin thickness (ft)	. This value is not required when temperature calculations are not requested.
43 to 48DP	Element surface area (ft^2)	. The loading of area is only applicable when absorbed heats are requested (card type 01). . Either all or none of the elements in a particular case may have an area input. Such an input causes the absorbed heat to be a rate (Btu/hr) rather than a flux (Btu/hr-ft^2).
51 to 56DP	Node number	. Each element may have an equivalent node number from 1 to 999. When a node number is loaded for a particular element, the program will then identify the respective answers by the node number and not element number. . If a node number is not given to a particular element, the respective answers will be identified by the element number. . The node number must have a decimal point.

Sun-Position Card

Data-card description:

.The Sun-position card is referred to as card type 05.

<u>Format or columns</u>	<u>Contents</u>	<u>Data comments</u>
1 to 2	05	.The 05 in columns 1 and 2 acts as a flag to the program to signify card type.
3 to 4RJ	Type of data for Sun position	.The code is used to signal what type of data is to be used to describe Sun position.
	<u>Data type</u>	
	Read in α , β , γ data	.If the Sun position is to be read in terms of angles α , β , and γ , a 3 must be punched in column 4. Sun position is then expressed directly with respect to the planet coordinate X_p , Z_p , and Y_p -axes in terms of α , β , and γ , respectively, as shown in figure 9. The planet coordinate system is summarized in table D-I.
	Read in ephemeris data	.If Sun position is taken from an ephemeris, a blank or any number other than 3 in column 4 will mean that instead of α , β , and γ being read in, i , ω , Ω , RA, and DEC will be read in.
		.Refer to the section of this report entitled "Celestial Mechanics Theory: Coordinate Systems" for definition of those angles required as input for the ephemeris data input option.
		.Sample calculations for obtaining RA and DEC of the Sun from an ephemeris are shown in appendix D.

Sun-Position Card - Continued

<u>Format or columns</u>	<u>Contents</u>	<u>Data comments</u>
6	<p>Compute Sun shade point code</p> <p><u>Option</u> <u>Code</u></p> <p>Compute ϕ_{in} Zero or and ϕ_{out} blank</p> <p>Input ϕ_{in} and 1 ϕ_{out}</p>	<p>. The points ϕ_{in} and ϕ_{out}, at which the vehicle passes in and out of the planet shadow, are normally computed from orbit geometry and Sun position. However, they may be read in if desired. This feature was provided principally as a debugging aid, but could conceivably be used in other ways.</p>
11 to 18DP	<p>Orbit inclination i, or α</p>	<p>. Load the angle of inclination i unless column 4 contains a 3, in which case, angle α is entered.</p> <p>. Both angles are in degrees of arc.</p> <p>. The angle of inclination i is the true inclination between the $X_c - Y_c$ plane and the orbital plane.</p> <p>. The value of i is always $\leq 90^\circ$.</p> <p>. The angle between the planet-Sun line and the X_p-axis is α.</p> <p>. The value of α is always $\leq 180^\circ$.</p>
19 to 26DP	<p>Argument of perifocus ω, or β</p>	<p>. Load ω, the argument of perifocus unless column 4 contains a 3, in which case, angle β is entered.</p> <p>. Both angles are in degrees of arc.</p>

Sun-Position Card - Continued

<u>Format or columns</u>	<u>Contents</u>	<u>Data comments</u>
27 to 34DP	Longitude Ω of the ascending node, or γ	<p>. The argument of perifocus ω is measured in the orbital plane, in the direction of travel from the right ascension of the ascending node to the perifocus. The vehicle direction of travel is always in the same direction as when the X_p-axis is rotated in the orbital plane towards Y_p through the smallest angle.</p> <p>. The value of ω is always $\leq 360^\circ$.</p> <p>. The angle between the planet-Sun line and the Z_p-axis is β.</p> <p>. The value of β is always $\leq 180^\circ$.</p> <p>. Load Ω, longitude of the ascending node, unless column 4 contains a 3, in which case, angle γ is entered.</p> <p>. Both angles are in degrees of arc.</p> <p>. The longitude Ω of the ascending node is measured counterclockwise in the X_c-Y_c plane from X_c to the line of nodes.</p> <p>. The value of Ω is always $\leq 360^\circ$.</p> <p>. The angle between the planet-Sun line and the Y_p-axis is γ.</p> <p>. The value of γ is always $\leq 180^\circ$.</p>
35 to 42DP	Right ascension RA of the Sun	<p>. Load RA of the Sun, unless column 4 contains a 3, in which case, the columns are ignored.</p> <p>. The right ascension should be given in degrees of arc.</p>

Sun-Position Card - Concluded

<u>Format or columns</u>	<u>Contents</u>	<u>Data comments</u>
45 to 50DP	Declination DEC of the Sun	.The RA is obtained from the ephemeris. See appendix D, which gives sample data preparation for RA and DEC of the Sun. .Load DEC, declination of Sun, unless column 4 contains a 3, in which case, the columns are ignored. .The DEC should be given in degrees of arc. .The DEC is also obtained from the ephemeris. .The program ignores this value unless a 1 is punched in column 6.
51 to 65DP	True anomaly θ_{in} at which vehicle enters planet shadow	
66 to 80DP	True anomaly θ_{out} at which vehicle leaves planet shadow	.The program ignores this value unless a 1 is punched in column 6.

Internal-Heat Table CardsData-card description:

- .The internal-heat table cards are referred to as card type 06.
- .Cards type 06 are optional and may be omitted entirely. An undefined heat table is assumed to contain no heat loads. Once a particular heat table is defined, it remains unchanged until another card type 06 redefines heat load for that table.
- .Each internal-heat table is loaded as step functions. The heat load is considered constant over a period of time, until a new switching time is given with a new constant heat-load value.

Internal-Heat Table Cards - Continued

.If the table contains switching times that are greater than the orbital period, those switching times are ignored, since the program resets the internal-heat tables to time zero at the end of each orbit.

<u>Format or columns</u>	<u>Contents</u>	<u>Data comments</u>
1 to 2	06	.The 06 in columns 1 and 2 acts as a flag to the program to signify card type.
4	Internal-heat table number	.This is the number of the internal-heat table being read in. .The program user may load in up to eight such tables.
5 to 6RJ	Number of heat values in table	.If greater than zero, this is the number of internal-heat values other than the starting value to be read into this table. .This number may be from 1 to 19. .To define or redefine an internal-heat-table so that it has no heat loads, blanks or zeros are punched in columns 5 and 6. .If the heat load is constant at some value other than zero, a 1 must be punched in column 6. Then columns 11 to 18 should contain the constant heat load, and columns 19 to 26 should contain a very large number such as 90 000, corresponding to the time switching occurs. An example of constant heat table is given in case 502 of the sample data deck (appendix G).
11 to 18DP	$Q_g(0)$, initial heat load for table indicated in column 4 (Btu/hr-ft ²)	
19 to 26DP	t_1 , time at which the first change of heat	

Internal-Heat Table Cards - Concluded

<u>Format or columns</u>	<u>Contents</u>	<u>Data comments</u>
27 to 34DP	load in this table will occur (min) $Q_g(t_1)$, heat-load value from t_1 (Btu/hr-ft ²)	
35 to 42DP	t_2 , time at which the second change of heat load in this table will occur (min)	
43 to 50DP	$Q_g(t_2)$, heat-load value from t_2 (Btu/hr-ft ²)	
51 to 65DP	t_3 , time at which the third change of heat load in this table will occur (min)	
66 to 80DP	$Q_g(t_3)$, heat load value from t_3 (Btu/hr-ft ²)	

Continuation cards for internal heat

If columns 5 and 6 contain a number from 4 to 19, then from one to four continuation cards are necessary. Continuation cards must follow immediately after the card type 06 that they complete and must be in chronological order. The format for continuation cards is exactly as described previously, except the first switching time of each continuation card is punched in columns 3 to 10, with a decimal point. Heat loads after switching are punched following corresponding switching times until all heat loads that were specified in columns 5 and 6 of the first card type 06 have been punched. An example of card type 06 continuation cards is given in case 502 of the sample data deck in appendix G.

Element Count Card

Data-card description:

. The element count card is referred to as card type 07.

<u>Format or columns</u>	<u>Contents</u>	<u>Data comments</u>
1 to 2	07	. The 07 in columns 1 and 2 acts as a flag to the program to signify card type.
4 to 6RJ	Total number of vehicle elements (NSATP)	. This must be the same as the total number of cards type 04. . The maximum allowable number of elements is 200.

End of Case Card

Data-card description:

<u>Format or columns</u>	<u>Contents</u>	<u>Data comments</u>
1 to 80	Blank	. Each case, including the first case, always ends with a blank card.

APPENDIX J

PROGRAM LISTING

```

$JOB  F4UP  PARKEK  005258 ED241 F007 355UF P 010 010 4020
'      ASG      A=NEWPCF
'      FOK      EXEC,EXEC
CMAIN PROGRAM (CALLED PILOT)
  DIMENSION TRASH(17),F(10,9,42),HIAB(9),ANGTAB(10),WH(18),XH(9),
  1YH(6),ZH(6),ELL(9),RPP(9),RR(9),CAYY(9),TIME(2)
  DIMENSION DT(2,200),T(2,200),SINL(200),COSL(200),SINO(200),
  1COSO(200),THICK(200),NCOAT(200),NSUBS(200),COSRS(200),PHIT(200),
  2GAMM(200),NDUTY(200)
  DIMENSION ESUN(8),EEE(8,42),RO(8,42),SP(8,42),TQINT(41,8)
  DIMENSION BUFFER(3)
  COMMON TRASH,F,HIAB,ANGTAB,WH,XH,YH,ZH,ELL,RPP,RR,CAYY,PI,PIH,
  1TWOP1,PI180,NOFINU,NQORT,IFIRST,NEWSIG
  COMMON KPLNET,NOKIEN,KTEMP,NUMRUN,NSATP,NPRINT,KREV,NP50,REV
  COMMON A,B,C,AYE,BEE,RP,RN,PEE,EL,R,CAY,BARL,S,ALP2,BFT2,GAM2,
  1ALPHA2,BETA2,GAMMA2,COSA,COSB,COSG,SINB,PHIMAX
  COMMON SIGMA,CSIGMA,SSIGMA,TSIGMA
  COMMON PHIZ2,DPHI2,PHIZ,DPHI,PHI,CPHI,SPHI,PHIN2,PHOT2,SUN
  COMMON TIMEZ,TABS,IELAPS,ZEIT,TIME,DEL TAT,XP,YP,DEF,OPSQ,J1,J2
  COMMON EP1P4,EPSIG2,IM,FTM,SAS2,SRASH
  COMMON G,KHRCR,RHO,CP,EPSLN,ITK,KITER
  COMMON QNET,QSAT,QINT,QEXT,QPLAN,QALB,QSUN,QOLD,QNEW,TBREAK
  COMMON GAM,PHIC,ALI,ALTI,ANGS,CTHEI,FD,FF
  COMMON PH11,PH12,ISIG,FUDGE,I4,BUFFER,RV,NLINE
  COMMON CUSLS,SINLS,UT,T,SINL,COSL,SINO,COSO,THICK,NCOAT,NSUBS,
  1COSRS,PH11,GAMM
  COMMON ESUN,EEE,RO,SP,TQINT,NDUTY,ANINCL,ASCNOD,ASNLNG,RGTASC,
  1DECLIN
  DIMENSION AA(6),AA1(6),P(6),P1(6)
  COMMON AA,AA1,P,P1,IORDER,IONDI,IFRRUR,THEIA,DTMAX,EN1,EN,FACT,
  1YNHAT,ENHATL,EMAG,UERROR,DTTEST
  COMMON HSUN,HALB,HPLAN,NODE
  DIMENSION HSUN(200),HALB(200),HPLAN(200),NODE(200)
  COMMON KETCH
  COMMON HASUN,HAALB,HAPLN,HATOT
  DIMENSION HASUN(200),HAALB(200),HAPLN(200),HATOT(200)
  COMMON ZAREA
  DIMENSION ZAREA(200)
  COMMON IMTHRU
  DIMENSION IMTHRU(200)
  COMMON IMHI,PNAME,PHIPLT,TIMPLT,NPLOI,LAST,JUMP,LMAX,IONCE
  DIMENSION PNAME(39),PHIPLT(190),TIMPLT(190)
  COMMON TYME1,TYME2,TYME
  COMMON/HKIK/JIKK
C *** SCRATCH TAPES UTILIZED BY LINK 1 ARE.. 4,9 AND 11
C *** SCRATCH TAPES UTILIZED BY LINK 2 ARE 9 AND 11
C*****
C *** OUTPUT TAPE IS 6, INPUT IS 5
C*****
C
C
  IONCE=0
255 JIJK=0
  IF(IONCE-1)2,1,2
  2 JUMP=1
  IONCE=1
  NPLOI=0
  1 IF(NPLOI)4,5,4
  4 TYME=TYME1+TYME2
  WRITE (6,9999)TYME1,TYME2,TYME
EXEC0000
EXEC0010
EXEC0020
EXEC0030
EXEC0040
EXEC0050
EXEC0060
EXEC0070
EXEC0080
EXEC0090
EXEC0100
EXEC0110
EXEC0120
EXEC0130
EXEC0140
EXEC0150
EXEC0160
EXEC0170
EXEC0180
EXEC0190
EXEC0200
EXEC0210
EXEC0220
EXEC0230
EXEC0240
EXEC0250
EXEC0260
EXEC0270
EXEC0280
EXEC0290
EXEC0300
EXEC0310
EXEC0320
EXEC0330
EXEC0340
EXEC0350
EXEC0360
EXEC0370
EXEC0380
EXEC0390
EXEC0400
EXEC0410
EXEC0420
EXEC0430
EXEC0440
EXEC0450
EXEC0460
EXEC0470
EXEC0480
EXEC0490
EXEC0500
EXEC0510
EXEC0520
EXEC0530
EXEC0540
EXEC0550
EXEC0560
EXEC0570

```

```

9999 FORMAT( /// 18H CALCULATION TIME=F6.2,13H , PLOT TIME=F6.2, EXEC0580
1 28H , TOTAL TIME FOR THIS CASE=F6.2,30H...ALL TIMES ARE IN MINUTEEXEC0590
25... ) EXEC0600
5 CONTINUE EXEC0610
C *** JUMP=2 MEANS WE HAVE RETURNED FROM LINK 2 (PLOT ROUTINES) EXEC0620
60 TO (3,290),JUMP EXEC0630
C          CONSTANIS FOR NUMERICAL INTEGRATION EXEC0640
3 IORDER=4 EXEC0650
FACT=1.0/(1.0-0.5**IORDER) EXEC0660
AA(1)=1.0/6.0 EXEC0670
AA(4)=AA(1) EXEC0680
AA(2)=AA(1)*2.0 EXEC0690
AA(3)=AA(2) EXEC0700
P(1)=0.0 EXEC0710
P(2)=.5 EXEC0720
P(3)=0.5 EXEC0730
P(4)=1.0 EXEC0740
IORDI=1 EXEC0750
IFIRST=1 EXEC0760
C CALL ROUTINE TO READ IN DATA AND PRINT HEADING EXEC0770
290 CALL TINPUT EXEC0780
IEKKOR=3 EXEC0790
IF(FUDGE+2669.0) 2669,2668,2669 EXEC0800
2668 IEKKOR=1 EXEC0810
2669 EN=0.0 EXEC0820
EMAG=0.0 EXEC0830
ENHAIL=0.0 EXEC0840
YNMAT=0.0 EXEC0850
C          CONSTANIS FOR DIFFERENTIAL EQUATION EXEC0860
SAS2=0.5*S EXEC0870
SRASH=SAS2*R EXEC0880
EP1P4=0.5*(1.0-R)*S EXEC0890
T4=SQRT(SQRT(S*(1.0-R)/6.856E-09)) EXEC0900
C          SET ALIERNATING INDICES EXEC0910
150 J1=1 EXEC0920
J2=2 EXEC0930
TIME(1)=0.0 EXEC0940
C CALL ROUTINE TO MOVE VEHICLE ALONG ITS PRESCRIBED ORBIT AND EXEC0950
C          PERFORM REQUIRED CALCULATION EXEC0960
CALL LOOP EXEC0970
IF(J1JK.NE.0)GO TO 255 EXEC0980
GO TO 290 EXEC0990
END EXEC1000

'      FOR      DECK1,DECK1      DK010000
SUBROUTINE HEAD      DK010010
C          PRINT OUT CAPTION PAGE      DK010020
WRITE (6,1)      DK010030
1 FORMAT(1H1////////////////)      DK010040
WRITE (6,2)      DK010050
2 FORMAT(40X,51HA COMPUTER PROGRAM FOR CALCULATING EXTERNAL THERMAL,DK010060
140X,      51HRADIATION HEAT LOADS AND TEMPERATURES OF SPACECRAFT,DK010070
246X,      38HORBITTING ABOUT THE PLANETS OR THE MOON      DK010080
3////      DK010090
464X,2HBY // 42X,47HMIDWEST RESEARCH INSTITUTE,KANSAS CITY,MISSOURIDK010100
5/63X,3HAND /44X,43HNASA MANNED SPACFCRAFT CENTER,HOUSTON,TEXAS      DK010110
6////////// 42X,52HDOCUMENTED IN THE NASA TECHNICAL REPORT      DK010120
7-R(S97) /44X,43HBY H.FINCH,R.VOGT,D.SOMMERVILLE,AND D.BLAND      DK010130
WRITE (6,611)      DK010140

```

611	FORMAT(1H1)	DK010150
	CALL FREAD	DK010160
	RETURN	DK010170
	END	DK010180
	FOR DECK2,DECK2	DK020000
	SUBROUTINE TINPUI	DK020010
	DIMENSION TRASH(17),F(10,9,42),HTAB(9),ANGTAB(10),WH(18),XH(9),	DK020020
	YH(6),ZH(6),ELL(9),RPP(9),RR(9),CAYY(9),TIME(2)	DK020030
	DIMENSION DT(2,200),T(2,200),SINL(200),COSL(200),SINO(200),	DK020040
	ICOSO(200),THICK(200),NCOAT(200),NSURS(200),COSRS(200),PHIT(200),	DK020050
	2GAMM(200),NDUTY(200)	DK020060
	DIMENSION ESUN(8),LEE(8,42),RO(8,42),SP(8,42),TQINT(41,8)	DK020070
	DIMENSION BUFFER(2)	DK020080
	COMMON TRASH,F,HTAB,ANGTAB,WH,XH,YH,ZH,ELL,RPP,RR,CAYY,PI,PIH,	DK020090
	1TWOP1,PIH0,NOFIND,NQORT,IFIRST,NEWSIG	DK020100
	COMMON KPLNE1,NOKIEN,KTEMP,NUMRUN,NSATP,NPRINT,KREV,NP50,REV	DK020110
	COMMON A,H,C,AYE,BEE,RP,RN,PEE,EL,R,CAY,BARL,S,ALP2,BFT2,GAM2,	DK020120
	1ALPHA2,BETA2,GAMMA2,COSA,COSB,COSG,SINB,PHIMAX	DK020130
	COMMON SIGMA,CSIGMA,SSIGMA,TSIGMA	DK020140
	COMMON PHI22,DPHI2,PHI2,DPHI,PHI,CPHI,SPHI,PHIN2,PHOT2,SUN	DK020150
	COMMON TIMEZ,TABS,IELAPS,ZEIT,TIME,DELTAT,XP,YP,DEE,DPSQ,J1,J2	DK020160
	COMMON EPTP4,EPSIG2,TM,FTM,SAS2,SRASH	DK020170
	COMMON G,HRHCP,RHO,CP,EPSLN,ITK,KITER	DK020180
	COMMON QNET,QSAT,QINT,QEXT,QPLAN,QALB,QSUN,QOLD,QNEW,TBREAK	DK020190
	COMMON GAM,PHIC,ALI,ALTI,ANGS,CTHEI,FD,FF	DK020200
	COMMON PHI1,PHI2,ISIG,FUDGE,T4,JUDGE,PUFFER,RV,NLINE	DK020210
	COMMON COSLS,SINLS,DT,T,SINL,COSL,SINO,COSO,THICK,NCOAT,NSURS,	DK020220
	1COSRS,PHIT,GAMM	DK020230
	COMMON ESUN,LEE,RO,SP,TQINT,NDUTY,ANINCL,ASCNOD,ASNLNG,RGTASC,	DK020240
	1UELLIN	DK020250
	DIMENSION AA(6),AA1(6),P(6),P1(6)	DK020260
	COMMON AA,AA1,P,P1,IORDER,IORD1,IERRUP,THEFA,DTMAX,EN1,EN,FACT,	DK020270
	1YNHAT,ENHATL,EMAG,UERROR,DTTEST	DK020280
	COMMON HSUN,HALB,HPLAN,NODE	DK020290
	DIMENSION HSUN(200),HALB(200),HPLAN(200),NODE(200)	DK020300
	COMMON KETCH	DK020310
	COMMON HASUN,HAALB,HAPLN,HATOT	DK020320
	DIMENSION HASUN(200),HAALB(200),HAPLN(200),HATOT(200)	DK020330
	COMMON ZAREA	DK020340
	DIMENSION ZAREA(200)	DK020350
	COMMON IMTHRU	DK020360
	DIMENSION IMTHRU(200)	DK020370
	COMMON IMHI,PNAME,PHIPLT,TIMPLT,NPLOI,LAST,JUMP,LMAX,IONCE	DK020380
	DIMENSION PNAME(39),PHIPLT(190),TIMPLT(190)	DK020390
	EQUIVALENCE(1RASH(15),C1),(TRASH(16),C2),(TRASH(17),C3)	DK020400
	DIMENSION W(7),Z(41),NZ(41)	DK020410
	EQUIVALENCE(1RASH(14),NBLANK),(Z(1),NZ(1))	DK020420
	DIMENSION TZ(200)	DK020430
	DIMENSION ELAMB(200),OMEGA(200)	DK020440
	DIMENSION TKALT(9)	DK020450
	COMMON TYME1,TYME2,TYME	DK020460
	COMMON AG, IN,J,KABG,KL, K,LPJ,LL,LN,L,LSHAUF,M,NCARD,	DK020470
	1NEWDC,NEWGAM,NEWMAT,NHEAD,N,NTRIG,PFE1,P6,PIN,POUT,ROP,SIGMA2	DK020480
	COMMON ELAMB,OMEGA,TKALT,TZ,W, AGNM,PGNM	DK020490
C ***	THE PURPOSE OF THE LAST 3 STATEMENTS (COMMON AG THRU PGNM)	DK020500
C	IS TO PRESERVE VALUES ESTABLISHED IN TINPUI. THIS COMMON	DK020510
C	(AND ASSOCIATED DIMENSION STATEMENTS) ALSO APPEARS IN THE	DK020520
C	MAIN PROGRAM OF LINK 2 .	DK020530

EQUIVALENCE (NAREA,AREA)	DK020540
NHEAD=1	DK020550
JFLAG=0	DK020560
DO 1005 IHL=1,39	DK020570
DATA 0000CT/0050505050505/	DK020580
1005 PNAME(IHL)=0000CI	DK020590
LN=0	DK020600
DO 392 I=1,N5ATP	DK020610
392 IM1HRU(I)=0	DK020620
KEWIND 4	DK020630
KEWIND 9	DK020640
KEWIND 11	DK020650
NEWGAM=0	DK020660
NEWDC=0	DK020670
NEWMAT=0	DK020680
IF(1FIKSI=1) 51,39,39	DK020690
C THIS SEQUENCE IS EXECUTED ONLY ONCE	DK020700
39 IFIRST=0	DK020710
KEICH=0	DK020720
CALL HEAD	DK020730
C SET PLANET COLD SIDE TEMPERATURES	DK020740
DO 391 J=3,8	DK020750
391 IKALI(J)=50.0	DK020760
IKALI(2)=186.0	DK020770
IKALI(5)=10.0	DK020780
IKALT(1)=200.0	DK020790
IKALI(4)=200.0	DK020800
IKALI(9)=200.0	DK020810
C READ COMMENT CARDS	DK020820
C NCAK() GREATER THAN 9 INDICATES COMMENT CARD	DK020830
C **** TAPE 3 WAS REPLACED BY TAPE 9 TO ACCOMMODATE THE CHAINED PROGRAMS	DK020840
C **** NECESSARY FOR PLOT OPTION. (SEPT. 14,1964)	DK020850
C COMMENTS ARE STORED ON TAPE 9 TO BE COPIED AFTER HEADING IS WRITTEN	DK020860
51 READ(5,6042) K,(Z(J),J=1,13)	DK020870
6042 FORMAT(12,13A6)	DK020880
IF(K)51,51,570	DK020890
570 IF(K=8) 6041,572,572	DK020900
572 WRITE(9,6042)K,(Z(J),J=1,13)	DK020910
IF(NHEAD=3)10000,10000,10001	DK020920
10000 DO 10002 LBJ=1,13	DK020930
LN=LN+LBJ	DK020940
10002 PNAME(LN)=Z(LBJ)	DK020950
NHEAD=NHEAD+1	DK020960
LN=LN+13	DK020970
10001 CONTINUE	DK020980
GO TO 51	DK020990
6041 JFLAG=1	DK021000
GO TO 52	DK021010
C BRANCH TO STORE INPUT DATA	DK021020
604 GO TO 605	DK021030
605 GO TO (1,2,3,4,5,6,7),NCARD	DK021040
1 NUMKUN=(100*K+L)*100+M	DK021050
NGOKI=N	DK021060
NPL01=w(1)	DK021070
GO TO 52	DK021080
2 NPKINT=MAX0(K,1)	DK021090
JUDGE=100*L+M	DK021100
PH1Z2=w(1)	DK021110
UPH1Z2=w(2)	DK021120
REV=w(3)	DK021130

PHIZ=PHIZ2*PI180	DK021140
DPHI=DPHI2*PI180	DK021150
C READ CASE DATA	DK021160
52 IF(JFLAG) 934,935,934	DK021170
934 READ(30,600) NCARD,K,L,M,N,(W(J),J=1,7)	DK021180
JFLAG=0	DK021190
GO TO 936	DK021200
935 READ(5,600) NCARD,K,L,M,N,(W(J),J=1,7)	DK021210
936 CONTINUE	DK021220
600 FORMAT(5I2,5F8.2,2E15.7)	DK021230
IF(NCARD) 99,99,604	DK021240
3 KPLNET=MAX0(K,1)	DK021250
C DO NOT PERMIT VARIABLE TEMPERATURES EXCEPT FOR THE MOON	DK021260
IF(KPLNET-2)2000,2005,2000	DK021270
2000 M=U	DK021280
2005 CONTINUE	DK021290
NORIEN=L	DK021300
KTEMP=M	DK021310
TM=TKALT(K)	DK021320
FUDGE=W(1)	DK021330
KOP=RPP(KPLNET)	DK021340
AG= W(6)*6076.1033	DK021350
PG= W(7)*6076.1033	DK021360
AGNM=W(6)	DK021370
PGNM=W(7)	DK021380
IF(AG-PG) 450,451,452	DK021390
450 AG=W(7) *6076.1033	DK021400
PG=W(6) *6076.1033	DK021410
AGNM=W(7)	DK021420
PGNM=W(6)	DK021430
WRITE (6,453)	DK021440
453 FORMAT (// 89H YOU INPUT AN ORBIT ALTIUDE MAX. LESS THAN THE MIN.	DK021450
1..1 REVERSED THEM AND SHALL CONTINUE. //)	DK021460
GO TO 452	DK021470
451 A= AG + KOP	DK021480
B= A	DK021490
C=SQRT(A*A -B*B)	DK021500
GO TO 460	DK021510
452 A= .5*(PG+ AG +2.0*KOP)	DK021520
C= A -PG -KOP	DK021530
B=SQRT(A*A -C*C)	DK021540
460 CONTINUE	DK021550
TM=.1714E-8*TM**4	DK021560
GO TO 52	DK021570
4 J=100*K+L	DK021580
IRASH(14)=WH(4)	DK021590
C A BLANK FIELD CAUSES PROGRAM TO USE PREVIOUS VALUE OF DATA	DK021600
READ(30,591) K,(Z(1),I=1,12),AREA,Z(13)	DK021610
591 FORMAT(I2,A4,A2,ZA1,4(A6,A2),A6,ZX,A6)	DK021620
READ(30,5914) XNODE	DK021630
5914 FORMAT (50X,F8.0)	DK021640
IF(NAREA-NBLANK)5911,5913,5911	DK021650
C *** IF AKFAS ARE INPUT, THERE MUST BE ONE FOR EACH ELEMENT	DK021660
5911 ZAREA(J)=W(5)	DK021670
KEICH=1	DK021680
5913 CONTINUE	DK021690
IF(NZ(5)-NBLANK) 401,402,401	DK021700
401 ELAMB(J)=W(1)*PI180	DK021710
NTRI6=1	DK021720
NEWGAM=1	DK021730

402	IF(NZ(7)-NBLANK) 403,404,403	DK021740
403	OMEGA(J)=W(2)*PI180	DK021750
4032	NTRIG=1	DK021760
	NEWGAM=1	DK021770
404	IF(NZ(9)-NBLANK) 405,406,405	DK021780
405	IZ(J)=W(3)	DK021790
406	IF(NZ(11)-NBLANK) 407,408,407	DK021800
407	THICK(J)=W(4)	DK021810
408	IF(NZ(2)-NBLANK) 409,410,409	DK021820
409	NCUAT(J)=M	DK021830
410	IF(NZ(3)-NBLANK) 411,412,411	DK021840
411	NSUBS(J)=N/10	DK021850
412	IF(NZ(4)-NBLANK) 413,414,413	DK021860
413	NDUTY(J)=MOD(N,10)	DK021870
414	NEWMAT=1	DK021880
4140	WRITE (4,591)K,(Z(I),I=1,12),AREA,Z(13)	DK021890
	IF(NZ(13)-NBLANK) 4141,42,4141	DK021900
4141	NODE(J)=XNODE	DK021910
42	IF(NODE(J)) 4142,4142,52	DK021920
4142	NODE(J)=J	DK021930
	GO TO 52	DK021940
5	KABG=K-3	DK021950
	LSHADE=L	DK021960
	PHIN2=W(6)	DK021970
	PHUT2=W(7)	DK021980
C	IF K IS 3, SUN POSITION IS GIVEN BY ALPHA,BETA,GAMMA	DK021990
C	IF NOT, IT IS GIVEN BY DATA FROM EPHEMERIS AND ORBIT DATA	DK022000
	IF(KABG) 501,508,501	DK022010
501	ANINCL=W(1)	DK022020
C	ASNLNG IS LARGE OMEGA	DK022030
C	ASCNOD IS SMALL OMEGA	DK022040
	ASCNOD=W(2)	DK022050
	ASNLNG=W(3)	DK022060
	KG(ASC=W(4)	DK022070
	DECLIN=W(5)	DK022080
	GO TO 52	DK022090
508	ALP2=W(1)	DK022100
	BET2=W(2)	DK022110
	GAM2=W(3)	DK022120
	ALPHA2=ALP2*PI180	DK022130
	GAMMA2=GAM2*PI180	DK022140
	BETA2=BET2 *PI180	DK022150
	COSA=COS(ALPHA2)	DK022160
	COSB=COS(BETA2)	DK022170
	COSG=COS(GAMMA2)	DK022180
	SINB=SIN(BETA2)	DK022190
	GO TO 52	DK022200
6	TQINT(40,K)=100000000.0	DK022210
	CALL QIIN(K,L,W(1))	DK022220
	NEWDC=1	DK022230
	GO TO 52	DK022240
7	NSATP=100*K+L	DK022250
	NSATP=MOD(NSATP,201)	DK022260
	GO TO 52	DK022270
C	WRITE CASE IDENTIFICATION	DK022280
C	WRITE TERMINATION MARK ON SCRATCH TAPES AND REWIND	DK022290
99	K=0	DK022300
	WRITE (9,6044)K,(WH(J),J=1,13)	DK022310
6044	FORMAT(I2,13A6)	DK022320
	REWIND 9	DK022330

	WRITE (11)K,K,(TQINT(J,1),J=1,41)	DK022340
	REWIND 11	DK022350
	WRITE (4,591)K,(Z(J),J=1,12),AREA,Z(13)	DK022360
	REWIND 4	DK022370
C	WRITE CASE HEADING	DK022380
	WRITE (6,640)NUMKUN	DK022390
640	FORMAT(9H1CASE NO.16)	DK022400
C	TRANSCRIBE COMMENTS	DK022410
	DO 101 L=1,15	DK022420
	READ (9,6044)K,(Z(J),J=1,13)	DK022430
	IF(K) 103,103,101	DK022440
101	WRITE (6,6043)(Z(J),J=1,13)	DK022450
6043	FORMAT(1X13A0)	DK022460
103	K=3*(NORLEN+2)	DK022470
102	REWIND 9	DK022480
	L=3*KTEMP+3	DK022490
	J=KPLNET+KPLNET	DK022500
	J1=1	DK022510
	WRITE (6,642)ZH(1),WH(J-1),WH(J),WH(4),ZH(2),ZH(3), XH(K-2),XH(K-	DK022520
	11),XH(K),WH(4),ZH(4),ZH(5),ZH(6),YH(L-2),YH(L-1),YH(L)	DK022530
642	FORMAT(16A6)	DK022540
	EL=EEL(KPLNET)	DK022550
	S=443.0*(HARL/EL)**2	DK022560
	CAY=CAYY(KPLNET)	DK022570
	RP=RPP(KPLNET)	DK022580
	K=KR(KPLNET)	DK022590
	KN=SQRT(A*A*A/CAY)/60.0	DK022600
	PEE=TWOP(*KN	DK022610
	PEE1=PEE	DK022620
C	IT IS NECESSARY TO RECOMPUTE SIN,COS OF LAMBDA AND OMEGA	DK022630
701	NEWGAM=0	DK022640
	NTRIG=0	DK022650
	DO 710 J=1,NSATP	DK022660
	SINO(J)=SIN(OMEGA(J))	DK022670
	COSO(J)=COS(OMEGA(J))	DK022680
	SINL(J)=SIN(ELAMB(J))	DK022690
	COSL(J)=COS(ELAMB(J))	DK022700
710	GAMM(J)=ARCOS(SINO(J)*COSL(J))	DK022710
715	IF(NORLEN) 718,720,720	DK022720
718	NTRIG=1	DK022730
C	FIND SIGMA	DK022740
720	NOFIND=LSHADE	DK022750
	CALL SIGBET(KABG,LSHADE)	DK022760
	SIGMA2=SIGMA/PI180	DK022770
C	PRINT SUN-SHADE POINTS	DK022780
	IF(PHIN2+900.0) 6431,6431,6432	DK022790
6431	PIN=0.0	DK022800
	POUT=0.0	DK022810
	SUN=2.0	DK022820
	GO TO 6433	DK022830
6432	PIN=AMOD(PHIN2+300.0,360.0)	DK022840
	POUT=AMOD(PHUT2+360.0,360.0)	DK022850
6433	WRITE (6,644)AGNM,PGNM,PHIZ2,DPH12,SIGMA2,BET2,PIN,POUT	DK022860
644	FORMAT (3X,28HMAX. * ORBIT ALT.(NM) * MIN.,4X,4HPH10,PX,4HDPHI,8X,	DK022870
	15HSIGMA,7X,4HBETA,8X,4HPHIN,8X,5HPOUT/1XF10.2,9XF11.2,6F12.5)	DK022880
	IF(KARG) 650,645,650	DK022890
645	WRITE (6,646)ALP2,GAM2	DK022900
646	FORMAT(7H ALPHA=F10.5,8H GAMMA=F10.5)	DK022910
	GO TO 990	DK022920
650	WRITE (6,651)ANINCL,ASCNOD,ASNLNG,KGLASC,DECLIN	DK022930

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651 FORMAT(13H INCLINATION=F10.5,24H ARGUMENT OF PERIFOCUS=F10.5,21H DK022940
1 LONG. OF ASC. NODE=F10.5,18H RIGHT ASCENSION=F10.5,14H DECLINATION=F10.5) DK022950
210N=F10.5) DK022960
990 IF(NEWDC) 150,150,110 DK022970
C TRANSCRIBE NEW INTERNAL HEAT LOADS DK022980
110 WRITE (6,109) DK022990
109 FORMAT(24H0NEW DUTY CYCLES READ IN) DK023000
DO 140 L=1,8 DK023010
READ (11)K,KL,(Z(J),J=1,41) DK023020
IF(K) 150,150,111 DK023030
111 N=KL+KL+1 DK023040
IF(Z(41)) 113,113,140 DK023050
113 N=2 DK023060
140 WRITE (6,139)K,(Z(J),J=1,N) DK023070
139 FORMAT(6H0INEXI2/(6H QIN=F8.4,4H I=F8.2,6H QIN=F8.4,4H T=F8.2) DK023080
1,6H QIN=F8.4,4H I=F8.2,6H QIN=F8.4,4H T=F8.2)) DK023090
150 IF(NEWMAT) 999,999,151 DK023100
C TRANSCRIBE NEW ELEMENTS DK023110
151 WRITE (6,149) DK023120
149 FORMAT(37H0ELEMENT COATING SUBSTRATE DUTY CYCLE5X5HLAMDA7X5HOMEGA7DK023130
1X4HT(0)6X9HTHICKNESS4X4HAREA4X8HNODE NO. ) DK023140
DO 170 L=1,200 DK023150
LL=L DK023160
READ (4,591)K,(Z(J),J=1,12),AREA,Z(13) DK023170
IF(K) 999,999,170 DK023180
170 WRITE (6,156)(Z(J),J=1,12),AREA,Z(13) DK023190
156 FORMAT(2XA4,0XA2,8XA1,9XA1,9XA6,A2, 4XA6,A2, 4XA6,A2, 4XA6,A2, DK023200
12X,A6,4X,A6) DK023210
C SET INITIAL TEMPERATURES DK023220
999 DO 998 J=1,NSATP DK023230
998 I(1,J)=TZ(J) DK023240
1000 KENDIND 11 DK023250
KENDIND4 DK023260
WRITE (6,461)REV DK023270
461 FORMAT (// 35H MAXIMUM NO. OF ORBITS REQUESTED= F7.3 ) DK023280
RETURN DK023290
END DK023300

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FOR DECK3,DECK3 DK030000
SUBROUTINE LOOP DK030010
DIMENSION TRASH(17),F(10,9,42),HTAB(9),ANG(AB(10),WH(18),XH(9), DK030020
1YH(6),ZH(6),ELL(9),RPP(9),RR(9),CAYY(9),TIME(2) DK030030
DIMENSION DT(2,200),T(2,200),SINL(200),COSL(200),SINO(200), DK030040
1COSO(200),THICK(200),NCOAT(200),NSUBS(200),COSRS(200),PHIT(200), DK030050
2GAMM(200),NDUTY(200) DK030060
DIMENSION ESUN(8),EEE(8,42),RO(8,42),SP(8,42),TQIN1(41,8) DK030070
DIMENSION BUFFER(2) DK030080
COMMON TRASH,F,HTAB,ANGTAB,WH,XH,YH,ZH,ELL,RPP,RR,CAYY,PI,PIH, DK030090
1IWOP1,PI180,NOFIND,NQORT,IFIRST,NEWSIG DK030100
COMMON KPLNET,NORIEN,KTEMP,NUMRUN,NSATP,NPRINT,KREV,NP50,REV DK030110
COMMON A,B,C,AYE,BEE,RP,RN,PEE,EL,R,CAY,BARL,S,ALP2,BET2,GAM2, DK030120
1ALPHA2,BETA2,GAMMA2,COSA,COSB,COSG,SINB,PHIMAX DK030130
COMMON SIGMA,CSIGMA,SSIGMA,TSIGMA DK030140
COMMON PHIZ2,DPH12,PHIZ,DPHI,PHI,CPH1,SPH1,PHIN2,PHOT2,SUN DK030150
COMMON TIME2,TABS,IELAPS,ZEIT,TIME,DELTAT,XP,YP,DFE,DPS0,J1,J2 DK030160
COMMON EPTP4,EPSIG2,TM,FTM,SAS2,SRASH DK030170
COMMON G,KHRLP,RHO,CP,EPSLN,ITK,KITEK DK030180
COMMON QNET,QSAT,QINT,QEXT,QPLAN,QALB,QSUN,GOLD,QNEW,TBREAK DK030190
COMMON GAM,PHIC,ALI,ALTI,ANGS,CHET,FD,FF DK030200

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COMMON PHI1,PHI2,ISIG,FUDGE,T4,JUDGE,PUFFER,RV,NLINE	DK030210
COMMON COSLS,SINLS,DT,T,SINL,COSL,SINO,COSO,THICK,NCOAT,NSUBS,	DK030220
1COSKS,PHIT,GAMM	DK030230
COMMON ESUN,EEE,R0,SP,TQINT,NDUTY,ANINCL,ASCNOD,ASNLCG,RGTASC,	DK030240
1DECLIN	DK030250
DIMENSION AA(6),AA1(6),P(6),P1(6)	DK030260
COMMON AA,AA1,P,P1,IORDER,IORD1,1ERROR,THETA,DTMAX,EN1,EN,FACT,	DK030270
1YNHAT,ENHATL,EMAG,1ERROR,DTTEST	DK030280
COMMON HSUN,HALB,HPLAN,NODE	DK030290
DIMENSION HSUN(200),HALB(200),HPLAN(200),NODE(200)	DK030300
COMMON KETCH	DK030310
COMMON HASUN,HAALB,HAPLN,HATOT	DK030320
DIMENSION HASUN(200),HAALB(200),HAPLN(200),HATOT(200)	DK030330
COMMON ZAREA	DK030340
DIMENSION ZAKEA(200)	DK030350
COMMON IMTHRU	DK030360
DIMENSION IMIHRU(200)	DK030370
COMMON IMHI,PNAME,PHIPLT,TIMPLT,NPLOT,LAST,JUMP,LMAX,IONCE	DK030380
DIMENSION PNAME(39),PHIPLT(190),TIMPLT(190)	DK030390
COMMON TYME1,TYME2,TYME	DK030400
DIMENSION TPKEV(200)	DK030410
DIMENSION ISIBL(200)	DK030420
COMMON/HKIK/JIJK	DK030430
DO 300 I=1,200	DK030440
300 IMTHRU(I)=0	DK030450
IMHI=NSATP	DK030460
CALL RESET	DK030470
DO 407 IZ=1,190	DK030480
PHIPLT(IZ)=0.0	DK030490
407 TIMPLT(IZ)=0.0	DK030500
C RESET LINE COUNT AND PRINT PAGE HEADING	DK030510
IF(JUDGE-359) 710,711,710	DK030520
710 IF(NQORT-2)600,600,601	DK030530
601 ND=NQORT-2	DK030540
GO TO 712	DK030550
600 ND=-NQORT	DK030560
GO TO 712	DK030570
711 IF(NQORT-2)603,603,604	DK030580
C WE CALL TOUT BECAUSE IT IS NOT POSSIBLE TO HAVE A BLOCK OUTPUT	DK030590
C WHEN NQORT IS GREATER THAN 2. TOUT WILL CATCH THIS ERROR,	DK030600
C PRINT A MESSAGE, AND CALL EXIT...	DK030610
604 CALL TOUT(L)	DK030620
CALL EXIT	DK030630
603 ND=-3	DK030640
712 CALL TALLY(NLINE,-1,ND)	DK030650
IF(NQORT*NQORT -3*NQORT)730,732,730	DK030660
C COMPUTE MAXIMUM TIME INTERVAL	DK030670
730 PHI1=(180.0-UPHI2*0.5)*PI180*0.5	DK030680
ZZ1=SIGN(PHI1)/COS(PHI1)	DK030690
ZZ1=ATAN((A-C)*ZZ1/B)*2.0	DK030700
IF(ZZ1)861,861,871	DK030710
861 ZZ1=ZZ1+2.0*PI	DK030720
871 SINZZ1=SIN(ZZ1)	DK030730
ZZ1=KN*(ZZ1-C*SINZZ1/A)	DK030740
PHI1=PHI1+UPHI2*PI180*0.5	DK030750
ZZ2=SIGN(PHI1)/COS(PHI1)	DK030760
ZZ2=ATAN((A-C)*ZZ2/B)*2.0	DK030770
IF(ZZ2) 961,961,971	DK030780
961 ZZ2=ZZ2+2.0*PI	DK030790
971 SINZZ1=SIN(ZZ2)	DK030800

C	DTMAX=RN*(ZZ2-C*SINZZ1/A)-ZZ1	DK030810
	SFT INITIAL TEMPERATURES FOR STEADY STATE TESTS	DK030820
972	DO 731 J=1,NSATP	DK030830
	IF(IMTHRU(J)) 731,733,731	DK030840
733	TPREV(J)=T(1,J)	DK030850
731	CONTINUE	DK030860
732	LINC=360.0/DPHI2+0.5	DK030870
	KREV=REV+.9999	DK030880
	ISUN=1	DK030890
C	TAPES 9 (A5) AND 11 (A6) ARE USED AS TEMPORARY STORAGE FOR TEMP.	DK030900
C	AND OTHER VALUABLE INFORMATION...	DK030910
500	IT=9	DK030920
	JT=11	DK030930
	REWIND IF	DK030940
	REWIND JT	DK030950
	KEEP=0	DK030960
	NPIS=0	DK030970
	NDU=0	DK030980
	NSS=0	DK030990
	PHIBAD=-999.0	DK031000
C ***	ESTABLISH PHI-IN(FEIN) AND PHI-OUT(FEOUT) VALUES TO BE USED IN	DK031010
C ***	TESTS FOR BAD PHI	DK031020
	FEIN=AMOD(PHI2+360.0,360.0)	DK031030
	FEOUT=AMOD(PHI2+360.0,360.0)	DK031040
501	DO 850 K=1,KREV	DK031050
	NSIBL=1	DK031060
	NORBIT=K	DK031070
	DO 444 IJK=1,NSATP	DK031080
444	ISIBL(IJK)=0	DK031090
	KV=K	DK031100
	LKKV=REV-KV	DK031110
	IF(NPLOT) 441,442,441	DK031120
441	IF(NQORT-3)442,431,431	DK031130
431	IF(URRV)432,433,433	DK031140
432	IF(K-1)433,433,442	DK031150
C ***	INITIALIZE AVG. EPP CALCULATION	DK031160
433	CALL HEAT (201)	DK031170
442	IF(URRV)741,750,750	DK031180
C	FRACTIONAL ORBIT	DK031190
741	LMAX=360.0*(URRV+1.0)/DPHI2+0.5	DK031200
	GO TO 751	DK031210
C	COMPLETE ORBIT	DK031220
750	LMAX=LINC	DK031230
751	DO 800 L=1,LMAX	DK031240
	EEL=L	DK031250
	LL=L	DK031260
	ISKIP=0	DK031270
752	PHIABS=PHI2+EEL*DPHI2	DK031280
C ***	IF PHIABS IS IN A TROUBLE SPOT, SKIP OUTPUT AT THAT POINT	DK031290
	FE=AMOD(PHIABS,360.0)	DK031300
	IF(FE-FEIN)418,419,419	DK031310
419	IF(FE-FEIN-.1)421,421,418	DK031320
418	IF(FE-FEOUT)422,422,423	DK031330
422	IF(FE-FEOUT+.1)423,421,421	DK031340
421	ISKIP=1	DK031350
	IF(ISUN-2)423,800,800	DK031360
423	CONTINUE	DK031370
	IF(SUN-2.0)3,10,10	DK031380
C	FIND POSITION IN ORBIT	DK031390
3	GO TO (200,250),ISUN	DK031400

200	TEST=(PHIABS-PHIN2)*(PHIABS-PHOT2)	DK031410
	IF(TEST)260,260,265	DK031420
260	IF(SUN)10,10,210	DK031430
C	SUN-SHADE POINT JUST PASSED -- REDEFINE PHIABS	DK031440
210	NSS=-1	DK031450
	IF(NDO)411,411,412	DK031460
C ***	PHIN2 , SUN	DK031470
411	PHIABS=PHIN2	DK031480
	ISUN=2	DK031490
	SUN=1.0	DK031500
	GO TO 10	DK031510
C ***	PHIN2 , SHADE	DK031520
412	PHIABS=PHIN2 + .1	DK031530
	SUN=0.0	DK031540
	GO TO 10	DK031550
265	TEST=(PHIABS-PHIN2-360.0)*(PHIABS-PHOT2-360.0)	DK031560
	IF(TEST) 260,260,266	DK031570
266	IF(SUN) 205,205,10	DK031580
C	SHADE-SUN POINT JUST PASSED -- REDEFINE PHIABS	DK031590
205	NSS=1	DK031600
	IF(NDO)413,413,414	DK031610
C ***	PHOT2 , SHADE	DK031620
413	PHIABS=PHOT2 -.1	DK031630
	ISUN=2	DK031640
	SUN=0.0	DK031650
C ***	IF FEOUT=.1 , AT FIRST CALCULATION POINT OF FIRST ORBIT, IS LESS	DK031660
	IF(NORBIT*LL-1)427,427,428	DK031670
C ***	THAN PHIZ2 (PHI AT TIME ZERO) WE SHALL ALWAYS SKIP THAT POINT	DK031680
427	IF (FEOUT-.1-PHIZ2)429,429,10	DK031690
429	PHIBAD=PHIABS	DK031700
428	IF (ABS(PHIABS-PHIBAD)-.000001)430,430,10	DK031710
C ***	PHOT2 , SUN	DK031720
414	PHIABS=PHOT2	DK031730
	SUN=1.0	DK031740
	GO TO 10	DK031750
C	SUN-SHADE POINTS ALREADY CHECKED DURING THIS INTERVAL	DK031760
250	CONTINUE	DK031770
	ISUN=1	DK031780
C	CONVERT PHI TO FIRST FOUR QUADRANTS	DK031790
10	PHI=AMOD(PHIABS,360.0)	DK031800
	IF (SKIP)425,425,424	DK031810
424	IF (ABS(FE-PHI)-.000001)426,426,425	DK031820
426	SKIP=0	DK031830
	GO TO 800	DK031840
425	CALL LOCUS	DK031850
101	IF (TABS=TIME(J1)+0.5*PEE) 102,12,12	DK031860
102	TABS=TABS+PEE	DK031870
	GO TO 101	DK031880
12	TIME(J2)=TABS	DK031890
	ZEIT=TABS	DK031900
	IF (KEEP)400,401,400	DK031910
401	IF (K-1)403,403,402	DK031920
402	KEEP=1	DK031930
	GO TO 400	DK031940
403	IF (NPL01)443,508,443	DK031950
443	NPTS=NPTS+1	DK031960
	IF (ABS(PHI)-.01)404,404,405	DK031970
404	PHIPLT(NPTS)=360.0	DK031980
	GO TO 406	DK031990
405	PHIPLT(NPTS)=AMOD(PHI+360.0,360.0)	DK032000

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406 TIMPLT(NPTS)=ZEIT                                DK032010
GO TO 508                                              DK032020
400 READ (IT)(T(J2,J),HSUN(J),HALB(J),HPLAN(J),HASUN(J),HAALB(J),HAPL DK032030
IN(J),HATOT(J),J=1,NSATP)                          DK032040
508 DO 13 J=1,NSATP                                  DK032050
IF(NQORT*NQORT -3*NQORT)503,507,503 .              DK032060
503 IF(IMTHRU(J))13,505,13                          DK032070
505 CALL TEMPER(J)                                    DK032080
GO TO 13                                              DK032090
507 CALL HEAT(J)                                      DK032100
13 CONTINUE                                           DK032110
IF(DRRV)21,23,23                                     DK032120
21 IF(K-1)23,23,22                                   DK032130
23 LAST=JT                                            DK032140
WRITE (JT)(T(J2,J),HSUN(J),HALB(J),HPLAN(J),HASUN(J),HAALB(J),HAPL DK032150
IN(J),HATOT(J),J=1,NSATP)                          DK032160
22 CONTINUE                                           DK032170
ND= MOD(L,NPRINT)                                    DK032180
14 IF(ISUN=2) 15,16,16                              DK032190
15 IF(ND) 17,16,17                                   DK032200
C PRINT AFTER EVERY NPRINT INCREMENTS              DK032210
16 CALL TOUT(L)                                       DK032220
17 JUB=J1                                             DK032230
J1=J2                                                 DK032240
J2=JUB                                                DK032250
430 IF(NSS)416,415,416                               DK032260
416 IF(NDO)417,417,415                               DK032270
417 NDO=1                                             DK032280
IF(NSS)210,210,205                                   DK032290
415 NDO=0                                             DK032300
NSS=0                                                 DK032310
IF(ISUN=2) 800,752,752                               DK032320
800 CONTINUE                                         DK032330
IF(NPLOT)439,438,439                                 DK032340
439 IF(NQORT=3)438,437,434                           DK032350
434 IF(DRRV) 435,436,436                             DK032360
435 IF(K-1)437,437,438                               DK032370
436 IF(K-KREV) 438,437,437                           DK032380
C *** CALCULATE AVG. EPP IF IT IS TIME TO DO SO    DK032390
437 CALL HEAT (202)                                  DK032400
438 CONTINUE                                         DK032410
IF(DRRV)99,410,410                                   DK032420
C IF TEMPERATURE CYCLE HAS STABILIZED, HALT COMPUTATION DK032430
410 IF(NQORT*NQORT -3*NQORT)802,99,802              DK032440
C IMTHRU(I) = 0 MEANS THAT THE TEMPERATURE OF ELEMENT I HAS DK032450
C NOT STABILIZED                                     DK032460
802 DO 805 IX=1,NSATP                                DK032470
IF(IMTHRU(IX))805,810,805                             DK032480
810 IF(ABS(TPREV(IX)-T(J1,IX))= 0.5) 803,803,805    DK032490
803 IMTHRU(IX)= K                                    DK032500
ISTBL(NSIBL)=IX                                       DK032510
NSTBL=NSTBL+1                                        DK032520
805 CONTINUE                                         DK032530
NSTBL=NSTBL-1                                        DK032540
WRITE (6,854)ZEIT,NORBIT                             DK032550
854 FORMAT(/ 20H THE OUTPUT AT TIME= F8.2,19H ENDS ORBIT NUMBER 13) DK032560
NLIN=NLIN+5                                           DK032570
WRITE (6,408)(ISIBL(IJK),IJK=1,NSTBL)               DK032580
408 FORMAT ( 68H THE TEMPERATURES OF THESE NODES STABILIZED DURING THE DK032590
1 LAST ORBIT... 1215 / (1X,2615) )                 DK032600

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DO 808 I=1,NSATP	DK032610
IF(IMTHRU(I))808,809,808	DK032620
808 CONTINUE	DK032630
C *** CALCULATE AVG. EPP IF ALL NODES HAVE STABILIZED	DK032640
IF(NPLOT)440,99,440	DK032650
440 CALL HEAT(202)	DK032660
GO TO 99	DK032670
809 CONTINUE	DK032680
DO 806 IX=1,NSATP	DK032690
IF(IMTHRU(IX))806,807,806	DK032700
807 IPREV(IX)=T(J1,IX)	DK032710
IMHI=IX	DK032720
806 CONTINUE	DK032730
C RESET STARTING INDEX IN INTERNAL HEAT TABLES	DK032740
DO 830 J=1,8	DK032750
IF(TQINT(41,J)-2.0) 830,830,828	DK032760
828 TQINT(41,J)=2.0	DK032770
830 CONTINUE	DK032780
KI=J1	DK032790
JT=JT	DK032800
JT=KI	DK032810
REWIND II	DK032820
REWIND JT	DK032830
850 CONTINUE	DK032840
99 CONTINUE	DK032850
CALL CLOCK(TIME1)	DK032860
IF(NPLOT)10003,10002,10003	DK032870
10002 TIME2=0.0	DK032880
WRITE(6,9999)TIME1	DK032890
9999 FORMAT(//// 33H CALCULATION TIME FOR THIS CASE = F6.2,	DK032900
1 11H MINUTES...)	DK032910
RETURN	DK032920
10003 CONTINUE	DK032930
C *** SNEAK AVG. EPP INTO LINK 2	DK032940
CALL HEAT(203)	DK032950
LMAX=NPIS	DK032960
WRITE(6,409)	DK032970
409 FORMAT(/// 6RH S-L 4020 PLOTS HAVE BEEN REQUESTED AND SHALL BE PROVIDED BY LINK 2)	DK032980
CALL MAIN2	DK032990
J1K=1	DK033000
RETURN	DK033010
END	DK033020
	DK033030
FOR DECK4,DECK4	DK040000
SUBROUTINE TOUT(L)	DK040010
DIMENSION TRASH(17),F(10,9,42),H1AB(9),ANG(AB(10),WH(18),XH(9),	DK040020
1YH(6),ZH(6),ELL(9),RPP(9),RR(9),CAYY(9),TIME(2)	DK040030
DIMENSION DT(2,200),T(2,200),SINL(200),COSL(200),SINU(200),	DK040040
1COSO(200),THICK(200),NCOAT(200),NSURF(200),COSRS(200),PHIT(200),	DK040050
2GAMM(200),NDUTY(200)	DK040060
DIMENSION ESUN(8),EEF(8,42),RO(8,42),SP(8,42),TQINT(41,8)	DK040070
DIMENSION BUFFER(2)	DK040080
COMMON TRASH,F,H1AB,ANGTAR,WH,XH,YH,ZH,ELL,RPP,RR,CAYY,PI,PIH,	DK040090
1IWOP1,PI180,NOFINU,NQORT,IFIRST,NEWSIG	DK040100
COMMON KPLNE1,NORLEN,KTEMP,NUMRUN,NSATP,NPRINT,KREV,NP50,REV	DK040110
COMMON A,H,C,AYF,BEE,RP,RN,PEF,EL,R,CAY,BARL,S,ALP2,BFT2,GAM2,	DK040120
1ALPHA2,BETA2,GAMMA2,COSA,COSB,COSG,SINB,PHIMAX	DK040130
COMMON SIGMA,CSIGMA,SSIGMA,TSIGMA	DK040140

COMMON PHIZ2,DPHI2,PHIZ,DPHI,XHI,CPHI,SPHI,PHIN2,PHOT2,SUN	DK040150
COMMON TIMEZ,TARS,TELAPS,ZEIT,TIME,DELTAT,XP,YP,DEE,UPS0,J1,J2	DK040160
COMMON EPTP4,EPS162,TM,FTM,SAS2,SRASH	DK040170
COMMON G,RHRC,P,RHO,CP,EPSLN,ITK,KITER	DK040180
COMMON QNET,QSAT,QINT,QEXT,QPLAN,QALB,QSUN,QOLD,QNEW,TBREAK	DK040190
COMMON GAM,PHIC,ALI,ALTI,ANGS,CTHET,FD,FF	DK040200
COMMON PH11,PHI2,ISIG,FUDGE,TPL,JUDGE,BUFFER,RV,NLINE	DK040210
COMMON COSLS,SINLS,DT,T,SINL,COSL,SIN0,COS0,THICK,NCOAT,NSUBS,	DK040220
1COSRS,PH11,GAMM	DK040230
COMMON ESUN,EEE,R0,SP,TQINT,NDUTY,ANINCL,ASCNOD,ASNLNG,RGTASC,	DK040240
1UECLIN	DK040250
DIMENSION AA(6),AA1(6),P(6),P1(6)	DK040260
COMMON AA,AA1,P,P1,IORDER,IORD1,IEKRUP,THETA,DTMAX,EN1,EN,FACT,	DK040270
1YNHAT,ENHATL,EMAG,UEERROR,DTTEST	DK040280
COMMON HSUN,HALB,HPLAN,NODE	DK040290
DIMENSION HSUN(200),HALB(200),HPLAN(200),NODE(200)	DK040300
COMMON KETCH	DK040310
COMMON HASUN,HAALB,HAPLN,HATOT	DK040320
DIMENSION HASUN(200),HAALB(200),HAPLN(200),HATOT(200)	DK040330
COMMON ZAKEA	DK040340
DIMENSION ZAKEA(200)	DK040350
PHI=XHI	DK040360
IF(PHI) 38,39,39	DK040370
38 PHI=PHI+360.0	DK040380
39 IF(JUDGE-359)40,100,40	DK040390
100 IF(NQORT-2)10,10,1000	DK040400
1000 WRITE (6,1001)NQORT	DK040410
1001 FORMAT (1H1,/////////35H YOU WANT A BLOCK OUTPUT FOR NQORT=I3 //	DK040420
1 61H THIS IS NOT POSSIBLE AT THE PRESENT TIME...I SHALL CALL EXIT)	DK040430
CALL EXIT	DK040440
40 K1=1	DK040450
IF(NQORT-2)512,512,513	DK040460
513 NARG=NQORT-2	DK040470
GO TO 514	DK040480
512 NARG = -NQORT	DK040490
514 CALL TALLY(NLINE,2,NARG)	DK040500
IF(NQORT-1)41,61,510	DK040510
510 IF(NQORT-2)51,51,511	DK040520
511 IF(NQORT-3)81,81,91	DK040530
C HEA) ONLY	DK040540
41 WRITE (6,411)PHI,ZEIT,NODE(1),HSUN(1),HALB(1),HPLAN(1)	DK040550
IF(NSATP-2)99,42,42	DK040560
42 DO 43 J=2,NSATP	DK040570
CALL TALLY(NLINE,1,-NQORT)	DK040580
43 WRITE (6,412)NODE(J),HSUN(J),HALB(J),HPLAN(J)	DK040590
411 FORMAT (/F6.1,F8.2,1X13,4F10.2)	DK040600
412 FORMAT(15X13,4F10.2)	DK040610
GO TO 99	DK040620
51 WRITE (6,411)PHI,ZEIT,NODE(1),T(J2,1),HSUN(1),HALB(1),HPLAN(1)	DK040630
IF(NSATP-2)99,52,52	DK040640
52 DO 53 J=2,NSATP	DK040650
CALL TALLY(NLINE,1,-NQORT)	DK040660
53 WRITE (6,412)NODE(J),T(J2,J),HSUN(J),HALB(J),HPLAN(J)	DK040670
GO TO 99	DK040680
61 WRITE (6,411)PHI,ZEIT,NODE(1),T(J2,1)	DK040690
IF(NSATP-2)99,62,62	DK040700
62 DO 63 J=2,NSATP	DK040710
CALL TALLY(NLINE,1,-NQORT)	DK040720
63 WRITE (6,412)NODE(J),T(J2,J)	DK040730
GO TO 99	DK040740


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C          INCIDENT AND ABSORBED HEATS                                DK040750
81 WRITE (6,810)PHI,ZEIT,NODE(1),HSUN(1),HALB(1),HPLAN(1),HASUN(1),HDK040760
   1AALB(1),HAPLN(1),HATOT(1)                                DK040770
810 FORMAT( /F6.1,F8.2,1X13,3F10.2,11X,4F10.2)              DK040780
   IF(NSATP=2)99,82,82                                         DK040790
82 DO 83 J=2,NSATP                                             DK040800
   CALL TALLY(NLINE,1,1)                                       DK040810
83 WRITE (6,830)NODE(J),HSUN(J),HALB(J),HPLAN(J),HASUN(J),HAALB(J),HADK040820
   1PLN(J),HATOT(J)                                           DK040830
830 FORMAT (15X13, 3F10.2,11X,4F10.2)                        DK040840
   GO TO 99                                                    DK040850
C          TEMPERATURE, INCIDENT HEAT, AND ABSORBED HEAT          DK040860
91 WRITE (6,910)PHI,ZEIT,NODE(1),T(J2,1),HSUN(1),HALB(1),HPLAN(1),HADK040870
   1SUN(1),HAALB(1),HAPLN(1),HATOT(1)                        DK040880
910 FORMAT (/F6.1,F8.2,1X13,4F10.2,11X,4F10.2)              DK040890
   IF(NSATP=2)99,92,92                                         DK040900
92 DO 93 J=2,NSATP                                             DK040910
   CALL TALLY(NLINE,1,2)                                       DK040920
93 WRITE (6,930)NODE(J),T(J2,J),HSUN(J),HALB(J),HPLAN(J),HASUN(J),HADK040930
   1ALB(J),HAPLN(J),HATOT(J)                                   DK040940
930 FORMAT (15X13,4F10.2,11X,4F10.2)                          DK040950
   GO TO 99                                                    DK040960
10 L8=(NSATP+9)/10                                           DK040970
   CALL TALLY(NLINE,L8+2,-3)                                   DK040980
   IF(NGORT=1) 11,21,21                                       DK040990
C          ONLY HEAT FLUXES ARE REQUIRED                            DK041000
11 WRITE (6,702)PHI,ZEIT                                     DK041010
12 CALL ARROUT(HSUN(1),L8,NSATP)                               DK041020
   GO TO 31                                                    DK041030
C          TEMPERATURES REQUIRED                                    DK041040
21 WRITE (6,701)PHI,ZEIT                                     DK041050
   DO 23 J=1,NSATP,10                                          DK041060
   NF=MIN0(10*J,NSATP)                                         DK041070
23 WRITE (6,705)J,(1(J2,N),N=J,NF)                            DK041080
705 FORMAT(14X14,2X10F10.2)                                   DK041090
   IF(NGORT=1) 99,99,25                                       DK041100
C          HEAT FLUXES AS WELL AS TEMPERATURES NEEDED           DK041110
25 CALL TALLY(NLINE,L8+2,-3)                                   DK041120
   WRITE (6,706)                                               DK041130
   GO TO 12                                                    DK041140
31 CALL TALLY(NLINE,L8+2,-3)                                   DK041150
32 WRITE (6,703)                                               DK041160
   CALL ARROUT(HALB(1),L8,NSATP)                               DK041170
   CALL TALLY(NLINE,L8+2,-3)                                   DK041180
34 WRITE (6,704)                                               DK041190
   CALL ARROUT(HPLAN(1),L8,NSATP)                              DK041200
701 FORMAT(/F6.1,F9.2,14H TEMPERATURES)                      DK041210
702 FORMAT(/F6.1,F9.2,10H Q SOLAR )                           DK041220
703 FORMAT(/17X8H0 ALBEDO)                                    DK041230
704 FORMAT(/17X8H0 PLANET)                                    DK041240
706 FORMAT(/17X8H0 SOLAR )                                    DK041250
99 RETURN                                                       DK041260
   END                                                         DK041270

      FOR DECK5,DECK5                                DK050000
      SUBROUTINE HEAT(J)                                DK050010
      DIMENSION TRASH(17),F(10,9,42),H1AB(9),ANGTAB(10),WH(18),XH(9),
      1YH(6),ZH(6),ELL(9),RPP(9),RR(9),CAYY(9),TIME(2)      DK050020
      DIMENSION DT(2,200),T(2,200),SINL(200),COSL(200),SINO(200),
      1                                DK050030
      1                                DK050040

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1COS0(200),THICK(200),NCOAT(200),NSUBS(200),COSKS(200),PHIT(200),
2GAMM(200),NDUTY(200)
DIMENSION ESUN(8),EEE(8,42),RO(8,42),SP(8,42),TQINT(41,8)
COMMON TRASH,F,HIAB,ANGTAB,WH,XH,YH,ZH,ELL,RPP,RR,CAYY,PI,PIH,
11WOP1,PI180,NOFIND,NQORT,IFIRST,NEWSIG
COMMON KPLNEI,NOKIEN,KTEMP,NUMRUN,NSATP,NPRINT,KREV,NP50,REV
COMMON A,H,C,AYE,BEE,RP,RN,PEE,EL,R,CAY,BARL,S,ALP2,BF12,GAM2,
1ALPHA2,BETA2,GAMMA2,COSA,COSB,COSG,SINB,PHIMAX
COMMON SIGMA,CSIGMA,SSIGMA,TSIGMA
COMMON PHIZ2,DPHI2,PHI2,DPHI,PHI,CPHI,SPHI,PHIN2,PHOT2,SUN
COMMON TIMEZ,TABS,TELAPS,ZEIT,TIME,DEL TAT,XP,YP,DFE,DPSQ,J1,J2
COMMON EP1P4,EPSIG2,TM,FTM,SAS2,SRASH
COMMON G,RHRCR,RHO,CP,EPSLN,ITK,KITEK
COMMON QNET,QSAT,QINT,QEXT,QPLAN,QALB,QSUN,GOLD,QNEW,TBREAK
COMMON GAM,PHIC,ALF,ALTI,ANGS,CTHET,FD,FF
COMMON PH11,PHI2,ISIG,FUDGE,T4,JUDGE,TPL,RUFFER,RV,NLINE
COMMON COSLS,SINLS,UT,T,SINL,COSL,SINO,COSO,THICK,NCOAT,NSUBS,
1COSRS,PHIT,GAMM
COMMON ESUN,EEE,RO,SP,TQINT,NDUTY,ANINCL,ASCNOD,ASNLCG,RGTASC,
1DECLIN
DIMENSION AA(6),AA1(6),P(6),P1(6)
COMMON AA,AA1,P,P1,IORDER,IOHD1,IERRUR,THETA,DTMAX,EN1,EN,FACT,
1YNHAT,ENHATL,EMAG,DERRUR,DTTEST
COMMON HSUN,HALB,HPLAN,NODE
DIMENSION HSUN(200),HALB(200),HPLAN(200),NODE(200)
COMMON KETCH
COMMON HASUN,HAALB,HAPLN,HATOT
DIMENSION HASUN(200),HAALB(200),HAPLN(200),HATOT(200)
COMMON ZAREA
DIMENSION ZAREA(200)
COMMON IMTHRU
DIMENSION IMTHRU(200)
DIMENSION DARK(200),BRITE(200),NBRITE(200)
1F(J-201)100,101,102
C *** INITIALIZE CALCULATION OF AVG. EPP (PLANET ABSORPTIVITY) TO BE
C OUTPUT ON SC-4020 PLOTS
101 DO 103 I=1,NSATP
1F (IMTHRU(I))103,108,103
108 DARK(I)=0.0
BRITE(I)=0.0
NBRITE(I)=0
103 CONTINUE
GO TO 999
102 IF(J-202)109,109,110
C *** CALCULATE AVG. EPP FOR SUN SIDE
109 DO 104 I=1,NSATP
104 BRITE(I)=BRITE(I)/FLOAT(NBRITE(I))
GO TO 999
C *** SAVE DARK(I) AND BRITE(I) IN TWO ARRAYS WHICH ARE IN COMMON BUT
C NO LONGER USED IN THIS CASE
110 DO 111 I=1,NSATP
HSUN(I) =DARK(I)
111 HALB(I)=BRITE(I)
GO TO 999
100 CONTINUE
JSAT=J
C DETERMINE VALUES NEEDED FOR J TH ELEMENT
GAM=GAMM(JSAT)
PHIC=PHI1(JSAT)
JC=NCOAT(JSAT)

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DK050050
DK050060
DK050070
DK050080
DK050090
DK050100
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DK050460
DK050470
DK050480
DK050490
DK050500
DK050510
DK050520
DK050530
DK050540
DK050550
DK050560
DK050570
DK050580
DK050590
DK050600
DK050610
DK050620
DK050630
DK050640

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	JS=NSUBS(JSAT)	DK050650
	AS=ESUN(JC)	DK050660
	TEMT=T(J1,JSAT)	DK050670
	CALL INTERP(EEE(1,1),TEMT,JC,EPSLN)	DK050680
	CALL INTERP(R0(1,1),TEMT,JS,RH0)	DK050690
	CALL INTERP(SP(1,1),TEMT,JS,CP)	DK050700
	CALL INTERP(EEE(1,1),TPL,JC,EPP)	DK050710
	EPSIG2=-3.428E-09*EPSLN	DK050720
	RHRCP=1.0/(RH0*CP*(HICK(JSAT)*60.0)	DK050730
	KODE=(3.0-SUN)*3.0*SUN	DK050740
	KODE=KODE+3*KTEMP+NORIEN+NORIFN	DK050750
5	GO TO (35,36,33,34,35,36,41,42,33,40,41,42),KODE	DK050760
C	SPINNING,CONSTANT PLANET TEMPERATURE	DK050770
33	FPLAN=0.25*FD	DK050780
	IF(SUN) 331,50,331	DK050790
331	FSUN=0.5	DK050800
	FALB=0.0	DK050810
	IF(CTHET) 60,60,332	DK050820
332	FALB=FD*CTHET	DK050830
	GO TO 60	DK050840
C	SPINNING,SHADE,VARIABLE PLANET TEMPERATURE	DK050850
34	FPLAN=FTM*FD/(S*(1.0-R))	DK050860
	GO TO 50	DK050870
C	ORIENTED,SHADE,CONSTANT PLANET TEMPERATURE	DK050880
35	ANGS=0.0	DK050890
	CALL GEOFAC(FF)	DK050900
	FPLAN=0.5*FF	DK050910
	GO TO 50	DK050920
C	ORIENTED,SHADE,VARIABLE PLANET TEMPERATURE	DK050930
36	ANGS=0.0	DK050940
	CALL GEOFAC(FF)	DK050950
	FPLAN=2.0*FTM*FF/(S*(1.0-R))	DK050960
	GO TO 50	DK050970
C	SPINNING, SUN, VARIABLE PLANET TEMPERATURE	DK050980
40	FPLAN=FD*CTHET	DK050990
	RUB=FTM*FD/(S*(1.0-R))	DK051000
	IF(FPLAN-RUB) 401,331,331	DK051010
401	FPLAN=RUB	DK051020
	GO TO 331	DK051030
C	ORIENTED, SUN, CONSTANT PLANET TEMPERATURE	DK051040
41	STANG=ANGS	DK051050
	ANGS=0.0	DK051060
	CALL GEOFAC(FF)	DK051070
	ANGS=STANG	DK051080
	FPLAN=0.5*FF	DK051090
411	CALL GEOFAC(FF)	DK051100
	FALB=FF+FF	DK051110
	IF(FALB) 412,415,415	DK051120
412	FALB=0.0	DK051130
415	FSUN=2.0*COSKS(JSAI)	DK051140
	IF(FSUN) 416,60,60	DK051150
416	FSUN=0.0	DK051160
	GO TO 60	DK051170
C	ORIENTED, SUN, VARIABLE PLANET TEMPERATURE	DK051180
42	CALL GEOFAC(FF)	DK051190
	FPLAN=FF+FF	DK051200
	FALB=FPLAN	DK051210
	STANG=ANGS	DK051220
	ANGS=0.0	DK051230
	CALL GEOFAC(FF)	DK051240

ANGS=STANG	DK051250
RUB=2.0*FTM*FF/(S*(1.0-R))	DK051260
IF(FPLAN-RUB) 420,420,422	DK051270
420 FPLAN=RUB	DK051280
422 IF(FALB) 412,415,415	DK051290
50 FSUN=0.0	DK051300
49 FALB=0.0	DK051310
60 QSUN=FSUN*SAS2	DK051320
QALB=FALB*SRASH	DK051330
QPLAN=FPLAN*EPTP4	DK051340
QSUN=AMAX1(QSUN,0.0)	DK051350
QALB=AMAX1(QALB,0.0)	DK051360
QPLAN=AMAX1(QPLAN,0.0)	DK051370
QEXT=QSUN+QALB+QPLAN	DK051380
QNET=AS*(QSUN+QALB)+EPP*QPLAN	DK051390
AQSUN=AS*QSUN	DK051400
AQALB=AS*QALB	DK051410
AQPLN=EPP*QPLAN	DK051420
ATOT=QNET	DK051430
IF(ZAREA(JSAT)) 501,501,500	DK051440
500 AQSUN=AQSUN* ZAREA(JSAT)	DK051450
AQALB=AQALB* ZAREA(JSAT)	DK051460
AQPLN=AQPLN* ZAREA(JSAT)	DK051470
ATOT=ATOT* ZAREA(JSAT)	DK051480
501 F4=1.0	DK051490
IF(NORIE) 607,606,607	DK051500
606 F4=4.0	DK051510
607 HSUN(JSAT)=F4*QSUN	DK051520
HALB(JSAT)=F4*QALB	DK051530
HPLAN(JSAT)=F4*QPLAN	DK051540
HASUN(JSAT)=F4*AQSUN	DK051550
HAALB(JSAT)=F4*AQALB	DK051560
HAPLN(JSAT)=F4*AQPLN	DK051570
HAIOT(JSAT)=ATOT*F4	DK051580
IF(NQORT*NQORT -3*NQORT)61,71,61	DK051590
61 CALL QIFIND(JSAT,TElaps-TIMEZ,TBREAK,QNEW,QOLD)	DK051600
71 GO TO 99	DK051610
99 IF(NQORT -3)999,105,105	DK051620
105 IF (SUN-1.0)106,107,107	DK051630
C *** EPP FOR SHADE SIDE (CONSTANT)	DK051640
106 LAKK(JSAT) =EPP	DK051650
GO TO 999	DK051660
C *** ACCUMULATE EPP FOR SUN SIDE	DK051670
107 NBRITE(JSAT) =NBRITE(JSAT) +1	DK051680
BRITE (JSAT) =BRITE(JSAT) +EPP	DK051690
999 RETURN	DK051700
END	DK051710
FOR DECK6,DECK6	DK060000
SUBROUTINE FREAD	DK060010
DIMENSION TRASH(17),F(10,9,42),HTAB(9),ANGTAB(10),WH(18),XH(9),	DK060020
1YH(6),ZH(6),ELL(9),RPP(9),RR(9),CAYY(9),TIME(2)	DK060030
DIMENSION DT(2,200),T(2,200),SINL(200),COSL(200),SINO(200),	DK060040
1COS0(200),THICK(200),NCOAT(200),NSURS(200),COSRS(200),PHIT(200),	DK060050
2GAMM(200),NDUTY(200)	DK060060
DIMENSION ESUN(8),EEE(8,42),RO(8,42),SP(8,42),TQINT(41,8)	DK060070
DIMENSION BUFFER(10)	DK060080
COMMON TRASH,F,HTAB,ANGTAB,WH,XH,YH,ZH,ELL,RPP,RR,CAYY,PI,PIH,	DK060090
11WUPI,PI180,NOFIND,NQORT,IFIRST,NEWSIG	DK060100

COMMON KPLNET,NOKIEN,KTEMP,NUMRUN,NSATP,NPRINT,KREV,NP50,REV	DK060110
COMMON A,B,C,AYE,BEE,RP,RN,PEE,EL,R,CAY,BAKL,S,ALP2,BET2,GAM2,	DK060120
1ALPHA2,BETA2,GAMMA2,COSA,COSB,COSG,SINB,PHIMAX	DK060130
COMMON SIGMA,CSIGMA,SSIGMA,TSIGMA	DK060140
COMMON PH12,DPH12,PH1Z,DPHI,PHI,CPH1,SPHI,PHIN2,PHOT2,SUN	DK060150
COMMON TIMEZ,TARS,IELAPS,ZEIT,TIME,DELTAT,XP,YP,UEE,UPSG,J1,J2	DK060160
COMMON EPTP4,EPSIG2,TM,FTM,SAS2,SRASH	DK060170
COMMON G,KHRCP,RHO,CP,EPSLN,ITK,KITEK	DK060180
COMMON QNET,QSAT,QINT,QEXT,QPLAN,QALB,QSUN,GOLD,QNEW,TBREAK	DK060190
COMMON GAM,PHIC,ALI,ALTI,ANGS,CTHET,FD,FF	DK060200
COMMON BUFEK	DK060210
COMMON COSLS,SINLS,DT,T,SINL,COSL,SINO,COSU,THICK,NCOAT,NSUBS,	DK060220
1COSRS,PHIT,GAMM	DK060230
COMMON ESUN,EEE,R0,SP,TQINT,NDUTY,ANINCL,ASCNOD,ASNLNG,RGTASC,	DK060240
1DECLIN	DK060250
DIMENSION Z(13)	DK060260
HTAB(1)=-1.E20	DK060270
HTAB(2)=100.0	DK060280
HTAB(3)=300.0	DK060290
HTAB(4)=600.0	DK060300
HTAB(5)=1000.0	DK060310
HTAB(6)=3000.0	DK060320
HTAB(7)=6000.0	DK060330
HTAB(8)=10000.0	DK060340
HTAB(9)=20000.0	DK060350
ANGTAB(1)=0.0	DK060360
ANGTAB(2)=20.0	DK060370
ANGTAB(3)=30.0	DK060380
ANGTAB(4)=40.0	DK060390
ANGTAB(5)=50.0	DK060400
ANGTAB(6)=60.0	DK060410
ANGTAB(7)=70.0	DK060420
ANGTAB(8)=80.0	DK060430
ANGTAB(9)=85.0	DK060440
ANGTAB(10)=90.0	DK060450
REAL(5,600) ((F(J,K,1),J=1,10),K=2,9)	DK060460
REAL(5,600) (((F(J,K,L),J=1,10),K=2,9),L=8,42)	DK060470
600 FORMAT(20F4,4)	DK060480
DO 47 L=2,7	DK060490
DO 47 K=2,9	DK060500
DO 47 J=1,10	DK060510
47 F(J,K,L)=F(J,K,1)	DK060520
C CONSTANTS WHICH ARE DEFINED ONLY FIRST TIME AROUND	DK060530
PI=3.1415927	DK060540
TWUPI=6.2831853	DK060550
PIH=1.5707963	DK060560
PI180=.017453293	DK060570
BAKL=48.89E10	DK060580
ELL(1)=48.89E10	DK060590
ELL(2)=48.89E10	DK060600
ELL(3)=255.3E10	DK060610
ELL(4)=74.81E10	DK060620
ELL(5)=19.03E10	DK060630
ELL(6)=1475.E10	DK060640
ELL(7)=467.9E10	DK060650
ELL(8)=941.3E10	DK060660
ELL(9)=35.43E10	DK060670
CAYY(1)=141.E14	DK060680
CAYY(2)=1.731E14	DK060690
CAYY(3)=44900.E14	DK060700

CAYY(4)=15.20E14	DK060710
CAYY(5)=7.66E14	DK060720
CAYY(6)=2435.E14	DK060730
CAYY(7)=13450.E14	DK060740
CAYY(8)=2058.E14	DK060750
CAYY(9)=114.8E14	DK060760
KPP(1)=20.9E6	DK060770
KPP(2)=5.702E6	DK060780
KPP(3)=229.3E6	DK060790
KPP(4)=10.87E6	DK060800
KPP(5)=8.151E6	DK060810
KPP(6)=81.51E6	DK060820
KPP(7)=188.7E6	DK060830
KPP(8)=83.6E6	DK060840
KPP(9)=20.34E6	DK060850
C *** EARTH ALBEDO CHANGED APR. 1966 (WAS=.39)	DK060860
KK(1)=.35	DK060870
KK(2)=.047	DK060880
KK(3)=.51	DK060890
KK(4)=.148	DK060900
KK(5)=.058	DK060910
KK(6)=.62	DK060920
KK(7)=.50	DK060930
KK(8)=.66	DK060940
KK(9)=.76	DK060950
DO 14 J=1,200	DK060960
KCOAT(J)=1	DK060970
KSUBS(J)=1	DK060980
NDUTY(J)=1	DK060990
THICK(J)=.01	DK061000
SIN0(J)=0.0	DK061010
COS0(J)=1.0	DK061020
SINL(J)=0.0	DK061030
COSL(J)=1.0	DK061040
14 GAMM(J)=90.0	DK061050
C READ IN HEADING INFORMATION	DK061060
READ(5,641) (ZH(J),J=1,6),(YH(J),J=1,6),(XH(J),J=1,9),(WH(J),J=1,18)	DK061070
641 FORMAT(13A6)	DK061080
C READ COMMENT CARDS	DK061090
30 READ(5,405) KK,(Z(J),J=1,13)	DK061100
405 FORMAT(I2,13A6)	DK061110
IF(9-KK)37,37,38	DK061120
37 WRITE(6,406) (Z(J),J=1,13)	DK061130
406 FORMAT(/1X13A6)	DK061140
GO TO 30	DK061150
C READ IN TABLES OF MATERIAL PROPERTIES	DK061160
38 CONTINUE	DK061170
300 READ(30,400) KCOAT,KSUBS	DK061180
400 FORMAT(2I2)	DK061190
C THERE ARE KCOAT COATING TABLES, KSUBS SUBSTRATE TABLES	DK061200
DO 301 M=1,KCOAT	DK061210
CALL TABLE(EEE(1,1),1,M)	DK061220
301 ESUN(M)=EEE(M,42)	DK061230
DO 302 M=1,KSUBS	DK061240
CALL TABLE(SP(1,1),0,M)	DK061250
302 CALL TABLE(RU(1,1),0,M)	DK061260
99 RETURN	DK061270
END	DK061280
	DK061290

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      FOR      DECK7,DECK7
      SUBROUTINE QIFIND(JSAT,TIMD,TB,QN,QA)
      DIMENSION TRASH(17),F(10,9,42),HIAB(9),ANGTAB(10),WH(18),XH(9),
      1YH(6),ZH(6),ELL(9),RPP(9),RR(9),CAYY(9),TIME(2)
      DIMENSION DT(2,200),T(2,200),SINL(200),COSL(200),SINO(200),
      1COS0(200),THICK(200),NCOAT(200),NSUBS(200),COSRS(200),PHIT(200),
      2GAMM(200),NDUTY(200)
      DIMENSION ESUN(8),EEE(8,42),RO(8,42),SP(8,42),TQINT(41,8)
      DIMENSION BUFFER(10)
      COMMON TRASH,F,HIAB,ANGTAB,WH,XH,YH,ZH,ELL,RPP,RR,CAYY,PI,PIH,
      1TWOPI,PI180,NOFIND,NGORT,IFIRST,NEWSIG
      COMMON KPLNE1,NOKIEN,KTEMP,NUMRUN,NSATP,NPRINT,KREV,NP50,REV
      COMMON A,B,C,AYE,BEE,RP,RN,PEE,EL,R,CAY,BARL,S,ALP2,BET2,GAM2,
      1ALPHA2,BETA2,GAMMA2,COSA,COSB,COSG,SINB,PHIMAX
      COMMON SIGMA,CSIGMA,SSIGMA,TSIGMA
      COMMON PHIZ2,DPHI2,PHI2,DPHI,PHI,CPH1,SPHI,PHIN2,PHOT2,SUN
      COMMON TIMEZ,TABS,TELAPS,ZEIT,TIME,DELTAT,XP,YP,DEE,DPSQ,J1,J2
      COMMON EPTP4,EPSIG2,TM,FTM,SAS2,SRASH
      COMMON G,RHRCP,RHO,CP,EPSLN,ITK,KITER
      COMMON QNET,QSAT,QINT,QEXT,QPLAN,QALB,QSUN,GOLD,QNEW,TBREAK
      COMMON GAM,PHIC,ALT,ALTI,ANGS,CTHET,FD,FF
      COMMON BUFFER
      COMMON COSLS,SINLS,DT,T,SINL,COSL,SINO,COS0,THICK,NCOAT,NSUBS,
      1COSRS,PHIT,GAMM
      COMMON ESUN,EEE,RO,SP,TQINT,NDUTY,ANINCL,ASCNOD,ASNLNG,RGTASC,
      1DECLIN
      DIMENSION AA(6),AA1(6),P(6),P1(6)
      COMMON AA,AA1,P,P1,IORDER,IORD1,IEKROR,THETA,DTMAX,EN1,EN,FACT,
      1YNHAT,ENHATL,EMAG,DError,DTTEST
      COMMON HSUN,HALB,HPLAN,NODE
      DIMENSION HSUN(200),HALB(200),HPLAN(200),NODE(200)
      COMMON KETCH
      COMMON HASUN,HAALB,HAPLN,HATOT
      DIMENSION HASUN(200),HAALB(200),HAPLN(200),HATOT(200)
      COMMON ZAREA
      DIMENSION ZAREA(200)
      COMMON IMTHRU
      DIMENSION IMTHRU(200)
      COMMON IMHI
      N=NDUTY(JSAT)
      TID=AMOD(TIMD,PEE)
      IF(TID) 92,94,95
      92 TID=TID+PEE
      C      FIND TIME AT START OF INTERVAL
      95 TA=TID-ABS(TIME(J1)-TIME(J2))
      K=IQINT(41,N)
      C      IF TQINT(41)=0, SET QINT=0.
      C      IF TQINT(41) IS 2., 4.,... USE THIS AS INDEX FOR TESTS
      IF(K) 96,96,5
      96 GOLD=0.0
      97 QNEW=QOLD
      C      NO CHANGE THIS TIME
      TBREAK=TID
      99 TB=TBREAK
      QN=QNEW
      QA=QOLD
      100 IF(IMHI -JSA1)101,101,105
      C      ADVANCE INTERNAL HEAT TABLE INDEX AFTER PROCESSING LAST ELEMENT
      101 DO 104 J=1,8

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	N=IQINT(41,J)	DK070590
	IF(N)104,104,102	DK070600
102	IF(TQINT(N,J)-TID) 103,104,104	DK070610
103	TQINT(41,J)=TQINT(41,J)+2.0	DK070620
	N=IQINT(41,J)	DK070630
	GO TO 102	DK070640
104	CONTINUE	DK070650
105	RETURN	DK070660
5	IF(TID-TQINT(K,N)) 6,6,7	DK070670
6	QOLD=TQINT(K-1,N)	DK070680
	GO TO 97	DK070690
7	TBREAK=TQINT(K,N)	DK070700
C	QINT CHANGES THIS TIME	DK070710
	QOLD=TQINT(K-1,N)	DK070720
	QNEW=TQINT(K+1,N)	DK070730
71	IF(K-38) 98,8,8	DK070740
98	K=K+2	DK070750
	IF(TID-TQINT(K,N)) 99,99,991	DK070760
C	MORE THAN ONE CHANGE THIS INTERVAL	DK070770
C	FIND AVERAGE QOLD	DK070780
991	QOLD=QNEW*(TQINT(K,N)-TQINT(K-2,N))+QOLD*(TQINT(K-2,N)-TA)	DK070790
	QNEW=TQINT(K+1,N)	DK070800
	GO TO 71	DK070810
8	TQINT(40,N)=100000000.0	DK070820
	GO TO 99	DK070830
	END	DK070840
	FOR DECK8,DECK8	DK080000
	SUBROUTINE TALLY(N,ND,NQ)	DK080010
	DIMENSION TRASH(17),F(10,9,42),HIAB(9),ANGTAB(10),WH(18),XH(9),	DK080020
	1YH(6),ZH(6),ELL(9),RPP(9),RR(9),CAYY(9),TIME(2)	DK080030
	DIMENSION DT(2,200),T(2,200),SINL(200),COSL(200),SINO(200),	DK080040
	1COS0(200),THICK(200),NCOAT(200),NSURS(200),COSRS(200),PHIT(200),	DK080050
	2GAMM(200),NDUTY(200)	DK080060
	DIMENSION ESUN(8),EEE(8,42),RO(8,42),SP(8,42),TQINT(41,8)	DK080070
	DIMENSION BUFFER(2)	DK080080
	COMMON TRASH,F,HIAB,ANGTAB,WH,XH,YH,ZH,ELL,RPP,RR,CAYY,PI,PIH,	DK080090
	1TWOP1,PI180,NOFIND,NQORT,IFIRST,NEWS16	DK080100
	COMMON KPLNE1,NOKIEN,KTEMP,NUMRUN,NSATP,NPRINT,KREY,NP50,REV	DK080110
	COMMON A,B,C,AYF,BEE,RP,RR,PEE,EL,R,CAY,BARL,S,ALP2,BFT2,GAM2,	DK080120
	1ALPHA2,BETA2,GAMMA2,COSA,COSB,COSG,SINB,PHIMAX	DK080130
	COMMON SIGMA,CSIGMA,SSIGMA,TSIGMA	DK080140
	COMMON PHIZ2,DPH12,PHIZ,DPHI,XHI,CPH1,SPHI,PHIN2,PHOT2,SUN	DK080150
	COMMON TIMEZ,TABS,IELAPS,ZEII,TIME,DEL TAT,XP,YP,DEE,DPSQ,J1,J2	DK080160
	COMMON EPTP4,EPS1G2,IM,FTM,SAS2,SRASH	DK080170
	COMMON G,RHRLP,RHO,CP,EPSLN,ITK,KITER	DK080180
	COMMON QNET,QSAT,QINT,QEXT,QPLAN,QALB,QSUN,QOLD,QNEW,TBREAK	DK080190
	COMMON GAM,PHIC,ALI,ALTI,ANGS,CTHET,FD,FF	DK080200
	COMMON PH11,PH12,ISIG,FUDGE,TPL,JUDGE,BUFFER,RV,NLINE	DK080210
	COMMON COSLS,SINLS,DT,T,SINL,COSL,SINO,COS0,THICK,NCOAT,NSURS,	DK080220
	1COSRS,PH11,GAMM	DK080230
	COMMON ESUN,EEE,RO,SP,TQINT,NDUTY,ANINCL,ASCNOD,ASNLNG,RTASC,	DK080240
	1UECLIN	DK080250
	DIMENSION AA(6),AA1(6),P(6),P1(6)	DK080260
	COMMON AA,AA1,P,P1,IORDER,IORD1,1EKROP,THEJA,DIIMAX,FN1,FN,FACT,	DK080270
	1YNHAT,ENHATL,FMAG,UEERROR,DTTEST	DK080280
	COMMON HSUN,HALR,HPLAN,NODE	DK080290
	DIMENSION HSUN(200),HALB(200),HPLAN(200),NODE(200)	DK080300
	COMMON KETCH	DK080310
	NC=NQ+4	DK080320
	IF(ND) 9,9,8	DK080330

	2GAMM(200)	DK090060
	DIMENSION ESUN(8),EEE(8,42),RO(8,42),SP(8,42),TQINT(41,8)	DK090070
	COMMON TRASH,F,H,AB,ANGTAB,WH,XH,YH,ZH,ELL,RPP,RR,CAYY,PI,PIH,	DK090080
	1TWUPI,PI180,NOFIND,NGORT,IFIRST,NEWSIG	DK090090
	COMMON KPLNET,NORIEN,KTEMP,NUMRUN,NSATP,NPRINT,KREV,NP50,REV	DK090100
	COMMON A,B,C,AYE,BEE,RP,RN,PEE,EL,R,CAY,BARL,S,ALP2,BFT2,GAM2,	DK090110
	1ALPHA2,BETA2,GAMMA2,COSA,COSB,COSG,SINB,PHIMAX	DK090120
	COMMON SIGMA,CSIGMA,SSIGMA,TSIGMA	DK090130
	COMMON PHIZ2,DPH12,PHIZ,DPHI,PHI,CPH1,SPHI,PHIN2,PHOT2,SUN	DK090140
	COMMON TIMEZ,TABS,TELAPS,ZEIT,TIME,DELTAT,XP,YP,DEE,UPSQ,J1,J2	DK090150
	COMMON EPTP4,EP5IG2,TM,FTM,SAS2,SRASH	DK090160
	COMMON G,RHRC,PC,RHO,CP,EP5LN,ITK,KITEK	DK090170
	COMMON QNET,QSAT,QINT,QEXT,QPLAN,QALB,QSUN,QOLD,QNEW,TBREAK	DK090180
	COMMON GAM,PHIC,ALT,ALTI,ANGS,CTHET,FD,FF	DK090190
	COMMON PH11,PH12,ISIG,FUDGE,T4,JUDGE,TPL,BUFFER,RV,NLINE	DK090200
	COMMON COSLS,SINLS,DT,T,SINL,COSL,SINO,COSU,THICK,NCUAT,NSUBS,	DK090210
	1COSRS,PHIT,GAMM	DK090220
	COMMON ESUN,EEE,RO,SP,TQINT,NDUTY,ANINCL,ASCNOD,ASNLNG,RGTASC,	DK090230
	1DECLIN	DK090240
	DIMENSION AA(6),AA1(6),P(6),P1(6)	DK090250
	COMMON AA,AA1,P,P1,IORDER,IORD1,IEKRUR,THETA,DTMAX,EN1,EN,FACT,	DK090260
	1YNHAT,ENHATL,EMAG,UEX,DTTEST	DK090270
	COMMON HSUN,HALB,HPLAN,NODE	DK090280
	DIMENSION HSUN(200),HALB(200),HPLAN(200),NODE(200)	DK090290
C	FINU TIME AS A FUNCTION OF PHI. ALSO FIND X AND Y	DK090300
	PHIH=0.5*PI180*PHI	DK090310
	PHIR=PHIH+PHIH	DK090320
	TPHIH=SIN(PHIH)/COS(PHIH)	DK090330
	ATP=(A-C)*TPHIH/B	DK090340
	EE=ATAN(ATP)*2.0	DK090350
	IF(EE) 8,8,9	DK090360
8	EE=EE+TWUPI	DK090370
9	SINEE=SIN(EE)	DK090380
	TELAPS=RN*(EE-C*SINEE/A)	DK090390
10	XP=COS(EE)*A -C	DK090400
	YP=SINEE*B	DK090410
C	FIND USEFUL QUANTITIES DEPENDING ONLY ON SATELLITE LOCATION	DK090420
	UPSQ=XP*XP+YP*YP	DK090430
	UEE=SQRT(UPSQ)	DK090440
	FD=1.0-SQRT(1.0-RP/UPSQ*RP)	DK090450
	CTHET=(XP*COSA+YP*COSG)/DEE	DK090460
	TPL=T4	DK090470
	IF(KTEMP) 208,208,201	DK090480
201	IF(CTHET) 205,205,202	DK090490
202	TPL=T4*SQRT(SQRT(4.0*CTHET))	DK090500
	IF(TPL -1M) 205,208,208	DK090510
205	TPL =TM	DK090520
208	AL1I=DEE-KP	DK090530
	CPHI=COS(PHIR)	DK090540
	SPHI=SIN(PHIR)	DK090550
C	IF SATELLITE IS ORIENTED, FIND PHIC,COSRS,SIN AND COS LAMDA	DK090560
	IF(NORIEN) 281,3,281	DK090570
281	SINLS=SPHI*CSIGMA-CPHI*SSIGMA	DK090580
	COSLS=-SPHI*SSIGMA-CPHI*CSIGMA	DK090590
	ANGS=ARCOS(CTHET)	DK090600
	AL1=3441.0*ALTI/RP	DK090610
	IF(NORIEN) 283,283,282	DK090620
282	CFLS=1.0	DK090630
	SFLS=0.0	DK090640
	FLSC=COSLS	DK090650

	FLSS=SINLS	DK090660
	GO TO 2831	DK090670
283	CFLS=COSLS	DK090680
	SFLS=SINLS	DK090690
	FLSC=1.0	DK090700
	FLSS=0.0	DK090710
2831	DO 287 K=1,NSATP	DK090720
	IF(NORLEN) 2832,2838,2838	DK090730
2832	COSX=COSLS*COSL(K)-SINLS*SINL(K)	DK090740
	GAMM(K)=ARCOS(COSX)	DK090750
2838	RNUM=COSL(K)*SFLS+SINL(K)*CFLS	DK090760
	DEN=((SINB*SINLS)**2+COSB**2)*(COSU(K)**2+RNUM**2)	DK090770
	RNUM=SINLS*SINB*RNUM+COSB*COSU(K)	DK090780
	COSX=RNUM/SQRT(DEN)	DK090790
	IF(COSX=1.0) 285,284,284	DK090800
284	PHIT(K)=0.0	DK090810
	GO TO 2861	DK090820
285	IF(COSX=1.0) 2850,2850,286	DK090830
2850	PHIT(K)=180.0	DK090840
	GO TO 2861	DK090850
286	PHIT(K)=ARCOS(COSX)	DK090860
2861	IF(NORLEN) 287,287,2862	DK090870
2862	COSRS(K)=SINB*(COSL(K)*FLSC+SINL(K)*FLSS)+COSB*COSU(K)	DK090880
287	CONTINUE	DK090890
3	TABS=TELAPS-TIMEZ+(RV-1.0)*PEE	DK090900
	IF(TABS) 98,98,99	DK090910
98	TABS=TABS+PEE	DK090920
99	RETURN	DK090930
	END	DK090940
	FOR DECK10,DECK10	DK100000
	SUBROUTINE TEMPER(JSAT)	DK100010
	DIMENSION TRASH(17),F(10,9,42),H1AB(9),ANGTAB(10),WH(18),XH(9),	DK100020
	1YH(6),ZH(6),ELL(9),RPP(9),RR(9),CAYY(9),TIME(2)	DK100030
	DIMENSION DT(2,200),T(2,200),SINL(200),COSL(200),SINO(200),	DK100040
	1COSU(200),THICK(200),NCOAT(200),NSURS(200),COSRS(200),PHIT(200),	DK100050
	2GAMM(200),NDUTY(200)	DK100060
	DIMENSION ESUN(8),EEE(8,42),RO(8,42),SP(8,42),TGINT(41,8)	DK100070
	DIMENSION BUFFER(8)	DK100080
	COMMON TRASH,F,H1AB,ANGTAB,WH,XH,YH,ZH,ELL,RPP,RR,CAYY,PI,PIH,	DK100090
	11WUPI,PI180,NOFIND,NQORT,IFIRST,NEWSIG	DK100100
	COMMON KPLNET,NORLEN,KTEMP,NUMRUN,NSATP,NPRINT,KRFV,NP50,REV	DK100110
	COMMON A,H,C,AYF,BEE,RP,RN,PEE,EL,R,CAY,BARL,S,ALP2,BFT2,GAM2,	DK100120
	1ALPHA2,BETA2,GAMMA2,COSA,COSB,COSG,SINB,PHIMAX	DK100130
	COMMON SIGMA,CSIGMA,SSIGMA,TSIGMA	DK100140
	COMMON PHIZ2,DPH12,PHIZ,DPHI,PHI,CPH1,SPH1,PHIN2,PHOT2,SUN	DK100150
	COMMON TIMEZ,TABS,TELAPS,ZEIT,TIME,DELTAT,XP,YP,DEE,DPSQ,J1,J2	DK100160
	COMMON EPTP4,EPSIG2,TM,FTM,SAS2,SRASH	DK100170
	COMMON G,KHRCR,RHO,CP,EPSLN,ITK,KITEK	DK100180
	COMMON QNET,QSAT,QINT,QEXT,QPLAN,QALB,QSUN,QOLD,QNEW,TBREAK	DK100190
	COMMON GAM,PHIC,ALI,AL1I,ANGS,CTHET,FD,FF	DK100200
	COMMON PH11,PHI2,BUFFER	DK100210
	COMMON COSLS,SINLS,DT,T,SINL,COSL,SINO,COSU,THICK,NCOAT,NSURS,	DK100220
	1COSRS,PH11,GAMM	DK100230
	COMMON ESUN,EEE,RO,SP,TGINT,NDUTY,ANINCL,ASCNOD,ASNLCG,RGTASC,	DK100240
	1UECLIN	DK100250
	DIMENSION AA(6),AA1(6),P(6),P1(6)	DK100260
	COMMON AA,AA1,P,P1,IORUER,IORU1,1EKRUR,THEJA,DTMAX,EN1,EN,FACT,	DK100270
	1YNHAT,ENHATL,EMAG,UEKRUR,DTTEST	DK100280

J=JSAT	DK100290
DELTAT=TIME(J2)-TIME(J1)	DK100300
TIME1 = TIME(J1)	DK100310
HORD=DTMAX**10ORDER	DK100320
CALL HEAT(J)	DK100330
C NEW LOGIC	DK100340
IFLAG = 1	DK100350
IF(QOLD-QNEW)60,50,60	DK100360
60 QINT = QOLD	DK100370
TLEFT=TIME(J2)-TBREAK	DK100380
C **** THE FOLLOWING CARD WAS ADDED 2/4/65 AT THE REQUEST OF MRI ****	DK100390
TLEFT=AMOD(TLEFT,PEE)	DK100400
DELTAT=DELTAT-TLEFT	DK100410
DELTAT=AMOD(DELTAT,PEE)	DK100420
IFLAG = 2	DK100430
GO TO 51	DK100440
61 IFLAG = 1	DK100450
TIME1 = TIME1 + DELTAT	DK100460
DELTAT = TLEFT	DK100470
DELTAT=AMOD(DELTAT,PEE)	DK100480
T(J1,J) = TEMP	DK100490
50 QINT = QNEW	DK100500
C BEGIN OLD LOGIC	DK100510
51 TEMP=T(J1,J)	DK100520
C10=10.0	DK100530
Q2=SQRT(QNET+QINT1)	DK100540
Q4=SQRT(Q2)	DK100550
PF2=SQRT(-0.5*EPSIG2)	DK100560
PF4=SQRT(PF2)	DK100570
UT1=C10/(RHRCP*(QNET+QINT+0.5*EPSIG2*TEMP**4))	DK100580
UT1=ABS(UT1)	DK100590
UT2=1.0/((Q2+PF2*TEMP**2)*(Q4+PF4*TEMP))	DK100600
DTTEST=AMIN1(UT1,UT2)	DK100610
IF(DTTES)52,52,53	DK100620
52 DTTEST=.00001*PEE	DK100630
53 ISTEPS=DELTAT/DTTEST+.999	DK100640
IF(ISTEPS)100,100,101	DK100650
100 ISTEPS=1	DK100660
101 STEPS=ISTEPS	DK100670
STEP=DELTAT/STEPS	DK100680
THETA=STEP/DTMAX	DK100690
TEMP=T(J1,J)	DK100700
TIME2 = TIME1	DK100710
DO 30 I = 1,ISTEPS	DK100720
F0 = PHIFN(TIME2,TEMP,STEP)	DK100730
TEMP=TEMP+STEP*FU	DK100740
GO TO (1,1,3),IERROR	DK100750
1 CALL FRRUK(STEP,F0)	DK100760
EN1=EN*HORD	DK100770
DERROR=(EN-EMAG)*HORD	DK100780
EMAG=EN	DK100790
WRITE (6,601)EN1,DERROR,TEMP	DK100800
601 FORMAT(5H0EN1=1PE14.7,5H DE=E14.7,7H TEMP=E14.7)	DK100810
3 CONTINUE	DK100820
30 TIME2 = TIME2 + STEP	DK100830
C NEW LOGIC	DK100840
GO TO (62,61),IFLAG	DK100850
C BEGIN OLD LOGIC	DK100860
62 T(J2,J) = TEMP	DK100870
11 RETURN	DK100880

END	DK100890
FOR DECK11,DECK11	DK110000
SUBROUTINE INIT	DK110010
DIMENSION TRASH(17),F(10,9,42),HTAB(9),ANGTAB(10),WH(18),XH(9),	DK110020
1YH(6),ZH(6),ELL(9),RPP(9),RR(9),CAYY(9),TIME(2)	DK110030
DIMENSION DT(2,200),T(2,200),SINL(200),COSL(200),SINO(200),	DK110040
1COS0(200),THICK(200),NCOAT(200),NSURS(200),COSRS(200),PHIT(200),	DK110050
2GAMM(200),NDUTY(200)	DK110060
DIMENSION ESUN(8),EEE(8,42),RO(8,42),SP(8,42),TGINT(41,8)	DK110070
DIMENSION BUFFER(6)	DK110080
COMMON TRASH,F,HTAB,ANGTAB,WH,XH,YH,ZH,ELL,RPP,RR,CAYY,PI,PIH,	DK110090
1TWOPI,PI180,NOFIND,NGORT,IFIRST,NEWSIG	DK110100
COMMON KPLNET,NOKIEN,KTEMP,NUMRUN,NSATP,NPRINT,KREV,NP50,REV	DK110110
COMMON A,B,C,AYE,BEE,RP,RN,PEE,EL,R,CAY,BARL,S,ALP2,BET2,GAM2,	DK110120
1ALPHA2,BETA2,GAMMA2,COSA,COSB,COSG,SINB,PHIMAX	DK110130
COMMON SIGMA,CSIGMA,SSIGMA,TSIGMA	DK110140
COMMON PHIZ2,DPHI2,PHIZ,DPHI,PHI,CPHI,SPHI,PHIN2,PHOT2,SUN	DK110150
COMMON TIMEZ,TABS,TELAPS,ZEIT,TIME,DELTAT,XP,YP,DEE,DPSQ,J1,J2	DK110160
COMMON EPIP4,EPSIG2,TM,FTM,SAS2,SRASH	DK110170
COMMON G,KHRLP,RHO,CP,EPSLN,ITK,KITEK	DK110180
COMMON QNET,QSAT,QINT,QEXT,QPLAN,QALB,QSUN,QOLD,QNEW,TBREAK	DK110190
COMMON GAM,PHIC,ALT,ALTI,ANGS,CTHEI,FD,FF	DK110200
COMMON PH11,PHI2,BUFFER,RV,NLINE	DK110210
COMMON COSLS,SINLS,UT,T,SINL,COSL,SINO,COS0,THICK,NCOAT,NSURS,	DK110220
1COSRS,PHIT,GAMM	DK110230
COMMON ESUN,EEE,RO,SP,TGINT,NDUTY	DK110240
C DETERMINE REFERENCE TIME	DK110250
PHI=PHIZ2	DK110260
KV=1.0	DK110270
CALL LUCUS	DK110280
TIMEZ=TIMEZ+1ABS	DK110290
IF(SUN=2.0) 89,80,80	DK110300
80 PHIN2=-10.0	DK110310
PHOT2=-10.0	DK110320
GO TO 100	DK110330
89 IF(NOFIND) 90,90,92	DK110340
C SET UP SUN-SHADE TESTS FOR FIRST ORBIT	DK110350
90 IF (PHI1) 1,2,2	DK110360
1 PHI1=PHI1+TWOPI	DK110370
2 IF (PHI2) 3,4,4	DK110380
3 PHI2=PHI2+TWOPI	DK110390
4 DIF=PHI1-PHI2	DK110400
IF (ABS(DIF)-PI) 8,8,5	DK110410
5 IF (DIF) 6,9,9	DK110420
6 PHI2=PHI2-TWOPI	DK110430
C INTERCHANGE PHI1 AND PHI2	DK110440
7 RUB=PHI1	DK110450
PHI1=PHI2	DK110460
PHI2=RUB	DK110470
GO TO 10	DK110480
8 IF (DIF) 10,10,7	DK110490
9 PHI1=PHI1-TWOPI	DK110500
10 I1=PHI1-PHI2	DK110510
I2=PHI2-PHI2	DK110520
IF (I1) 11,11,13	DK110530
11 IF (I2) 12,12,97	DK110540
12 I1=I1+TWOPI	DK110550
I2=I2+TWOPI	DK110560

13	IF (T1) 14,14,13	DK110570
	IF (T2) 97,97,98	DK110580
14	IF (T2) 98,98,97	DK110590
C	ORBIT BEGINS IN SHADE	DK110600
97	SUN=0.0	DK110610
	GO TO 99	DK110620
C	ORBIT BEGINS IN SUN	DK110630
98	SUN=1.0	DK110640
99	PHI1=PHI2+T1	DK110650
	PHI2=PHI2+T2	DK110660
	PHI12=PHI1/P1180	DK110670
	PHI22=PHI2/P1180	DK110680
100	RETURN	DK110690
92	SUN=1.0	DK110700
	IF ((PHOT2-PHI22)*(PHI22-PHI12)) 100,100,93	DK110710
93	SUN=0.0	DK110720
	GO TO 100	DK110730
	END	DK110740
	FOR DECK12,DECK12	DK120000
	SUBROUTINE FIND	DK120010
	DIMENSION X(3),Y(3),U(4),V(4)	DK120020
	EQUIVALENCE(X(1),TRASH(1)),(Y(1),TRASH(4)),(U(1),TRASH(7))	DK120030
	EQUIVALENCE(V(1),TRASH(11)),(C1,TRASH(15)),(C2,TRASH(16))	DK120040
	EQUIVALENCE(C3,TRASH(17))	DK120050
	DIMENSION TRASH(17),F(10,9,42),HTAB(9),ANGTAB(10),WH(18),XH(9),	DK120060
	YH(6),ZH(6),ELL(9),RPP(9),RR(9),CAY(9),TIME(2)	DK120070
	DIMENSION DT(2,200),T(2,200),SINL(200),COSL(200),SINO(200),	DK120080
	ICOSO(200),THICK(200),NCOAT(200),NSUBS(200),COSKS(200),PHIT(200),	DK120090
	2GAMM(200),NDUTY(200)	DK120100
	DIMENSION ESUN(8),EEE(8,42),RO(8,42),SP(8,42),TQINT(41,8)	DK120110
	DIMENSION BUFFER(8)	DK120120
	COMMON TRASH,F,HTAB,ANGTAB,WH,XH,YH,ZH,ELL,RPP,RR,CAY,PI,PIH,	DK120130
	11WUPI,PI180,NOFINO,NOORT,IFIRST,NEWSIG	DK120140
	COMMON KPLNE1,NORIEN,KTEMP,NUMRUN,NSATP,NPRINT,KREV,NP50,REV	DK120150
	COMMON A,H,C,AYE,BEE,RF,RN,PEE,EL,R,CAY,BARL,S,ALP2,BET2,GAM2,	DK120160
	1ALPHA2,BETA2,GAMMA2,COSA,COSB,COSG,SINB,PHIMAX	DK120170
	COMMON SIGMA,CSIGMA,SSIGMA,TSIGMA	DK120180
	COMMON PHI22,DPHI2,PHI2,DPHI,PHI,CPHI,SPHI,PHI12,PHOT2,SUN	DK120190
	COMMON TIMEZ,TARS,IELAPS,ZEIT,TIME,DELTA1,XP,YP,DEF,DPS0,J1,J2	DK120200
	COMMON EPIP4,EPSIG2,TM,FTM,SAS2,SRASH	DK120210
	COMMON G,RHRCR,RHO,CP,EPSLN,ITK,KITER	DK120220
	COMMON QNET,QSAT,QINT,QEXT,QPLAN,QALB,QSUN,GOLD,QNEW,TBREAK	DK120230
	COMMON GAM,PHIC,ALI,ALII,ANGS,CTHEI,FD,FF	DK120240
	COMMON PHI1,PHI2,BUFFER	DK120250
	COMMON COSLS,SINLS,DT,T,SINL,COSL,SINO,COSO,THICK,NCOAT,NSUBS,	DK120260
	1COSRS,PHI1,GAMM	DK120270
	COMMON ESUN,EEE,RO,SP,TQINT,NDUTY	DK120280
	DIMENSION KC(4)	DK120290
	PHI1=-1000.0	DK120300
	PHI2=-1000.0	DK120310
C	PHI1 AND PHI2 =-1000 WILL SHOW THAT NO SUN-SHADE POINTS WERE FOUND	DK120320
	XS1=0.0	DK120330
	XS2=0.0	DK120340
	YS1=0.0	DK120350
	YS2=0.0	DK120360
1	SINSG=SSIGMA	DK120370
	COSSG=CSIGMA	DK120380
	IF(ABS(COSB)-.01) 2,2,12	DK120390

C	SUN IS IN ORBITAL PLANE	DK120400
2	CALL BETA90	DK120410
	GO TO 99	DK120420
C	SUN IS NOT IN ORBITAL PLANE	DK120430
12	U2=C2*A	DK120440
	U1=C1*A*A	DK120450
	U=C/A	DK120460
	D31=C3-U1	DK120470
	UT(1,1)=C3+C3	DK120480
	US=D2*D2	DK120490
	UU=C3-1.0	DK120500
	A4=D31*D31+DS	DK120510
	A3=2.0*D*(DT(1,1)*U31+DS)	DK120520
	A2=D*D*(DT(1,1)*(DT(1,1)+D31)+DS)-2.0*DU*D31-DS	DK120530
	A1=D*DT(1,1)*(D*U*DT(1,1)-2.0*DU)	DK120540
	A0=(DU-D*U*C3)**2	DK120550
C	CALL ROUTINE TO FACTOR QUARTIC	DK120560
	CALL QUART(A4,A3,A2,A1,A0)	DK120570
37	KR=U	DK120580
	SSHA1=0.0	DK120590
	SSHA2=0.0	DK120600
	JC=1	DK120610
	DO 375 J=1,4	DK120620
	IF (ABS(V(J))-100) 371,375,375	DK120630
371	KC(JC)=J	DK120640
	JC=JC+1	DK120650
375	CONTINUE	DK120660
	IF(JC-3) 49,376,380	DK120670
C	TWO REAL ROOTS, PUT THEM FIRST AND SECOND	DK120680
376	KK=2	DK120690
	GO TO 39	DK120700
C	FOUR REAL ROOTS, LOOK FOR A REPEATED ROOT	DK120710
380	DO 385 K=1,2	DK120720
	KP=K+1	DK120730
	DO 384 J=KP,4	DK120740
	UIF=ABS(U(J)-U(K))-0.001	DK120750
	IF(UIF) 381,384,384	DK120760
C	PUT REPEATED ROOTS TOGETHER, FIRST OR LAST PAIR	DK120770
C	TO AVOID TROUBLE IN GETTING CORRESPONDING Y	DK120780
381	N=3-K	DK120790
	KUB=U(N)	DK120800
	CRASH=V(N)	DK120810
	U(N)=U(J)	DK120820
	V(N)=V(J)	DK120830
	U(J)=KUB	DK120840
	V(J)=CRASH	DK120850
	GO TO 386	DK120860
384	CONTINUE	DK120870
385	CONTINUE	DK120880
386	KK=4	DK120890
C	FIND VALUES OF Y CORRESPONDING TO EACH REAL X	DK120900
39	J=1	DK120910
3907	K=KC(J)	DK120920
41	CALL WYE(U(1),K,Y1,P1,NP)	DK120930
410	SS=P1-SIGMA	DK120940
C	IS PH1 A SUPERFLUOUS ROOT	DK120950
	IF(COS(SS)) 42,60,60	DK120960
42	IF(KR) 43,43,44	DK120970
C	FIRST SUN-SHADE CHANGE POINT	DK120980
43	SSHA1=SS	DK120990

	PH11=P1	DK121000
	XS1=U(K)*A	DK121010
	YS1=Y1*A	DK121020
	KR=1	DK121030
	GO TO 60	DK121040
C	SECOND SUN-SHADE CHANGE POINT	DK121050
44	SSHA2=SS	DK121060
	PH12=P1	DK121070
	XS2=U(K)*A	DK121080
	YS2=Y1*A	DK121090
45	KR=2	DK121100
	GO TO 99	DK121110
C	IF X IS A REPEATED ROOT USE + AND - Y VALUES	DK121120
60	IF(NP) 70,70,61	DK121130
61	NP=0	DK121140
	J=J+1	DK121150
	P1=-P1	DK121160
	Y1=-Y1	DK121170
	GO TO 410	DK121180
70	J=J+1	DK121190
	IF(J-KK) 3907,3907,99	DK121200
99	RETURN	DK121210
	END	DK121220
	FOR DECK13,DECK13	DK130000
	SUBROUTINE SUNOR(SL,CL,SO,CO,SB,CB,CU,GM,N0)	DK130010
	COSLP=SO*CL	DK130020
	SINLP=SO*SL	DK130030
	IF(N0) 250,99,205	DK130040
205	GM=ARCCOS(COSLP)	DK130050
	GO TO 90	DK130060
250	CD=COSLP	DK130070
	COSLP=SB*CD-CB*CU	DK130080
	CO=CO*SB+CB*CD	DK130090
90	CL=COSLP	DK130100
	SL=SINLP	DK130110
99	RETURN	DK130120
	END	DK130130
	FOR DECK14,DECK14	DK140000
	SUBROUTINE BETA90	DK140010
	DIMENSION TRASH(17),F(10,9,42),HTAB(9),ANGTAB(10),WH(18),XH(9),	DK140020
	1YH(6),ZH(6),ELL(9),RPP(9),RR(9),CAYY(9),TIME(2)	DK140030
	DIMENSION DT(2,200),T(2,200),SINL(200),COSL(200),SINU(200),	DK140040
	1COS0(200),THICK(200),NCOAT(200),NSUBS(200),COSKS(200),PHIT(200),	DK140050
	2GAMM(200),NDUTY(200)	DK140060
	DIMENSION ESUN(8),EEE(8,42),RO(8,42),SP(8,42),TQINT(41,8)	DK140070
	DIMENSION BUFFER(8)	DK140080
	COMMON TRASH,F,HTAB,ANGTAB,WH,XH,YH,ZH,ELL,RPP,RR,CAYY,PI,PIH,	DK140090
	11WUPI,PI180,NOFIND,NQORT,IFIRST,NFWSIG	DK140100
	COMMON KPLNET,NORIEN,KTEMP,NUMRUN,NSATP,NPRINT,KREV,NP50,REV	DK140110
	COMMON A,B,C,AYF,BEE,RP,RN,PEE,EL,K,CAY,BAKL,S,ALP2,BET2,GAM2,	DK140120
	1ALPHA2,BETA2,GAMMA2,COSA,COSB,COSG,SINB,PHIMAX	DK140130
	COMMON SIGMA,CSIGMA,SSIGMA,TSIGMA	DK140140
	COMMON PHIZ2,DPHI2,PHIZ,DPHI,PHI,CPH1,SPHI,PHIN2,PHOT2,SUN	DK140150
	COMMON TIMEZ,TARS,1ELAPS,ZEIT,TIME,DELTAI,XP,YP,DEF,DPS0,J1,J2	DK140160
	COMMON EPTP4,EPSIG2,TM,FTM,SAS2,SRASH	DK140170
	COMMON G,KHRCR,RHO,CP,EPSLN,ITK,KITEK	DK140180

COMMON QNET,QSAT,QINT,QEXT,QPLAN,QALB,QSUN,QOLD,QNEW,TBREAK	DK140190
COMMON GAM,PHIC,AL1,AL11,ANGS,CTHET,FD,FF	DK140200
COMMON PH11,PH12,BUFFER	DK140210
COMMON COSLS,SINLS,DT,T,SINL,COSL,SINO,COSU,THICK,NCOAT,NSUBS,	DK140220
1COSRS,PH1T,GAMM	DK140230
COMMON ESUN,LEE,KO,SP,TQINT,NDUTY	DK140240
DIMENSION PAK(2),P(2)	DK140250
PAK(1)=-1.0	DK140260
PAK(2)=1.0	DK140270
ARP=A*RP	DK140280
BRP=B*RP	DK140290
CRP=C*RP	DK140300
BB=B*P	DK140310
IF(CSIGMA) 20,1,20	DK140320
1 DO 10 J=1,2	DK140330
CP=-ARP/(CRP+BB*PAK(J))	DK140340
SP(1,1)=SQRT(1.0-CP**2)	DK140350
IF(SSIGMA) 3,3,2	DK140360
2 SP(1,1)=-SP(1,1)	DK140370
3 TP=SP(1,1)/CP	DK140380
P(J)=ATAN(TP)	DK140390
IF(CP) 4,10,10	DK140400
4 P(J)=P(J)+PI	DK140410
10 CONTINUE	DK140420
11 PH11=P(1)	DK140430
PH12=P(2)	DK140440
99 RETURN	DK140450
20 DO 40 J=1,2	DK140460
BSCR=BB*SSIGMA+PAK(J)*CRP	DK140470
IF(BSCR) 28,21,28	DK140480
21 SP(1,1)=AKP*PAK(J)/(BB*CSIGMA)	DK140490
CP=SQRT(1.0-SP(1,1)**2)	DK140500
IF(CP*CSIGMA+SP(1,1)*SSIGMA) 23,23,22	DK140510
22 CP=-CP	DK140520
23 IF(CP) 26,24,26	DK140530
24 P(J)=0.5*PI	DK140540
IF(SP(1,1)) 25,40,40	DK140550
25 P(J)=1.5*PI	DK140560
GO TO 40	DK140570
26 TP=SP(1,1)/CP	DK140580
P(J)=ATAN(TP)	DK140590
IF(CP) 27,40,40	DK140600
27 P(J)=P(J)+PI	DK140610
GO TO 40	DK140620
28 BS=-BR*CSIGMA	DK140630
CC=PAK(J)*ARP	DK140640
RUB=BS*BS+BSCR*BSCR	DK140650
I1=-CC*BSCR/RUB	DK140660
I2=BS*SQRT(RUB-CC*CC)/RUB	DK140670
29 CP=I1+I2	DK140680
SP(1,1)=- (CP*BSCR+CC)/BS	DK140690
IF(CP*CSIGMA+SP(1,1)*SSIGMA) 23,30,30	DK140700
30 CP=I1-I2	DK140710
SP(1,1)=- (CP*BSCR+CC)/BS	DK140720
GO TO 23	DK140730
40 CONTINUE	DK140740
GO TO 11	DK140750
END	DK140760

	FOR DECK15,DECK15	DK150000
	SUBROUTINE WYE(U,KK,Y1,P1,NP)	DK150010
	DIMENSION X(3),Y(3),U(4),V(4)	DK150020
	EQUIVALENCE(X(1),TRASH(1)),(Y(1),TRASH(4))	DK150030
	EQUIVALENCE(V(1),TRASH(11)),(C1,TRASH(15)),(C2,TRASH(16))	DK150040
	EQUIVALENCE(C3,TRASH(17))	DK150050
	DIMENSION TRASH(17),F(10,9,42),HIAB(9),ANGTAB(10),WH(18),XH(9),	DK150060
	IYH(6),ZH(6),ELL(9),RPP(9),RR(9),CAYY(9),TIME(2)	DK150070
	DIMENSION DT(2,200),T(2,200),SINL(200),COSL(200),SINO(200),	DK150080
	ICOSO(200),THICK(200),NCOAT(200),NSUBS(200),COSKS(200),PHIT(200),	DK150090
	2GAMM(200),NDUTY(41,8)	DK150100
	DIMENSION ESUN(8),EEE(8,42),RO(8,42),SP(8,42),TQINT(41,8)	DK150110
	DIMENSION BU-FER(10)	DK150120
	COMMON TRASH,F,HIAB,ANGTAB,WH,XH,YH,ZH,ELL,RPP,RR,CAYY,PI,PIH,	DK150130
	11WUPI,PI1H0,NOFIND,NQORT,IFIRST,NEWSIG	DK150140
	COMMON KPLNET,NOKIEN,KTEMP,NUMRUN,NSATP,NPRINT,KREV,NP50,REV	DK150150
	COMMON A,B,C,AYF,BEE,RP,RRN,PEE,EL,R,CAY,BARL,S,ALP2,BFT2,GAM2,	DK150160
	1ALPHA2,BETA2,GAMMA2,COSA,COSB,COSG,SINB,PHIMAX	DK150170
	COMMON SIGMA,CSIGMA,SSIGMA,TSIGMA	DK150180
	COMMON PHIZ2,DPH12,PHIZ,DPHI,PHI,CPH1,SPHI,PHIN2,PHOT2,SUN	DK150190
	COMMON TIMEZ,TABS,TELAPS,ZEIT,TIME,DELTA1,XP,YP,DEE,DPSQ,J1,J2	DK150200
	COMMON EPTP4,EPSIG2,TM,FTM,SAS2,SRASH	DK150210
	COMMON G,RHRC,P,RHO,CP,EPSLN,11K,KITEK	DK150220
	COMMON QNET,QSAT,QINT,QEXT,QPLAN,QALB,QSUN,GOLD,QNEW,TBREAK	DK150230
	COMMON GAM,PHIC,ALI,ALTI,ANGS,CTHET,FD,FF	DK150240
	COMMON BU-FER	DK150250
	COMMON COSLS,SINLS,DT,T,SINL,COSL,SINO,COSO,THICK,NCOAT,NSUBS,	DK150260
	1COSRS,PHI1,GAMM	DK150270
	COMMON ESUN,EEE,RO,SP,TQINT,NDUTY	DK150280
	X(1)=U(KK)	DK150290
50	B0A=B/A	DK150300
	COA=C/A	DK150310
	DISC=1.0-(X(1)+COA)**2	DK150320
	IF(DISC+.0001) 90,2,2	DK150330
2	YDEN=2.0*SSIGMA*CSIGMA*(1.0-COSB**2)*X(1)	DK150340
	IF(ABS(YDEN)-.0020) 3,4,4	DK150350
C	NO UNIQUE Y FOR THIS X VALUE	DK150360
3	Y(1)=B0A*SQR1(DISC)	DK150370
	NP=1	DK150380
	GO TO 10	DK150390
C	UNIQUE Y FOR THIS X VALUE	DK150400
4	YNUM=X(1)*X(1)*((COSB*CSIGMA)**2+SSIGMA**2)-(PEE/A)**2	DK150410
	YNUM=YNUM+B0A*B0A*DISC*((COSB*SSIGMA)**2+CSIGMA**2)	DK150420
	Y(1)=YNUM/YDEN	DK150430
	NP=0	DK150440
10	IF(X(1)) 15,11,15	DK150450
11	PHI=1.57079365	DK150460
	GO TO 20	DK150470
15	PHI=ATAN(Y(1)/X(1))	DK150480
	IF(X(1)) 16,20,20	DK150490
16	PHI=PHI+3.1415927	DK150500
20	TORB=(Y(1)/B0A)**2-DISC	DK150510
	IF(ABS(TORB)-.001) 21,21,90	DK150520
21	TSHADE=(COSB*(X(1)*CSIGMA+Y(1)*SSIGMA)**2+(X(1)*SSIGMA-Y(1)*CSIGMA	DK150530
	1A)**2	DK150540
	TSHADE=(A/BEE)**2*(TSHADE-1.0	DK150550
	IF(ABS(TSHADE)-.001) 95,95,90	DK150560
90	P1=SIGMA	DK150570
	GO TO 96	DK150580
95	P1=PHI	DK150590

```

96      Y1=Y(1)
      RETURN
      END
DK150600
DK150610
DK150620

      FOR      DECK16,DECK16
      SUBROUTINE SIGBEI(KABG,LSHADE)
      DIMENSION TRASH(17),F(10,9,42),HTAB(9),ANGTAB(10),WH(18),XH(9),
      1YH(6),ZH(6),ELL(9),RPP(9),RR(9),CAYY(9),TIME(2)
      DIMENSION DT(2,200),T(2,200),SINL(200),COSL(200),SINO(200),
      1COSO(200),THICK(200),NCOAT(200),NSURS(200),COSRS(200),PHIT(200),
      2GAMM(200),NDUTY(200)
      DIMENSION ESUN(8),EEE(8,42),RO(8,42),SP(8,42),TQINT(41,8)
      DIMENSION BUFFER(8)
      COMMON TRASH,F,HTAB,ANGTAB,WH,XH,YH,ZH,ELL,RPP,RR,CAYY,PI,PIH,
      1IWUPI,PI180,NOFIND,NQORT,IFIRST,NEWS16
      COMMON KPLNET,NORIEN,KTEMP,NUMRUN,NSATP,NPRINT,KREV,NP50,REV
      COMMON A,B,C,AYE,BEE,RP,RN,PEE,EL,R,CAY,BARL,S,ALP2,BET2,GAM2,
      1ALPHA2,BETA2,GAMMA2,COSA,COSB,COSG,SINB,PHIMAX
      COMMON SIGMA,CSIGMA,SSIGMA,TSIGMA
      COMMON PHIZ,DPHI2,PHIZ,DPHI,PHI,CPHI,SPHI,PHIN2,PHOT2,SUN
      COMMON TIMEZ,TABS,IELAPS,ZEIT,TIME,DELTAT,XP,YP,DEE,DPSQ,J1,J2
      COMMON EP1P4,EPS1G2,TM,FTM,SA52,SRASH
      COMMON G,RHRC,P,RH0,CP,EPSLN,ITK,KITEH
      COMMON QNET,QSAT,QINT,QEXT,QPLAN,QALB,QSUN,QOLD,QNEW,TBREAK
      COMMON GAM,PHIC,ALF,ALTI,ANGS,CTHET,FD,FF
      COMMON PH11,PHI2,BUFFER
      COMMON COSLS,SINLS,DT,T,SINL,COSL,SINO,COSO,THICK,NCOAT,NSURS,
      1COSRS,PH11,GAMM
      COMMON ESUN,EEE,RO,SP,TQINT,NDUTY,ANINCL,ASCNOD,ASNLNG,RGTASC,
      1DECLIN
      EQUIVALENCE(1TRASH(15),C1),(1TRASH(16),C2),(1TRASH(17),C3)
      FIND SIGMA IF PART OF ORBIT MAY BE SHADED
      SUN=1.0
      IF(KABG) 600,700,600
      IF KABG IS 0, INPUT IS ALPHA, BETA, GAMMA
      600 CW=COS(ASCNOD*PI180)
      SW=SIN(ASCNOD*PI180)
      SO=SIN(ASNLNG*PI180)
      CO=COS(ASNLNG*PI180)
      CI=COS(ANINCL*PI180)
      SI=SIN(ANINCL*PI180)
      SR=SIN(RGTASC*PI180)
      CR=COS(RGTASC*PI180)
      CD=COS(DECLIN*PI180)
      SD=SIN(DECLIN*PI180)
      COSA=((CW*CO-SW*SO*CI)*CR+SR*(CW*SO+SW*CO*CI))*CD+SW*SI*SD
      ALP2=ARCUS(COSA)
      COSB=CD*SI*(SO*CR-CO*SR)+CI*SD
      BET2=ARCUS(COSB)
      SINB=SIN(PI180*BET2)
      COSG=CD*(SR*(CO*CW*CI-SO*SW)-(CR*(CW*SIN*CI+SW*CO))+CW*SI*SD
      GAM2=ARCUS(COSG)
      700 IF(ABS(COSA)-.01) 701,710,710
      701 IF(ABS(COSG)-.01) 702,7020,7020
      7020 IF(COSG) 704,703,703
      702 SUN=2.0
      **** THE FOLLOWING CARD WAS ADDED 12/15/66 TO ELIMINATE TERMINATOR
      ORBIT PROBLEMS.
      SSIGMA=0.0
      ****
      DK160000
      DK160010
      DK160020
      DK160030
      DK160040
      DK160050
      DK160060
      DK160070
      DK160080
      DK160090
      DK160100
      DK160110
      DK160120
      DK160130
      DK160140
      DK160150
      DK160160
      DK160170
      DK160180
      DK160190
      DK160200
      DK160210
      DK160220
      DK160230
      DK160240
      DK160250
      DK160260
      DK160270
      DK160280
      DK160290
      DK160300
      DK160310
      DK160320
      DK160330
      DK160340
      DK160350
      DK160360
      DK160370
      DK160380
      DK160390
      DK160400
      DK160410
      DK160420
      DK160430
      DK160440
      DK160450
      DK160460
      DK160470
      DK160480
      DK160490
      DK160500
      DK160510
      DK160520
      DK160530
      DK160540

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**** THE FOLLOWING CARD WAS ADDED 12/15/66 TO ELIMINATE TERMINATOR
      ORBIT PROBLEMS. ****
      CSIGMA=1.0
      SIGMA=0.0
      GO TO 722
703  SIGMA=PIH
      SIGMA2=90.0
      SSIGMA=1.0
      GO TO 705
704  SIGMA=1.5*PI
      SIGMA2=270.0
      SSIGMA=-1.0
705  CSIGMA=0.0
      ISIGMA=SSIGMA*1.E 20
      GO TO 722
'10  IF (ABS(COSG)-.01) /11,715,715
'11  IF (COSA) 713,712,712
'12  SIGMA=0.0
      CSIGMA=1.0
      GO TO 714
'13  SIGMA=PI
      CSIGMA=-1.0
'14  SSIGMA=0.0
      ISIGMA=0.0
      GO TO 722
'15  CSIGMA=COSA/SQRT(COSA**2+COSG**2)
      SSIGMA=SQRT(1.0-CSIGMA**2)
      IF (COSG) 716,717,717
16   SSIGMA=-SSIGMA
17   CALL DVCHK (K000FX)
      GO TO(717,717),K000FX
171  ISIGMA=SSIGMA/CSIGMA
      CALL DVCHK (K000FX)
      GO TO(703,718),K000FX
18   SIGMA=ATAN(ISIGMA)
      IF (CSIGMA) /19,720,720
19   SIGMA=SIGMA+PI
20   IF (SIGMA) 721,722,722
21   SIGMA=SIGMA+PI+PI
22   IF (NORIEH) 723,720,723
23   DO 725 J=1,NSATP
      CALL SUNOR(SINL(J),COSL(J),SINO(J),COSO(J),SINB,COSB,COSHS(J),
1GAMM(J),NORIEH)
25   CONTINUE
      FIND INTERSECTION OF ELLIPSES
2J9  IF (SUN-2.0) 210,199,199
10   IF (ABS(COSB)-.01) 1001,211,211
11   AYE=ABS(KP/COSB)
      BEE=KF
      BEESQ=BEE*BEE
      C1=((CSIGMA*COSB)**2+SSIGMA**2)/BEESQ
      C2=(COSB*COSB-1.0)*2.0*B*SSIGMA*CSIGMA/BEESQ
      C3=((SSIGMA*COSB)**2+CSIGMA**2)*(H/REF)**2
1J01 IF (LSHADE) 199,1002,199
J02  CALL FIND
J9   CONTINUE
      CALL INII
      RETURN
      END

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DK160550
 DK160560
 DK160570
 DK160580
 DK160590
 DK160600
 DK160610
 DK160620
 DK160630
 DK160640
 DK160650
 DK160660
 DK160670
 DK160680
 DK160690
 DK160700
 DK160710
 DK160720
 DK160730
 DK160740
 DK160750
 DK160760
 DK160770
 DK160780
 DK160790
 DK160800
 DK160810
 DK160820
 DK160830
 DK160840
 DK160850
 DK160860
 DK160870
 DK160880
 DK160890
 DK160900
 DK160910
 DK160920
 DK160930
 DK160940
 DK160950
 DK160960
 DK160970
 DK160980
 DK160990
 DK161000
 DK161010
 DK161020
 DK161030
 DK161040
 DK161050
 DK161060
 DK161070
 DK161080
 DK161090
 DK161100
 DK161110
 DK161120
 DK161130

FOR DECK17,DECK17	DK170000
SUBROUTINE INTERP(PROP,TEMP,JX,ANS)	DK170010
PROP IS EEE(EPSILON),RO(RHO), OR SP(SPEC. HEAT)	DK170020
TEMP IS TEMPERATURE OF ELEMENT	DK170030
JX IS INDEX INDICATING WHICH COATING TABLE TO USE-OR SUBSTRATE	DK170040
ANS IS EPSILON, RHO, OR CP - RESPECTIVELY	DK170050
DIMENSION PROP(8,42)	DK170060
DO 16 I=1,42,2	DK170070
1 IF(TEMP-PROP(JX,I))11,13,12	DK170080
1 IF(TEMP-PROP(JX,I+2))16,14,15	DK170090
2 IF(TEMP-PROP(JX,I+2))15,14,16	DK170100
3 ANS=PROP(JX,I+1)	DK170110
RETURN	DK170120
4 ANS=PROP(JX,I+3)	DK170130
RETURN	DK170140
3 ANS=PROP(JX,I+1)+((PROP(JX,I+3)-PROP(JX,I+1))*	DK170150
1(TEMP-PROP(JX,I))/(PROP(JX,I+2)-PROP(JX,I))	DK170160
RETURN	DK170170
3 CONTINUE	DK170180
RETURN	DK170190
END	DK170200
FOR DECK18,DECK18	DK180000
SUBROUTINE AKROUI(A,LR,NSATP)	DK180010
DIMENSION A(200)	DK180020
1 DO 20 J=1,LR	DK180030
I4=10*J-9	DK180040
NF=MIN0(I4+9,NSATP)	DK180050
WRITE (6,705)I4,(A(K),K=I4,NF)	DK180060
75 FORMAT(14X14,2X10F10.2)	DK180070
9 RETURN	DK180080
END	DK180090
FOR DECK19,DECK19	DK190000
FUNCTION ARCOS(C)	DK190010
IF(C) 3,2,3	DK190020
ARCOS=90.0	DK190030
RETURN	DK190040
3 ARCOS=ATAN(SQRT((1.0/C)**2-1.0))/.017453293	DK190050
IF(C) 4,99,99	DK190060
4 ARCOS=180.0-ARCOS	DK190070
GO TO 99	DK190080
END	DK190090
FOR DECK20,DECK20	DK200000
SUBROUTINE QUART (A4,A3,A2,A1,A0)	DK200010
DIMENSION X(5),Y(3),U(4),V(4)	DK200020
COMMON X,Y,U,V	DK200030
DIMENSION SC(8)	DK200040
SC(1)=A4	DK200050
SC(2)=A3	DK200060
SC(3)=A2	DK200070
SC(4)=A1	DK200080
SC(5)=A0	DK200090
CALL DDFERI(SC(1))	DK200100
DO 5 J=1,4	DK200110

	K=J+J	DK200120
	U(J)=SC(K-1)	DK200130
5	V(J)=SC(K)	DK200140
9	RETURN	DK200150
	END	DK200160
	FOR DECK21,DECK21	DK210000
	FUNCTION PHIN(XN,YN,STEP)	DK210010
	DIMENSION TRASH(17),F(10,9,42),HTAB(9),ANGTAB(10),WH(18),XH(9),	DK210020
	1YH(6),ZH(6),ELL(9),RPP(9),RR(9),CAYY(9),TIME(2)	DK210030
	DIMENSION DT(2,200),T(2,200),SINL(200),COSL(200),SINO(200),	DK210040
	1COSO(200),THICK(200),NCOAT(200),NSURS(200),COSRS(200),PHIT(200),	DK210050
	2GAMM(200),NDUTY(200)	DK210060
	DIMENSION ESUN(8),EEE(8,42),RO(8,42),SP(8,42),TQINT(41,8)	DK210070
	DIMENSION BUFFER(8)	DK210080
	COMMON TRASH,F,HTAB,ANGTAB,WH,XH,YH,ZH,ELL,RPP,RR,CAYY,PI,PIH,	DK210090
	1TWOPI,PI180,NOFIND,NQORT,IFIRST,NEWSIG	DK210100
	COMMON KPLNET,NORIEN,KTEMP,NUMRUN,NSATP,NPRINT,KREV,NP50,REV	DK210110
	COMMON A,B,C,AYF,BEE,RP,RN,PEE,EL,R,CAY,BARL,S,ALP2,BFT2,GAM2,	DK210120
	1ALPHA2,BETA2,GAMMA2,COSA,COSB,COSG,SINB,PHIMAX	DK210130
	COMMON SIGMA,CSIGMA,SSIGMA,TSIGMA	DK210140
	COMMON PHIZ2,DPH12,PHI2,DPHI,PHI,CPH1,SPHI,PHIN2,PHOT2,SUN	DK210150
	COMMON TIME2,TABS,TELAPS,ZEIT,TIME,DELTAT,XP,YP,DEE,UPSQ,J1,J2	DK210160
	COMMON EPTP4,EPSIG2,TM,FTM,SAS2,SRASH	DK210170
	COMMON G,RHRCR,RHO,CP,EPSLN,ITK,KITEK	DK210180
	COMMON QNET,QSAT,QINT,QEXT,QPLAN,QALB,QSUN,GOLD,QNEW,TBREAK	DK210190
	COMMON GAM,PHIC,ALT,ALTI,ANGS,CTHET,FD,FF	DK210200
	COMMON PHI1,PHI2,BUFFER	DK210210
	COMMON COSLS,SINLS,UT,T,SINL,COSL,SINO,COSU,THICK,NCOAT,NSUBS,	DK210220
	1COSRS,PHI1,GAMM	DK210230
	COMMON ESUN,EEE,RO,SP,TQINT,NDUTY,ANINCL,ASCNOD,ASNLNG,RGTASC,	DK210240
	1DECLIN	DK210250
	DIMENSION AA(6),AA1(6),P(6),P1(6)	DK210260
	COMMON AA,AA1,P,P1,IORDER,IORD1,1EKROR,THETA,DTMAX,EN1,EN,FACT,	DK210270
	1YNHAT,ENHATL,EMAG,DEERROR,DTTEST	DK210280
	ARGX=XN	DK210290
	ARGY=YN	DK210300
	PHIFN = 0.0	DK210310
	NU=1	DK210320
10	FNK=FOFXY(ARGX,ARGY)	DK210330
	PHIFN=PHIFN+AA(NU)*FNK	DK210340
	IF (NU-IORDER) 11,12,12	DK210350
11	NU=NU+1	DK210360
	RASH=P(NU)*STEP	DK210370
	ARGX=XN+RASH	DK210380
	ARGY=YN+RASH*FNK	DK210390
	GO TO 10	DK210400
12	RETURN	DK210410
	END	DK210420
	FOR DECK22,DECK22	DK220000
	FUNCTION GFN(X,Y)	DK220010
	DIMENSION TRASH(17),F(10,9,42),HTAB(9),ANGTAB(10),WH(18),XH(9),	DK220020
	1YH(6),ZH(6),ELL(9),RPP(9),RR(9),CAYY(9),TIME(2)	DK220030
	DIMENSION DT(2,200),T(2,200),SINL(200),COSL(200),SINO(200),	DK220040
	1COSO(200),THICK(200),NCOAT(200),NSURS(200),COSRS(200),PHIT(200),	DK220050
	2GAMM(200),NDUTY(200)	DK220060
	DIMENSION ESUN(8),EEE(8,42),RO(8,42),SP(8,42),TQINT(41,8)	DK220070

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DIMENSION BUFFER(8)
COMMON TRASH,F,HIAB,ANGTAB,WH,XH,YH,ZH,ELL,RPP,RR,CAYY,PI,PIH,
1TWOP1,PI180,NOFIND,NQORT,IFIRST,NEWSIG
COMMON KPLNE1,NORLEN,KTEMP,NUMRUN,NSATP,NPRINT,KRFV,NP50,REV
COMMON A,B,C,AYE,BEE,RP,RN,PEE,EL,R,CAY,BARL,S,ALP2,BFT2,GAM2,
1ALPHA2,BETA2,GAMMA2,COSA,COSB,COSG,SINB,PHIMAX
COMMON SIGMA,CSIGMA,SSIGMA,TSIGMA
COMMON PHIZ2,DPH12,PHIZ,DPHI,PHI,CPH1,SPHI,PHIN2,PHOT2,SUN
COMMON TIMEZ,TABS,TELAPS,ZEIT,TIME,DELTAT,XP,YP,DEE,DPSQ,J1,J2
COMMON EPIP4,EPSIG2,TM,FTM,SAS2,SRASH
COMMON G,RHRCR,RHO,CP,EPSLN,ITK,KITEK
COMMON QNET,QSAT,QINT,QEXT,QPLAN,QALB,QSUN,QOLD,QNEW,TBREAK
COMMON GAM,PHIC,ALI,ALTI,ANGS,CTHET,FD,FF
COMMON PH11,PHI2,BUFFER
COMMON COSLS,SINLS,DT,T,SINL,COSL,SINO,COSO,THICK,NCOAT,NSUBS,
1COSRS,PHIT,GAMM
COMMON ESUN,EEE,RO,SP,TQINT,NDUTY,ANINCL,ASCNOD,ASNLNG,RGTASC,
1DECLIN
DIMENSION AA(6),AA1(6),P(6),P1(6)
COMMON AA,AA1,P,P1,IORDER,IORD1,IEKRRP,THETA,DTMAX,EN1,EN,FACT,
1YNHAT,ENHATL,EMAG,UEKRRR,DTTEST
GFN=2.0*EPSIG2*Y**3
RETURN
END

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DK220080
DK220090
DK220100
DK220110
DK220120
DK220130
DK220140
DK220150
DK220160
DK220170
DK220180
DK220190
DK220200
DK220210
DK220220
DK220230
DK220240
DK220250
DK220260
DK220270
DK220280
DK220290
DK220300
DK220310

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      FOR      DECK23,DECK23
FUNCTION FOFXY(X,Y)
DIMENSION TRASH(17),F(10,9,42),HIAB(9),ANGTAB(10),WH(18),XH(9),
1YH(6),ZH(6),ELL(9),RPP(9),RR(9),CAYY(9),TIME(2)
DIMENSION DT(2,200),T(2,200),SINL(200),COSL(200),SINO(200),
1COSO(200),THICK(200),NCOAT(200),NSURS(200),COSRS(200),PHIT(200),
2GAMM(200),NDUTY(200)
DIMENSION ESUN(8),EEE(8,42),RO(8,42),SP(8,42),TQINT(41,8)
DIMENSION BUFFER(8)
COMMON TRASH,F,HIAB,ANGTAB,WH,XH,YH,ZH,ELL,RPP,RR,CAYY,PI,PIH,
1TWOP1,PI180,NOFIND,NQORT,IFIRST,NEWSIG
COMMON KPLNE1,NORLEN,KTEMP,NUMRUN,NSATP,NPRINT,KRFV,NP50,REV
COMMON A,B,C,AYE,BEE,RP,RN,PEE,EL,R,CAY,BARL,S,ALP2,BFT2,GAM2,
1ALPHA2,BETA2,GAMMA2,COSA,COSB,COSG,SINB,PHIMAX
COMMON SIGMA,CSIGMA,SSIGMA,TSIGMA
COMMON PHIZ2,DPH12,PHIZ,DPHI,PHI,CPH1,SPHI,PHIN2,PHOT2,SUN
COMMON TIMEZ,TABS,TELAPS,ZEIT,TIME,DELTAT,XP,YP,DEE,DPSQ,J1,J2
COMMON EPIP4,EPSIG2,TM,FTM,SAS2,SRASH
COMMON G,RHRCR,RHO,CP,EPSLN,ITK,KITEK
COMMON QNET,QSAT,QINT,QEXT,QPLAN,QALB,QSUN,QOLD,QNEW,TBREAK
COMMON GAM,PHIC,ALI,ALTI,ANGS,CTHET,FD,FF
COMMON PH11,PHI2,BUFFER
COMMON COSLS,SINLS,DT,T,SINL,COSL,SINO,COSO,THICK,NCOAT,NSURS,
1COSRS,PHIT,GAMM
COMMON ESUN,EEE,RO,SP,TQINT,NDUTY,ANINCL,ASCNOD,ASNLNG,RGTASC,
1DECLIN
DIMENSION AA(6),AA1(6),P(6),P1(6)
COMMON AA,AA1,P,P1,IORDER,IORD1,IEKRRP,THETA,DTMAX,EN1,EN,FACT,
1YNHAT,ENHATL,EMAG,UEKRRR,DTTEST
FOFXY=RHRCR*(QNET+QINT+0.5*EPSIG2*Y**4)
RETURN
END

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DK230000
DK230010
DK230020
DK230030
DK230040
DK230050
DK230060
DK230070
DK230080
DK230090
DK230100
DK230110
DK230120
DK230130
DK230140
DK230150
DK230160
DK230170
DK230180
DK230190
DK230200
DK230210
DK230220
DK230230
DK230240
DK230250
DK230260
DK230270
DK230280
DK230290
DK230300
DK230310

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      FOR      DECK24,DECK24
      FUNCTION DELTA(XN,YN,STEP,F0)
      DIMENSION TRASH(17),F(10,9,42),H1AB(9),ANGTAB(10),WH(18),XH(9),
1YH(6),ZH(6),ELL(9),RPP(9),RR(9),CAYY(9),TIME(2)
      DIMENSION DT(2,200),T(2,200),SINL(200),COSL(200),SINO(200),
1COS0(200),THICK(200),NCOAT(200),NSURS(200),COSRS(200),PHIT(200),
2GAMM(200),NDUTY(200)
      DIMENSION ESUN(8),EEE(8,42),RO(8,42),SP(8,42),TQINT(41,8)
      DIMENSION BUFFER(8)
      COMMON TRASH,F,H1AB,ANGTAB,WH,XH,YH,ZH,ELL,RPP,RR,CAYY,PI,PIH,
1IWOP1,PI180,NOFIND,NQORT,IFIRST,NEWSIG
      COMMON KPLNE1,NORLEN,KTEMP,NUMRUN,NSATP,NPRINT,KREV,NP50,REV
      COMMON A,B,C,AYF,BEE,RP,RN,PEE,EL,R,CAY,BARL,S,ALP2,BET2,GAM2,
1ALPHA2,BETA2,GAMMA2,COSA,COSB,COSG,SINB,PHIMAX
      COMMON SIGMA,CSIGMA,SSIGMA,TSIGMA
      COMMON PHIZ2,DPH12,PHIZ,DPHI,PHI,CPH1,SPHI,PHIN2,PHOT2,SUN
      COMMON TIMEZ,TABS,IELAPS,ZEIT,TIME,DELTA1,XP,YP,DEE,DPS0,J1,J2
      COMMON EPIP4,EPSIG2,TM,FTM,SAS2,SRASH
      COMMON G,KHRC,P,RHO,CP,EPSLN,ITK,KITEK
      COMMON QNET,QSAT,QINT,QEXT,QPLAN,QALB,QSUN,GOLD,QNEW,TBREAK
      COMMON GAM,PHIC,ALI,ALTI,ANGS,CTHE1,FD,FF
      COMMON PH11,PHI2,BUFFER
      COMMON COSLS,SINLS,UT,T,SINL,COSL,SINO,COSU,THICK,NCOAT,NSURS,
1COSRS,PHIT,GAMM
      COMMON ESUN,EEE,RO,SP,TQINT,NDUTY,ANINCL,ASCNOU,ASNLNG,RGTASC,
1DECLIN
      DIMENSION AA(6),AA1(6),P(6),P1(6)
      COMMON AA,AA1,P,P1,IORDER,IORD1,IERROR,THETA,DTMAX,EN1,EN,FACT,
1YNHAT,ENHATL,EMAG,UEXERROR,DTTEST
      STEP1=STEP*0.5
      F1=PHIFN(XN,YN,STEP1)
      XN1=XN+STEP1
      YN1=YN+STEP1*F1
      F2=PHIFN(XN1,YN1,STEP1)
      DELTA=STEP1*(F1+F2)-F0*STEP
      RETURN
      END

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DK240000
DK240010
DK240020
DK240030
DK240040
DK240050
DK240060
DK240070
DK240080
DK240090
DK240100
DK240110
DK240120
DK240130
DK240140
DK240150
DK240160
DK240170
DK240180
DK240190
DK240200
DK240210
DK240220
DK240230
DK240240
DK240250
DK240260
DK240270
DK240280
DK240290
DK240300
DK240310
DK240320
DK240330
DK240340
DK240350
DK240360

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      FOR      DECK25,DECK25
      SUBROUTINE ERROR(STEP,F0)
      DIMENSION TRASH(17),F(10,9,42),H1AB(9),ANGTAB(10),WH(18),XH(9),
1YH(6),ZH(6),ELL(9),RPP(9),RR(9),CAYY(9),TIME(2)
      DIMENSION DT(2,200),T(2,200),SINL(200),COSL(200),SINO(200),
1COS0(200),THICK(200),NCOAT(200),NSURS(200),COSRS(200),PHIT(200),
2GAMM(200),NDUTY(200)
      DIMENSION ESUN(8),EEE(8,42),RO(8,42),SP(8,42),TQINT(41,8)
      DIMENSION BUFFER(8)
      COMMON TRASH,F,H1AB,ANGTAB,WH,XH,YH,ZH,ELL,RPP,RR,CAYY,PI,PIH,
1IWOP1,PI180,NOFIND,NQORT,IFIRST,NEWSIG
      COMMON KPLNE1,NORLEN,KTEMP,NUMRUN,NSATP,NPRINT,KREV,NP50,REV
      COMMON A,B,C,AYF,BEE,RP,RN,PEE,EL,R,CAY,BARL,S,ALP2,BET2,GAM2,
1ALPHA2,BETA2,GAMMA2,COSA,COSB,COSG,SINB,PHIMAX
      COMMON SIGMA,CSIGMA,SSIGMA,TSIGMA
      COMMON PHIZ2,DPH12,PHIZ,DPHI,PHI,CPH1,SPHI,PHIN2,PHOT2,SUN
      COMMON TIMEZ,TABS,IELAPS,ZEIT,TIME,DELTA1,XP,YP,DEE,DPS0,J1,J2
      COMMON EPIP4,EPSIG2,TM,FTM,SAS2,SRASH
      COMMON G,KHRC,P,RHO,CP,EPSLN,ITK,KITEK
      COMMON QNET,QSAT,QINT,QEXT,QPLAN,QALB,QSUN,GOLD,QNEW,TBREAK
      COMMON GAM,PHIC,ALI,ALTI,ANGS,CTHE1,FD,FF

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DK250000
DK250010
DK250020
DK250030
DK250040
DK250050
DK250060
DK250070
DK250080
DK250090
DK250100
DK250110
DK250120
DK250130
DK250140
DK250150
DK250160
DK250170
DK250180
DK250190
DK250200

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	COMMON PH11,PHI2,BUFFER	DK250210
	COMMON COSLS,SINLS,DT,T,SINL,COSL,SINO,COSO,THICK,NCOAT,NSURS,	DK250220
	1COSRS,PH11,GAMM	DK250230
	COMMON ESUN,EEE,RO,SP,TQINT,NDUTY,ANINCL,ASCNOD,ASNLNG,RGTASC,	DK250240
	1DECLIN	DK250250
	DIMENSION AA(6),AA1(6),P(6),P1(6)	DK250260
	COMMON AA,AA1,P,P1,IORDER,IORD1,IERROR,THETA,DTMAX,EN1,EN,FACT,	DK250270
	1YNHAT,ENHATL,EMAG,DERROR,DTTEST	DK250280
	TLAST=TIME(J1)	DK250290
	TJ=T(J1,1)	DK250300
	DTIME=STEP	DK250310
C	HENRICI ERROR FUNCTION.	DK250320
	GO TO (1,1,3),IEKKOR	DK250330
1	DEL=DELTA(TLAST,TJ,STEP,F0)	DK250340
2	DERROR = -((THETA/DTMAX)**IORDER)*DEL*FACT*THETA	DK250350
	EN=EN*(1.0+DTIME*GFN(TLAST,TJ))+DERROR	DK250360
3	RETURN	DK250370
	END	DK250380
	FOR DECK26,DECK26	DK260000
	SUBROUTINE GEOFAC(FAC)	DK260010
	DIMENSION TRASH(17),F(10,9,42),HTAB(9),ANGTAB(10),WH(18),XH(9),	DK260020
	1YH(6),ZH(6),ELL(9),RPP(9),RR(9),CAYY(9),TIME(2)	DK260030
	DIMENSION DT(2,200),T(2,200),SINL(200),COSL(200),SINO(200),	DK260040
	1COSO(200),THICK(200),NCOAT(200),NSURS(200),COSKS(200),PHIT(200),	DK260050
	2GAMM(200),NDUTY(200)	DK260060
	DIMENSION ESUN(8),EEE(8,42),RO(8,42),SP(8,42),TQINT(41,8)	DK260070
	DIMENSION BUFFER(10)	DK260080
	COMMON TRASH,F,HTAB,ANGTAB,WH,XH,YH,ZH,ELL,RPP,RR,CAYY,PI,PIH,	DK260085
	1TWOP1,PI180,NOFIND,NQORT,IFIRST,NEWSIG	DK260090
	COMMON KPLNE1,NORIEN,KTEMP,NUMRUN,NSATP,NPRINT,KREV,NP50,REV	DK260100
	COMMON A,B,C,AYE,BEE,RP,RN,PEE,EL,U,CAY,BARL,W,ALP2,BFT2,GAM2,	DK260110
	1ALPHA2,BETA2,GAMMA2,COSA,COSB,COSG,SINB,PHIMAX	DK260120
	COMMON SIGMA,CSIGMA,SSIGMA,TSIGMA	DK260130
	COMMON PHIZ2,UPHI2,PHIZ,DPHI,PHI,CPHI,SPHI,PHIN2,PHOT2,SUN	DK260140
	COMMON TIMEZ,TABS,IELAPS,ZEIT,TIME,DELTA1,XP,YP,DEE,DPS0,J1,J2	DK260150
	COMMON EPIP4,EPSIG2,TM,FTM,SAS2,SRASH	DK260160
	COMMON G,RHRCP,RHO,CP,EPSLN,ITK,KITEK	DK260170
	COMMON QNET,QSAT,QINT,QEXT,QPLAN,QALB,QSUN,QOLD,QNEW,TBREAK	DK260180
	COMMON GAM,PHIC,ALT,ALTI,ANGS,CTHEI,FD,FF	DK260190
	COMMON BUFFER	DK260200
	COMMON COSLS,SINLS,DT,T,SINL,COSL,SINO,COSO,THICK,NCOAT,NSURS,	DK260210
	1COSRS,PH11,GAMM	DK260220
	COMMON ESUN,EEE,RO,SP,TQINT,NDUTY,ANINCL,ASCNOD,ASNLNG,RGTASC,	DK260230
	1DECLIN	DK260240
	DIMENSION R(8),Q(4),P(2),S(2)	DK260250
	GAM30=GAM/30.0	DK260260
	NG30=GAM30	DK260270
	GN30=NG30	DK260280
	GF=GAM30-GN30	DK260290
5	P30=PHIC/30.0	DK260300
	NP30=P30	DK260310
	PN30=NP30	DK260320
	PF=P30-PN30	DK260330
	IF (NP30-6) 6,4,4	DK260340
4	NP30=5	DK260350
	PF=1.0	DK260360
6	NP51=7*NG30+NP30+1	DK260370
C	IF TABLES FOR ZERO H ARE USED, DO J=2,9	DK260380

DO 10 J=3,9	DK260390
NH=J-1	DK260400
IF (HTAB(J)-ALT) 10,7,7	DK260410
10 CONTINUE	DK260420
7 HF=(ALT-HTAB(NH))/(HTAB(NH+1)-HTAB(NH))	DK260430
11 DO 13 J=2,10	DK260440
ANGU=ANGS	DK260450
NA=J-1	DK260460
IF (ANGTAB(J)-ANGU) 13,12,12	DK260470
13 CONTINUE	DK260480
12 AF=(ANGU-ANGTAB(NA))/(ANGTAB(NA+1)-ANGTAB(NA))	DK260490
14 DO 25 JJ=1,2	DK260500
N=JJ-1	DK260510
K=NPS1+7*N	DK260520
DO 15 L=1,2	DK260530
KL=K+L-1	DK260540
LM=(L-1)*4	DK260550
R(LM+1)=F(NA,NH,KL)	DK260560
R(LM+2)=F(NA,NH+1,KL)	DK260570
R(LM+3)=F(NA+1,NH,KL)	DK260580
15 R(LM+4)=F(NA+1,NH+1,KL)	DK260590
DO 18 J=1,4	DK260600
K=J+J	DK260610
18 Q(J)=R(K-1)+HF*(R(K)-R(K-1))	DK260620
DO 20 J=1,2	DK260630
K=J+J	DK260640
20 P(J)=Q(K-1)+AF*(Q(K)-Q(K-1))	DK260650
S(JJ)=P(1)+PF*(P(2)-P(1))	DK260660
IF (NPS1-35) 25,22,22	DK260670
22 S(2)=0.0	DK260680
GO TO 26	DK260690
25 CONTINUE	DK260700
26 FAC=S(1)+GF*(S(2)-S(1))	DK260710
IF (ALT-HTAB(9)) 27,27,98	DK260720
27 IF (CUS(ANGS*PI18U)+.6428) 98,98,99	DK260730
99 IF (FAC) 98,100,100	DK260740
100 IF (1.0-FAC) 998,999,999	DK260750
98 FAC=0.0	DK260760
GO TO 999	DK260770
998 FAC=1.0	DK260780
999 RETURN	DK260790
END	DK260800
FOR DECK27,DECK27	DK270000
SUBROUTINE Q1IN(KK,LL,W)	DK270010
DIMENSION TQINT(41,8),TRASH(7818),W(1)	DK270020
COMMON TRASH,TQINT	DK270030
L=LL	DK270040
K=KK	DK270050
IF (L) 61,61,62	DK270060
C IF L=0, TABLE ENTRIES ARE CLEARED	DK270070
61 TQINT(1,K)=0.0	DK270080
TQINT(41,K)=0	DK270090
GO TO 99	DK270100
62 TQINT(41,K)=2.0	DK270110
DO 63 J=1,7	DK270120
63 TQINT(J,K)=W(J)	DK270130
C IF L=1,2 OR 3, NO MORE CARDS ARE READ	DK270140
C IF L IS LARGER, READ MORE CARDS	DK270150

IF(3-L) 64,99,99	DK270160
64 READ(5,65) (IQIN(2*J,K),TQINT(2*J+1,K),J=4,L)	DK270170
99 TQINT(2*L+2,K)=100000000.0	DK270180
WRITE (11)K,L,(TQINT(J,K),J=1,41)	DK270190
RETURN	DK270200
65 FORMAT(2X6F8.2,2F15.2)	DK270210
END	DK270220
FOR DECK28,DECK28	DK280000
SUBROUTINE DDVETA (C, RT, MTYPE)	DK280010
DIMENSION C(4), RT(3), X(2)	DK280020
C THE FOLLOWING STATEMENT(S) HAVE BEEN MANUFACTURED BY THE TRANSLATOR---	DK280030
C	DK280040
DOUBLE PRECISION C , RT , X , A , B	DK280050
DOUBLE PRECISION Q , R , USQR , CORRECT	DK280060
DOUBLE PRECISION PHI1 , DATAN2 , POD , FXP0 , DLOG	DK280070
DOUBLE PRECISION DEXP , DCOS , DSIN , CLOD , PLOD	DK280080
DO 10 L=2,4	DK280090
10 C(L)=C(L)/C(1)	DK280100
A=(3.000*C(3)-C(2)**2)/3.000	DK280110
B=(2.000*(C(2)**3-9.000*C(2)*C(3)+27.000*C(4))/27.000	DK280120
Q=B**2/4.000+A**3/27.000	DK280130
IF (B) 600, 700, 600	DK280140
700 IF (ABS(A)-.001) 12, 12, 300	DK280150
600 IF (ABS(Q)-10.0**(-15)*B**2/4.0) 12, 300, 300	DK280160
300 IF (Q) 11, 12, 13	DK280170
12 MTYPE=0	DK280180
GO TO 14	DK280190
13 MTYPE=1	DK280200
GO TO 15	DK280210
11 MTYPE=-1	DK280220
Q = DABS(Q)	DK280230
R = DUSQR(Q)	DK280240
CORRECT=-B/2.000	DK280250
PHI1=DATAN2(R,CORRECT)/3.000	DK280260
POD = DSQR(R**2/4.000+Q)	DK280270
IF (POD) 73, 70, 73	DK280280
70 X(1)=0.000	DK280290
X(2)=0.000	DK280300
GO TO 17	DK280310
73 EXP0=(DLOG(POD))/3.000	DK280320
POD=DEXP(EXP0)	DK280330
X(1)= POD*DCOS(PHI1)	DK280340
X(2)= POD*DSIN(PHI1)	DK280350
GO TO 17	DK280360
14 POD=-R/2.000	DK280370
IF (POD) 83, 80, 83	DK280380
80 X(1)=0.000	DK280390
GO TO 84	DK280400
83 EXP0=(DLOG(DABS(POD)))/3.000	DK280410
X(1)=(POD /DABS(POD))*DEXP(EXP0)	DK280420
84 RT(1)=2.000*X(1)	DK280430
RT(2)=-X(1)	DK280440
RT(3)=RT(2)	DK280450
GO TO 199	DK280460
15 Q=USQR(Q)	DK280470
CLOD=-B/2.000+Q	DK280480
PLOD=CLOD-2.000*Q	DK280490
IF (CLOD) 96, 95, 96	DK280500

95	X(1)=0.000	DK280510
	GO TO 97	DK280520
96	EXP0=(DLOG(DABS(CLOUD)))/3.000	DK280530
	X(1)=(CLOUD/DABS(CLOUD))*DEXP(EXP0)	DK280540
97	IF (PLOW) 91, 90, 91	DK280550
90	X(2)=0.000	DK280560
	GO TO 16	DK280570
91	EXP0=(DLOG(DABS(PL00)))/3.000	DK280580
	X(2)=(PLOW/DABS(PL00))*DEXP(EXP0)	DK280590
16	RT(1)=X(1)+X(2)-C(2)/3.000	DK280600
	RT(2)=-.5000*(X(1)+X(2))-C(2)/3.000	DK280610
	RT(3)=.866025403784438500*(X(1)-X(2))	DK280620
	RETURN	DK280630
17	RT(1)=2.000*X(1)	DK280640
	RT(2)=-X(1)+X(2)*1.73205080756887700	DK280650
	RT(3)=-X(1)-X(2)*1.73205080756887700	DK280660
199	DO 200 L=1,3	DK280670
200	RT(L)=RT(L)-C(2)/3.000	DK280680
18	RETURN	DK280690
	END	DK280700
	FOR DECK29,DECK29	DK290000
	SUBROUTINE DDFER1 (SC)	DK290010
	DIMENSION C(8), A(4), Y(3)	DK290020
	DIMENSION SC(8)	DK290030
C	THE FOLLOWING STATEMENT(S) HAVE BEEN MANUFACTURED BY THE TRANSLATOR---	DK290040
C		DK290050
	DOUBLE PRECISION C, A, Y, CLOUD, PLOW	DK290060
	DOUBLE PRECISION P, Q, K, U	DK290070
	DOUBLE PRECISION BIGA, DSQRT, UTSC1, DISC2, ABSZ	DK290080
	DOUBLE PRECISION CORRECT, PHI, DATAN2, DCOS, USIN	DK290090
	DOUBLE PRECISION ZAP, ZSQRE1, ZSQRE2	DK290100
C	DIMENSION C(8), A(4), Y(3)	DK290110
C	DIMENSION SC(8)	DK290120
	DO 400 L=1,5	DK290130
400	C(L)=SC(L)	DK290140
	DO 401 L=9,16	DK290150
401	C(L)=0.0	DK290160
	A(1)=1.000	DK290170
	CLOUD=.25000*C(2)/C(1)	DK290180
14	IF (C(2)) 100, 101, 100	DK290190
100	PL00=C(1)**2	DK290200
	P=C(3)/C(1)-.37500*(C(2)**2)/PLOW	DK290210
	Q=C(4)/C(1)-.500*(C(3)*C(2))/(PLOW)+.12500*((C(2)/C(1))**3)	DK290220
	K=((C(2)**2)*C(3))/(16.000*(C(1)**3))-(3.000*(C(2)**4))/(256.000*(C(1)**4))-	DK290230
	.2500*(C(4)*C(2))/PLOW+C(5)/C(1)	DK290240
102	A(2)=-P	DK290250
	A(3)=-4.000*K	DK290260
	A(4)=4.000*P*R-Q**2	DK290270
	CALL DDVE1A (A,Y, MTYPE)	DK290280
	IF (MTYPE) 15, 15, 16	DK290290
15	U=Y(1)	DK290300
	DO 666 L=2,3	DK290310
	IF (U-Y(L)) 76, 666, 666	DK290320
76	U=Y(L)	DK290330
666	CONTINUE	DK290340
	GO TO 21	DK290350
16	IF (Y(2)) 270, 271, 270	DK290360
271	IF (ABS(Y(3))-10.0**(-7)) 272, 17, 17	DK290370

272	Y(2) = 0.000	DK290380
	Y(3)=0.000	DK290390
	GO TO 15	DK290400
270	PL0D = DAMS(Y(3)/Y(2))-5.0D-7	DK290410
	IF (PL0D) 18, 18, 17	DK290420
17	U=Y(1)	DK290430
	GO TO 21	DK290440
18	PL0D=Y(1)-Y(2)	DK290450
	IF (PL0D) 20, 17, 17	DK290460
20	U=Y(2)	DK290470
21	PL0D=U-P	DK290480
	IF (ABS(PL0D)-10.0**(-10)) 200, 150, 150	DK290490
150	BIGA=DSQRT(PL0D)	DK290500
	DISC1=BIGA**2-2.0DU*U+2.0D0*W/BIGA	DK290510
	DISC2=DISC1-4.0DU*W/BIGA	DK290520
	IF (ABS(DISC1)-10.0**(-15)) 204, 205, 205	DK290530
204	DISC1=0.000	DK290540
205	IF (DISC1) 24, 22, 23	DK290550
22	C (2)=0.000	DK290560
	C (1)=-.50D0*BIGA	DK290570
	C (3)=C(1)	DK290580
	C (4)=0.000	DK290590
	GO TO 25	DK290600
23	DISC1=DSQRT(DISC1)	DK290610
	C (1)=.50DU*(DISC1-BIGA)	DK290620
	C (2)=0.000	DK290630
	C (4)=0.000	DK290640
	C (3)= C(1)-DISC1	DK290650
	GO TO 25	DK290660
24	C(1)=-.50D0*BIGA	DK290670
	C (3)= C (1)	DK290680
	PL0D=DAMS(DISC1)	DK290690
	C(2) =.50DU*DSQRT(PL0D)	DK290700
	C(4)=-C(2)	DK290710
25	IF (ABS(DISC2)-10.0**(-15)) 210, 211, 211	DK290720
210	DISC2=0.000	DK290730
211	IF (DISC2) 28, 20, 27	DK290740
26	C (6)=0.000	DK290750
	C (8)= C (6)	DK290760
	C (5)=.50DU*BIGA	DK290770
	C (7)= C (5)	DK290780
	GO TO 29	DK290790
27	DISC2=DSQRT(DISC2)	DK290800
	C (6)=0.000	DK290810
	C (5)=.50DU*(DISC2+BIGA)	DK290820
	C (8)=0.000	DK290830
	C (7)= C (5)-DISC2	DK290840
	GO TO 29	DK290850
28	C (5)=.50DU*BIGA	DK290860
	C (7)= C (5)	DK290870
	PL0D=DAMS(DISC2)	DK290880
	C (6)=.50DU*DSQRT(PL0D)	DK290890
	C(8)=-C(6)	DK290900
29	DO 30 L=1,4	DK290910
	C (2*L-1)= C (2*L-1)-CL0D	DK290920
30	CONTINUE	DK290930
	DO 166 L=1,8	DK290940
166	SC(L)=C(L)	DK290950
	RETURN	DK290960
101	P=C(3) /C(1)	DK290970

Q=C(4) /C(1)	DK290980
R=C(5) /C(1)	DK290990
GO TO 102	DK291000
200 PL00=P**2-4.000*K	DK291010
IF (ABS(PL00)-10.0**(-15)) 201, 202, 202	DK291020
201 PL00=0.000	DK291030
202 IF (PL00) 203, 300, 300	DK291040
203 ABSZ=DSQRT(P**2-PL00)/2.000	DK291050
CORRECT=-P	DK291060
PH1 =DATAN2(DSQRT(-PL00),CORRECT)	DK291070
C(1)=ABSZ*DCOS(PH1)	DK291080
C(2)=ABSZ*DSIN(PH1)	DK291090
C(3)= C(1)	DK291100
C(4)=-C(2)	DK291110
C(5)=-C(1)	DK291120
C(6)= C(2)	DK291130
C(7)= C(5)	DK291140
C(8)=-C(2)	DK291150
GO TO 29	DK291160
300 ZAP=DSQRT(PL00)	DK291170
ZSQRE1=.5000*(-P+ZAP)	DK291180
ZSQRE2=ZSQRE1-ZAP	DK291190
IF (ZSQRE1) 301, 302, 302	DK291200
301 C(2)=DSQRT(-ZSQRE1)	DK291210
C(4)=-C(2)	DK291220
C(1)=0.0000	DK291230
C(3)=0.0000	DK291240
GO TO 310	DK291250
302 C(1)=DSQRT(ZSQRE1)	DK291260
C(3)=-C(1)	DK291270
C(2)=0.0000	DK291280
C(4)=0.0000	DK291290
310 IF (ZSQRE2) 303, 304, 304	DK291300
303 C(6)=DSQRT(-ZSQRE2)	DK291310
C(8)=-C(6)	DK291320
C(5)=0.0000	DK291330
C(7)=0.0000	DK291340
GO TO 29	DK291350
304 C(5)=DSQRT(ZSQRE2)	DK291360
C(7)=-C(5)	DK291370
C(6)=0.0000	DK291380
C(8)=0.0000	DK291390
GO TO 29	DK291400
END	DK291410
FOR DECK30,DECK30	DK300000
SUBROUTINE TABLE(DATA,MX,MM)	DK300010
C MM IS NUMBER OF COATING MATERIALS(OR SUBSTRATE)	DK300020
C DATA IS RU FOR RHO, SP FOR SPEC. HEAT, AND EEE FOR EPSILON	DK300030
C MX=0 DESIGNATES SUBSTRATE TABLE	DK300040
C MX=1 DESIGNATES COATING TABLE	DK300050
DIMENSION DATA(8,42)	DK300060
M=MM	DK300070
K1=1	DK300080
KL=K1+5	DK300090
110 READ(5,201) (DATA(M,K),K=K1,KL),CODE1,COUF2	DK300100
IF(DATA(M,KL-5)-10000.0)111,115,115	DK300110
111 IF(DATA(M,KL-3)-10000.0)112,116,116	DK300120
112 IF(DATA(M,KL-1)-10000.0)113,117,117	DK300130

113	IF (KL-42) 114, 118, 121	DK300140
114	K1=K1+6	DK300150
	KL=K1+5	DK300160
	GO TO 110	DK300170
115	DATA(M,42)=DATA(M,KL-4)	DK300180
	NK=KL-3	DK300190
	GO TO 118	DK300200
116	DATA(M,42)=DATA(M,KL-2)	DK300210
	NK=KL-1	DK300220
	GO TO 118	DK300230
117	DATA(M,42)=DATA(M,KL)	DK300240
	NK=KL+1	DK300250
118	DATA(M,41)=10000.0	DK300260
	IF (MX) 121, 119, 120	DK300270
119	WRITE (6,202) M, CODE1, CODE2	DK300280
	GO TO 122	DK300290
120	WRITE (6,203) M, CODE1, CODE2	DK300300
	GO TO 122	DK300310
121	WRITE (6,204)	DK300320
122	CONTINUE	DK300330
	WRITE (6,206) (DATA(M,J), J=1,NK)	DK300340
	RETURN	DK300350
201	FORMAT(6E12.8,A6,A2)	DK300360
202	FORMAT(20H0SUBSTRATE MATERIAL I2,4H IS A6,A2)	DK300370
203	FORMAT(18H0COATING MATERIAL I2,4H IS A6,A2)	DK300380
204	FORMAT(20H ERROR IN INPUT DATA)	DK300390
206	FORMAT(1P6E16.7)	DK300400
	END	DK300410
	FOR DECK31, DECK31	DK310000
	SUBROUTINE MAIN2	DK310010
CPLTP	IS THE MAIN PROGRAM OF LINK 2 (PLUT ROUTINES)	DK310020
	DIMENSION TRASH(17), F(10,9,42), HIAB(9), ANGTA(10), WH(18), XH(9),	DK310030
	1YH(6), ZH(6), ELL(9), RPP(9), RR(9), CAYY(9), TOME(2)	DK310040
	DIMENSION DT(2,200), T(2,200), SINL(200), COSL(200), SINO(200),	DK310050
	1COS0(200), THICK(200), NCOAT(200), NSURS(200), COSRS(200), PHIT(200),	DK310060
	2GAMM(200), NDUTY(200)	DK310070
	DIMENSION ESUN(8), EEE(8,42), RO(8,42), SP(8,42), TQINT(41,8)	DK310080
	DIMENSION BUFFER(2)	DK310090
	COMMON TRASH, F, HIAB, ANGTA, WH, XH, YH, ZH, ELL, RPP, RR, CAYY, PI, PIH,	DK310100
	1IWUP1, PI180, NOFINU, NQORT, IFIRST, NEWSIG	DK310110
	COMMON KPLNET, NORIEN, KTEMP, NUMRUN, NSATP, NPRINT, KRFV, NP50, REV	DK310120
	COMMON A, B, C, AYE, BEE, RPRN, PELF, EL, R, CAY, BAKL, S, ALP2, BFT2, GAM2,	DK310130
	1ALPHA2, BETA2, GAMMA2, COSA, COSB, COSG, SINB, PHIMAX	DK310140
	COMMON SIGMA, CSIGMA, SSIGMA, TSIGMA	DK310150
	COMMON PHIZ2, DPH12, PHIZ, DPH1, PHI, CPH1, SPHI, PHIN2, PHO12, SUN	DK310160
	COMMON TIMEZ, TABS, IELAPS, ZEIT, TOME, DELTAT, XP, YP, DEE, DPS0, J1, J2	DK310170
	COMMON EIP4, EPS1G2, TM, FTM, SAS2, SRASH	DK310180
	COMMON G, RHRC, RH0, CP, EPSLN, ITK, KITEK	DK310190
	COMMON QNET, QSAT, QINT, QEXT, QPLAN, QALB, QSUN, GOLD, QNEW, TBREAK	DK310200
	COMMON GAM, PHIC, ALI, ALTI, ANG, CTHEI, FD, FF	DK310210
	COMMON PH11, PH12, ISIG, FUDGE, T4, JUDGE, RUFFFR, RV, NLINE	DK310220
	COMMON COSLS, SINLS, DT, T, SINL, COSL, SINO, COS0, THICK, NCOAT, NSURS,	DK310230
	1COSRS, PHIT, GAMM	DK310240
	COMMON ESUN, EEE, RO, SP, TQINT, NDUTY, ANINCL, ASCNOD, ASNLNG, RGTASC,	DK310250
	1UECLIN	DK310260
	DIMENSION AA(6), AA1(6), P(6), PO(6)	DK310270
	COMMON AA, AA1, P, PO, IORDER, IORD1, IERROR, THETA, DIMAX, EN1, EN, FACT,	DK310280
	1YNHAI, ENHATL, EMAG, DEERROR, DTTEST	DK310290

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COMMON HSUN,HALB,HPLAN,NODE
DIMENSION HSUN(200),HALB(200),HPLAN(200),NODE(200)
COMMON KE1CH
COMMON HASUN,HAALB,HAPLN,HATOT
DIMENSION HASUN(200),HAALB(200),HAPLN(200),HATOT(200)
COMMON ZAREA
DIMENSION ZAREA(200)
COMMON IMTHRU
DIMENSION IMTHRU(200)
COMMON IMHI,PNAME,PHIPLT,TIMPLT,NPLDT,LAST,JUMP,LMAX,IONCE
DIMENSION PNAME(39),PHIPLT(190),TIMPLT(190)
COMMON TYME1,TYME2,TYME
COMMON AG,IN,J,KABG,KL,K,LFJ,LL,LN,L,LSHAUF,M,NCAKD,
1NEWDC,NEWGAM,NEWMAT,NHEAD,N,NTRIG,PEE1,PG,PIN,POUT,POP,SIGMA2
COMMON ELAMB,OMEGA,TKALT,TZ,W,AGNM,PGNM
DIMENSION ELAMB(200),OMEGA(200),NZ(41),TKALT(9),TZ(200),W(7),Z(41)
EQUIVALENCE (TRASH(15),C1),(TRASH(16),C2),(TRASH(17),C3)
EQUIVALENCE (TRASH(14),NBLANK),(Z(1),NZ(1))
DIMENSION S1(190),S2(190),A1(190),A2(190),P1(190),P2(190),Q1(190)
1,Q2(190),T1(190),T2(190),TX(1,200),PHIO(190)
DIMENSION AS(200),BKITE(200),DARK(200),XLAMB(200),XMEGA(200)
DIMENSION THATS(16),ALL(20),FOLKS(13)
COMMON/HLJK/160,JJ,KK,KUN,NFIRST,NALL,ALL1,NALL1
CALL RESET
NJ= 2*KPLNET
NK= 3*(NORIFN+2)
NL= 3*KTEMP +3
THATS( 1)=ZH(1)
THATS( 2)=WH(NJ-1)
THATS( 3)=WH(NJ)
THATS( 4)=WH(4)
THATS( 5)=ZH(2)
THATS( 6)=ZH(3)
THATS( 7)=XH(NK-2)
THATS( 8)=XH(NK-1)
THATS( 9)=XH(NK)
THATS(10)=WH(4)
THATS(11)=ZH(4)
THATS(12)=ZH(5)
THATS(13)=ZH(6)
THATS(14)=YH(NL-2)
THATS(15)=YH(NL-1)
THATS(16)=YH(NL)
CALL RINDEC (AGNM ,NALL1,ALL1,ADUM)
ALL1=FUTA(ALL1)
1F(NALL1-5)48,49,49
48 NALL1=NALL1+1
DATA QU00HL/6H =/
49 ALL( 1)=QU00HL
DATA QU01HL/6HMAX(AL/
ALL( 2)=QU01HL
DATA QU02HL/6HT)MIN=/
ALL( 3)=QU02HL
CALL RINDEC (PGNM ,NALL,ALL( 4),ADUM)
ALL(4)=FUTA(ALL(4))
DATA QU03HL/6H PHIO=/
ALL( 5)=QU03HL
CALL RINDEC (PHI2 ,NALL,ALL( 6),ADUM)
ALL(6)=FUTA(ALL(6))
DATA QU04HL/6H DPHI=/

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DK310300
DK310310
DK310320
DK310330
DK310340
DK310350
DK310360
DK310370
DK310380
DK310390
DK310400
DK310410
DK310420
DK310430
DK310440
DK310450
DK310460
DK310470
DK310480
DK310490
DK310500
DK310510
DK310520
DK310530
DK310540
DK310550
DK310560
DK310570
DK310580
DK310590
DK310600
DK310610
DK310620
DK310630
DK310640
DK310650
DK310660
DK310670
DK310680
DK310690
DK310700
DK310710
DK310720
DK310730
DK310740
DK310750
DK310760
DK310770
DK310780
DK310790
DK310800
DK310810
DK310820
DK310830
DK310840
DK310850
DK310860
DK310870
DK310880
DK310890

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ALL(7)=Q004HL	DK310900
CALL BINDEC (DPH12 ,NALL,ALL(8),ADUM)	DK310910
ALL(8)=FDTA(ALL(8))	DK310920
DATA Q005HL/6H SIGM=	DK310930
ALL(9)=Q005HL	DK310940
CALL BINDEC (SIGMA2,NALL,ALL(10),ADUM)	DK310950
ALL(10)=FDTA(ALL(10))	DK310960
DATA Q006HL/6H BETA=	DK310970
ALL(11)=Q006HL	DK310980
CALL BINDEC (BET2 ,NALL,ALL(12),ADUM)	DK310990
ALL(12)=FDTA(ALL(12))	DK311000
DATA Q007HL/6H PHIN=	DK311010
ALL(13)=Q007HL	DK311020
CALL BINDEC (PIN ,NALL,ALL(14),ADUM)	DK311030
ALL(14)=FDTA(ALL(14))	DK311040
DATA Q008HL/6H POU=	DK311050
ALL(15)=Q008HL	DK311060
CALL BINDEC (POU1 ,NALL,ALL(16),ADUM)	DK311070
ALL(16)=FDTA(ALL(16))	DK311080
IF (KARG)65,60,65	DK311090
DATA Q009HL/6H ALPH=	DK311100
60 ALL(17)=Q009HL	DK311110
CALL BINDEC (ALP2 ,NALL,ALL(18),ADUM)	DK311120
ALL(18)=FDTA(ALL(18))	DK311130
ALL(19)=Q010HL	DK311140
DATA Q010HL/6H GAMM=	DK311150
CALL BINDEC (GAM2 ,NALL,ALL(20),ADUM)	DK311160
ALL(20)=FDTA(ALL(20))	DK311170
NALL=120	DK311180
GO TO 66	DK311190
65 NALL=96	DK311200
DATA Q011HL/6H INCL.=	DK311210
FOLKS(1)=Q011HL	DK311220
CALL BINDEC (ANINCL,NF,FOLKS(2),ADUM)	DK311230
FOLKS(2)=FDTA(FOLKS(2))	DK311240
DATA Q012HL/6H ARG 0/	DK311250
FOLKS(3)=Q012HL	DK311260
DATA Q013HL/6H PER=	DK311270
FOLKS(4)=Q013HL	DK311280
CALL BINDEC (ASCND,NF,FOLKS(5),ADUM)	DK311290
FOLKS(5)=FDTA(FOLKS(5))	DK311300
DATA Q014HL/6H LUNG./	DK311310
FOLKS(6)=Q014HL	DK311320
DATA Q015HL/6H OF ASC/	DK311330
FOLKS(7)=Q015HL	DK311340
DATA Q016HL/6H NUDE=	DK311350
FOLKS(8)=Q016HL	DK311360
CALL BINDEC (ASNLNG,NF,FOLKS(9),ADUM)	DK311370
FOLKS(9)=FDTA(FOLKS(9))	DK311380
DATA Q017HL/6H KA.=	DK311390
FOLKS(10)=Q017HL	DK311400
CALL BINDEC (KGTASC,NF,FOLKS(11),ADUM)	DK311410
FOLKS(11)=FDTA(FOLKS(11))	DK311420
DATA Q018HL/6H DEC.=	DK311430
FOLKS(12)=Q018HL	DK311440
CALL BINDEC (UFCLIN,NF,FOLKS(13),ADUM)	DK311450
FOLKS(13)=FDTA(FOLKS(13))	DK311460
66 CONTINUE	DK311470
*** RETRIEVE ABSORPTIVITIES (SOLAR AND AVERAGE PLANET)	DK311480
50 DO 51 I=1,NSATP	DK311490

XLAMB(I)=ELAMB(I)*57.29578	DK311500
XMEGA(I)=OMEGA(I)*57.29578	DK311510
JC= NCOAT(I)	DK311520
AS(I)=ESUN(JC)	DK311530
BRITE(I)= HALB(I)	DK311540
51 DAKK(I)= HSUN(I)	DK311550
JT=LAST	DK311560
RUN=NUMRUN	DK311570
10 IF (NQORT*NQORT -2*NQORT)12,13,12	DK311580
13 IGO= -1	DK311590
GO TO 15	DK311600
12 IGO= 0	DK311610
15 CONTINUE	DK311620
DO 19 II=1,LMAX	DK311630
19 PHIO(II)=PHIPLT(II)	DK311640
*** PHIZ2 IS THE INITIAL TRUE ANOMALY VALUE	DK311650
IF(PHIZ2=.01)31,31,32	DK311660
32 NSWTCH=1	DK311670
GO TO 18	DK311680
31 NSWTCH=0	DK311690
18 CONTINUE	DK311700
NFIRST=0	DK311710
N30=0	DK311720
DO 5 I=1,NSATP, 2	DK311730
REWIND JT	DK311740
JJ= I	DK311750
KK=I+1	DK311760
DO 20 L=1,LMAX	DK311770
*** READ BINARY SCRAICH TAPE JT TO OBTAIN INFORMATION RECORDED BY	DK311780
*** SUBROUTINE LOOP IN LINK 1	DK311790
*** INFORMATION WILL BE RETRIEVED TWO(2) NODES PER PASS	DK311800
READ (JT)(TX(1,J),HSUN(J),HALB(J),HPLAN(J),HASUN(J),HAALB(J), HAP	DK311810
1LN(J),HATOT(J),J=1,NSATP)	DK311820
IF(IGO) 25,30,30	DK311830
*** RETRIEVE INCIDENT HEATS	DK311840
25 S1(L)=HSUN(JJ)	DK311850
S2(L)=HSUN(KK)	DK311860
A1(L)=HALB(JJ)	DK311870
A2(L)=HALB(KK)	DK311880
P1(L)=HPLAN(JJ)	DK311890
P2(L)=HPLAN(KK)	DK311900
GO TO 16	DK311910
*** RETRIEVE ABSORBED HEATS	DK311920
30 S1(L)=HASUN(JJ)	DK311930
S2(L)=HASUN(KK)	DK311940
A1(L)=HAALB(JJ)	DK311950
A2(L)=HAALB(KK)	DK311960
P1(L)=HAPLN(JJ)	DK311970
P2(L)=HAPLN(KK)	DK311980
Q1(L)=HATOT(JJ)	DK311990
Q2(L)=HATOT(KK)	DK312000
16 IF(NQORT*NQORT-3*NQORT)17,20,17	DK312010
17 I1(L)= TX(1,JJ)	DK312020
I2(L)= TX(1,KK)	DK312030
N30=1	DK312040
20 CONTINUE	DK312050
IF(N30)40,40,35	DK312060
35 IF(NSWTCH)40,40,45	DK312070
*** ARRANGE TRUE ANOMALY ARRAY (PHIPLT) IN ASCENDING ORDER IF NOT	DK312080
ALREADY IN SUCH FORM (WILL NOT BE ASCENDING WHEN PHIZ NOT =0.0)	DK312090

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45 CALL ACCEND(PHIO(1),PHIPLT(1),T1(1),I2(1),LMAX)          DK312100
40 CONTINUE                                                  DK312110
** SUBROUTINE DRAW CONSTRUCTS IDENTIFIED PLOTS OF HEATS      DK312120
** AND/OR TEMPERATURES                                       DK312130
** NODES ARE PLOTTED TWO (2) AT A TIME                      DK312140
CALL DRAW(NODE,PHIPLT,TIMPLT,S1,S2,A1,A2,P1,P2,Q1,Q2,T1,T2,  DK312150
1PHIO,PNAME,IMTHRU,XLAMB,XMEGA,AS,BRITE,DARK,ZAREA,THATS,ALL,FOLKS,DK312160
2LMAX,KETCH,NWORT,NSATP,KABG)                                DK312170
5 CONTINUE                                                    DK312180
JUMP=2                                                        DK312190
CALL FILMAV(1)                                                DK312200
CALL CLOCK(TIME2)                                              DK312210
** RETURN TO PILOT IN LINK 1 WITH JUMP =2                    DK312220
RETURN                                                        DK312230
END                                                            DK312240

FOR DECK32,DECK32                                             DK320000
SUBROUTINE DRAW(NODE,PHIPLT,TIMPLT,S1,S2,A1,A2,P1,P2,Q1,Q2,T1,T2, DK320010
1PHIO,PNAME,IMTHRU,XLAMB,XMEGA,AS,BRITE,DARK,ZAREA,THATS,ALL,FOLKS,DK320020
2LMAX,KETCH,NWORT,NSATP,KABG)                                DK320030
DIMENSION T1(190),I2(190),TELL( 7),TELL2( 7),RL(9)          DK320040
DIMENSION NODE(200),IMTHRU(200),PHIPLT(190),TIMPLT(190),S1(190), DK320050
IS2(190),A1(190),A2(190),P1(190),P2(190),Q1(190),Q2(190)    DK320060
DIMENSION TELL(7),PNAME(39)                                   DK320070
DIMENSION S(190),A(190),P(190),Q(190)                       DK320080
DIMENSION XNAME1(13),XNAME2(13),XNAME3(13)                   DK320090
DIMENSION TOP(4),BOT(4),YL(9),TL(9),PL(9)                    DK320100
DIMENSION YNAM (5),TANOM(2),YTEMP(4)                         DK320110
DIMENSION PHIO(190),TAN(9),LOC(9),NC(9),BCD(9)               DK320120
DIMENSION AS(200),BRITE(200),DARK(200),ZAREA(200)           DK320130
DIMENSION ACKOSS(19),ELAMB(200),OMEGA(200)                   DK320140
DIMENSION THATS(16),ALL(20),FOLKS(13)                        DK320150
COMMON/BLJK/160,JJ,KK,RUN,NFIRST,NALL,ALL1,NALL1            DK320160
NONE=NFIRST                                                    DK320170
IF (NFIRST) 2,2,1                                             DK320180
** THE INSTRUCTIONS FROM HERE TO STATEMENT NO. 1 ARE REACHED ONCE DK320190
IN EACH CASE
2 NFIRST =1                                                    DK320200
DO 3 I=1,13                                                    DK320210
3 XNAME1(I)=PNAME(I)                                           DK320220
DO 4 I=1,13                                                    DK320230
4 XNAME2(I)=PNAME(I+13)                                         DK320240
DO 5 I=1,13                                                    DK320250
5 XNAME3(I)=PNAME(I+26)                                         DK320260
** SET UP RCU SENTENCES TO BE WRITTEN AS PLOT IDENTIFICATION DK320270
DAIA Q00UHL/6HTEMPER/                                         DK320280
YTEMP(1)=Q00UHL                                               DK320290
DAIA Q001HL/6HATURE /                                         DK320300
YTEMP(2)=Q001HL                                               DK320310
DAIA Q002HL/6HDEG. R/                                         DK320320
YTEMP(3)=Q002HL                                               DK320330
DAIA Q003HL/6HANKINE/                                         DK320340
YTEMP(4)=Q003HL                                               DK320350
DAIA Q00UCT/0310506232422/                                     DK320360
IANOM(1)=Q00UCT                                                DK320370
DAIA Q001CT/0750505050505/                                     DK320380
IANOM(2)=Q001CT                                                DK320390
DAIA Q004HL/6HNODE N/                                         DK320400
TELL(1)=Q004HL                                                 DK320410

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DATA Q005HL/6HNUMBER /	DK320430
TELL(2)=Q005HL	DK320440
DATA Q006HL/6HCASE N/	DK320450
TELL(4)=Q006HL	DK320460
TELL(5)=Q005HL	DK320470
DATA Q007HL/6H L/	DK320480
ACKROSS(1)=Q007HL	DK320490
DATA Q008HL/6HAMBDA=	DK320500
ACKROSS(2)=Q008HL	DK320510
DATA Q009HL/6H OMEGA=	DK320520
ACKROSS(4)=Q009HL	DK320530
IF(NQORT=3)70,71,71	DK320540
DATA Q010HL/6H PLANE/	DK320550
71 ACKROSS(6)=Q010HL	DK320560
DATA Q011HL/6HT ABS(/	DK320570
ACKROSS(7)=Q011HL	DK320580
DATA Q012HL/6HSUN) =/	DK320590
ACKROSS(8)=Q012HL	DK320600
DATA Q013HL/6H PLNET/	DK320610
ACKROSS(10)=Q013HL	DK320620
DATA Q014HL/6H ABS(S/	DK320630
ACKROSS(11)=Q014HL	DK320640
DATA Q015HL/6HHAUE)=/	DK320650
ACKROSS(12)=Q015HL	DK320660
DATA Q016HL/6H SOLAR/	DK320670
ACKROSS(14)=Q016HL	DK320680
DATA Q017HL/6H ABS =/	DK320690
ACKROSS(15)=Q017HL	DK320700
DATA Q018HL/6H SURFA/	DK320710
ACKROSS(17)=Q018HL	DK320720
DATA Q019HL/6HCE A.=/	DK320730
ACKROSS(18)=Q019HL	DK320740
70 NTUP=30	DK320750
** CONVERT RUN NO. TO BCD	DK320760
CALL BINDEC (RUN,NCRUN,TELL(6),DUMMY)	DK320770
TELL(6)=FUTA(TELL(6))	DK320780
IF(NQORT* NQORT -2*NQORT)25,24,25	DK320790
DATA Q020HL/6HARSONB/	DK320800
25 YNAM(1) = Q020HL	DK320810
DATA Q021HL/6HEID /	DK320820
YNAM(2) = Q021HL	DK320830
GO TO 26	DK320840
DATA Q022HL/6HINCIDE/	DK320850
24 YNAM(1) =Q022HL	DK320860
DATA Q023HL/6HNT /	DK320870
YNAM(2) =Q023HL	DK320880
DATA Q024HL/6H HEAT, /	DK320890
26 YNAM(3) =Q024HL	DK320900
DATA Q025HL/6HBTU/HR/	DK320910
YNAM(4)=Q025HL	DK320920
IF (KETCH)27,28,27	DK320930
DATA Q026HL/6H FI**2/	DK320940
28 YNAM(5) =Q026HL	DK320950
GO TO 29	DK320960
DATA Q027HL/6H...../	DK320970
27 YNAM(5) =Q027HL	DK320980
29 CONTINUE	DK320990
XM1N=0.0	DK321000
XMAL=TIMPLT(LMAX)	DK321010
IL(1)=TIMPLT(1)	DK321020

TL(9)=XMAX	DK321030
DEL=(XMAX-XMIN)/8.0	DK321040
TL(2)=DEL	DK321050
DO 6 I=3,8	DK321060
IX=1-1	DK321070
6 TL(I)=TL(IX)+DEL	DK321080
*** SCALE DETERMINES A SUITABLE MAX,MIN, AND 7 VALUES IN BETWEEN	DK321090
CALL SCALE (PHIPLT(LMAX),PHIPLT(1),PL(1),PL(2),PL(3),PL(4),PL(5),	DK321100
1PL(6),PL(7),PL(8),PL(9))	DK321110
PMIN= PL(1)	DK321120
PMAX= PL(9)	DK321130
1 IF(NQORT-1)11,100,11	DK321140
11 CONTINUE	DK321150
*** BEGIN PLOT SCHEME , STARTING WITH ELEMENT JJ	DK321160
IFIRST=1	DK321170
INODE=JJ	DK321180
DO 10 I=1,LMAX	DK321190
S(I)=S1(I)	DK321200
A(I)=A1(I)	DK321210
P(I)=P1(I)	DK321220
10 Q(I)=Q1(I)	DK321230
15 CALL RSET(1)	DK321240
*** DETERMINE LIMITS OF HEAT ARRAYS	DK321250
CALL HILOW(S(1),LMAX,TOP(1),BOT(1))	DK321260
CALL HILOW(A(1),LMAX,TOP(2),BOT(2))	DK321270
CALL HILOW(P(1),LMAX,TOP(3),BOT(3))	DK321280
IF(IGO)16,17,17	DK321290
17 CALL HILOW(Q(1),LMAX,QMAX,BOT(4))	DK321300
CALL HILOW(ROT(1),4,DUM,QMIN)	DK321310
GO TO 18	DK321320
16 BOT(4)=10000.0	DK321330
TOP(4)=0.0	DK321340
CALL HILOW(TOP(1),4,QMAX,DUM)	DK321350
CALL HILOW(ROT(1),4,DUM,QMIN)	DK321360
18 CONTINUE	DK321370
*** SCALE DETERMINES A SUITABLE MAX,MIN, AND 7 VALUES IN BETWEEN	DK321380
CALL SCALE(QMAX,QMIN,YL(1),YL(2),YL(3),YL(4),YL(5),YL(6),YL(7),	DK321390
1 YL(8),YL(9))	DK321400
QMIN=YL(1)	DK321410
QMAX=YL(9)	DK321420
*** GENERATE GRID AND PLOT HEATS	DK321430
CALL GRIDGN (63,1023,0,960,12,12,10,10)	DK321440
CALL PLOT11 (1,1,XMIN,XMAX,QMIN,QMAX,TIMPLT(1),S(1),LMAX,1,1HS)	DK321450
CALL PLOT11 (1,1,XMIN,XMAX,QMIN,QMAX,TIMPLT(1),A(1),LMAX,1,1HA)	DK321460
CALL PLOT11 (1,1,XMIN,XMAX,QMIN,QMAX,TIMPLT(1),P(1),LMAX,1,1HP)	DK321470
IF(IGO)30,20,20	DK321480
20 CALL PLOT11 (1,1,XMIN,XMAX,QMIN,QMAX,TIMPLT(1),Q(1),LMAX,1,1HQ)	DK321490
30 IF(NONE)93,94,93	DK321500
94 DO 7 I=1,9	DK321510
*** FIND TRUE ANOMALY (TAN(I)) CORRESPONDING TO TIME TL(1)	DK321520
CALL XINIRP(LMAX,TIMPLT(1),TL(I),PHIU(1),TAN(I))	DK321530
*** CONVERT EACH TAN(I) (TRUE ANOMALY) TO BCU EQUIVALENT	DK321540
IF(TAN(I))44,43,44	DK321550
DATA Q002CT/005050505050505/	DK321560
43 BCU(1)=Q002C1	DK321570
GO TO 8	DK321580
44 CALL RINDEC(IAN(1),NC(I),BCD(I),DUMMY)	DK321590
BCU(I)=FDTA(BCD(1))	DK321600
DUMMY=FDIA(DUMMY)	DK321610
8 NC(1)=0	DK321620

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*** FIND LOCATION OF TL(I) IN RASTER COUNTS
LTEMP=SCALEX(TL(I),1)
LOC(I)=LTEMP-21
7 CONTINUE
LOC(9)=LOC(9)-22
*** PUT LABELS AND IDENTIFICATION ON THE PRESENT FRAME
93 DO 95 I=1,9
CALL PRINT(LOC(I),981,8,0,NC(I),BCU(I))
CALL LABELY(YL(I),1,0)
95 CALL LABELX(TL(I),1,0)
CALL PRINT(63,1003, 8,0,78,XNAME1(1))
CALL PRINT(63,1013, 8,0,78,XNAME2(1))
CALL PRINT(63,1023, 8,0,78,XNAME3(1))
CALL PRINT( 1,360,0,12,30,YNAM(1))
CALL PRINT( 1,970,0,0,6,6,HMIN. )
CALL PRINT( 1,981,0,0,12,TANOM(1))
CALL BINDEC(ELAMB(INODE),NDUM,ACROSS( 3),DUM)
ACROSS(3)=FDTA(ACROSS(3))
DUM=FDTA(DUM)
CALL BINDEC(OMEGA(INODE),NDUM,ACROSS( 5),DUM)
ACROSS(5)=FDTA(ACROSS(5))
IF(NGORT-3)75,73,73
73 CALL BINDEC(BRITE(INODE),NDUM,ACROSS( 9),DUM)
ACROSS(9)=FDTA(ACROSS(9))
CALL BINDEC( DARK(INODE),NDUM,ACROSS(13),DUM)
ACROSS(13)=FDTA(ACROSS(13))
CALL BINDEC( AS(INODE),NDUM,ACROSS(16),DUM)
ACROSS(16)=FDTA(ACROSS(16))
NTOP=96
IF(ZAREA(INODE))74,75,74
74 CALL BINDEC(ZAREA(INODE),NDUM,ACROSS(19),DUM)
ACROSS(19)=FDTA(ACROSS(19))
NTOP=114
75 CONTINUE
CALL PRINT( 70, 8,8,0, NTOP,ACROSS(1))
CALL PRINT(138,20,8,0, 96,THATS(1))
CALL PRINT( 63,32,8,0, NALL,ALL(1))
CALL PRINT( 67,32,7,0, NALL1,ALL1 )
IF(KARG)/6,77,76
76 CALL PRINT(231,44,8,0, 78,FOLKS(1))
77 CONTINUE
IF(IFIRS1-1)54,54,55
54 XNODE=NODE(JJ)
GO TO 56
55 XNODE=NODE(KK)
C *** CONVERT NODE NO. TO BCU AND WRITE A SENTENCE INCLUDING SAME
56 CALL BINDEC (XNODE,NCNODE,TELL(3),DUMMY )
TELL(3)=FDTA(TELL(3))
NCHAR=12+ NCNODE
CALL PRINT( 63,992,12,0,NCHAR,TELL(1))
NCHAR=12+ NCKUN
CALL PRINT(280,992,12,0,NCHAR,TELL(4))
C *** TERMINATE THIS PLOT AND PROCEED
CALL DMPBUF
IF(IFIRS1-1)41,41,42
42 TELL2(4)=TELL(3)
GO TO 60
41 TELL1(4)=TELL(3)
DO 50 I=1,LMAX
S(I)=S2(I)

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DK321630
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DK322130
DK322140
DK322150
DK322160
DK322170
DK322180
DK322190
DK322200
DK322210
DK322220

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      A(I)=A2(I)
      P(I)=P2(I)
50  Q(I)=Q2(I)
      IFIRST = 2
      NONE=1
      INODE= KK
      IF(INODE=NSATP)15,15,60
      60 IF(NQORT*NGORT= 3*NGORT )100,102,100
      100 IF(IMTHRU(JJ)+IMTHRU(KK) )104,102,104
C *** PLOT STABILIZED TEMPERATURES OF ELEMENTS JJ AND KK IF REQUIRED
      104 CALL HILOW (T1(1),LMAX,T1MAX,T1MIN)
      CALL HILOW (T2(1),LMAX,T2MAX,T2MIN )
      TMAX= AMAX1(T1MAX,T2MAX)
      TMIN= AMIN1(T1MIN,T2MIN)
C *** SCALE DETERMINES A SUITABLE MAX,MIN, AND 7 VALUES IN BETWEEN
      CALL SCALE (TMAX,TMIN,KL(1),KL(2),KL(3),KL(4),KL(5),KL(6),KL(7),
      1KL(8),KL(9) )
C *** GENERATE GRID THEN PLOT STABILIZED TEMPERATURES AND IDENTIFY
      CALL RSEI(1)
      CALL GRIDGN (63,1023,0,960,12,12,10,10 )
      IF(IMTHRU(JJ))67,67,68
      68 CALL PLOI1 (1,1,PMIN,PMAX,KL(1),KL(9),PH1PLT(1),T1(1),LMAX,1,1H1 )
      67 IF(IMTHRU(KK))66,66,69
      69 IF(KK=NSATP)65,65,66
      65 CALL PLOI1 (1,1,PMIN,PMAX,KL(1),KL(9),PH1PLT(1),T2(1),LMAX,1,1H2 )
C *** TIDENT CONSTRUCTS SENTENCES TELL1 AND TELL2 WHICH DESCRIBE
C *** TEMPERATURE CURVES
      66 CALL TIDENT (JJ,KK,IMTHRU(1),TELL1(1),TELL2(1),NSATP,NCJJ,NCKK)
      DO 101 I=1,9
      CALL LABELY(KL(I),1,0)
      101 CALL LABELX(PL(I),1,0)
      NNC= 18+ NCNODE
      CALL PRINT( 63,981, 8,0,NNC,TELL1(1))
      NCJ= 6+ NCJJ
      CALL PRINT(255,981, 8,0,NCJ ,TELL1(5))
      CALL PRINT(351,981, 8,0, 6,TELL1(7))
      CALL PRINT( 63,992, 8,0,NNC,TELL2(1))
      NCK= 6+ NCKK
      CALL PRINT(255,992, 8,0,NCJ,TELL2(5))
      CALL PRINT(351,992, 8,0, 6,TELL2(7))
      CALL PRINT(63,1003, 8,0,78,XNAME1(1))
      CALL PRINT(63,1013, 8,0,78,XNAME2(1))
      CALL PRINT(63,1023, 8,0,78,XNAME3(1))
      CALL PRINT( 1,969,8,0,12,TANUM(1))
      CALL PRINT(10,365,8,12,24,YTEMP(1))
C *** TERMINATE THIS PLOT AND RETURN TO MAIN2 WHERE WE WILL OBTAIN
C *** INFORMATION PERTAINING TO THE NEXT TWO ELEMENTS
      CALL DMPBUF
      102 RETURN
      END
      '      FOR DECK33,DECK33
      SUBROUTINE SKALE(TOP,BOTTOM,A,B,C,D,E,F,G,H,XI)
CSKALE
C      TRUNCATE,ROUND AND FLOAT, INSERT 7 VALUES BETWEEN 2 GIVEN VALUES
C      TOP = GIVEN TOP VALUE(MAX)
C      BOTTOM = GIVEN BOTTOM VALUE(MIN)
C      A = SCALED MINIMUM
C      B = INSERTED VALUE 1
C      C = INSERTED VALUE 2
C      D = INSERTED VALUE 3

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C	E = INSERTED VALUE 4	DK330100
C	F,G,H, AND XI ARE INSERTED AT PROPER LOCATION	DK330110
	FINC=TOP-BOTTOM	DK330120
	AST = ALOG(FINC)/ALOG(10.0)	DK330130
	IF (AST) 5,10,10	DK330140
5	AST=AST-1.0	DK330150
10	NAST=AST	DK330160
	P10AST=10.0**NAST	DK330170
	FMAX=P10AST*(AINT(TOP/P10AST)+1.0)	DK330180
	FMIN=P10AST*AINT(BOTTOM/P10AST)	DK330190
	IF (BOTTOM) 15,20,20	DK330200
15	FMIN=FMIN-P10AST	DK330210
	IF (TOP) 17,20,20	DK330220
17	TOP = TOP - P10AST	DK330230
20	SC=(FMAX-FMIN)/8.0	DK330240
	A=FMIN	DK330250
	B=A+SC	DK330260
	C=B+SC	DK330270
	D=C+SC	DK330280
	E=D+SC	DK330290
	F=E+SC	DK330300
	G=F+SC	DK330310
	H=G+SC	DK330320
	XI=FMAX	DK330330
	RETURN	DK330340
	END	DK330350
	 FOR DECK34,DECK34	DK340000
	SUBROUTINE XINTRP(N,A,AA,B,BB)	DK340010
	DIMENSION A(182), B(182)	DK340020
	DO 100 I=1,N	DK340030
	K=1	DK340040
	IF (AA-A(I)) 1,2,100	DK340050
100	CONTINUE	DK340060
	BB=0.0	DK340070
	GO TO 5	DK340080
2	BB=B(K)	DK340090
	GO TO 5	DK340100
1	X=((AA-A(K-1))*(B(K)-B(K-1)))/(A(K)-A(K-1))	DK340110
	BB=B(K-1)+X	DK340120
5	RETURN	DK340130
	END	DK340140
	 FOR DECK35,DECK35	DK350000
	SUBROUTINE ACCEND(X0,X,Y,Z,N)	DK350010
C ***	ARRANGE X,Y, AND Z IN ORDER OF INCREASING X VALUES BUT KEEP	DK350020
	***ORIGINAL ORDER OF X ARKAY IN X0	DK350030
	DIMENSION X(1),Y(1),Z(1)	DK350040
	DIMENSION X0(1)	DK350050
	DO 104 J=1,N	DK350060
104	X(J)=X0(J)	DK350070
	K=1	DK350080
101	SMALL=X(K)	DK350090
	DO 100 I=K,N	DK350100
	DUMY=X(I)	DK350110
	SMALL=AMIN1(SMALL,DUMY)	DK350120
	IF (SMALL-X(I)) 100,102,100	DK350130
102	INDEX=I	DK350140

100 CONTINUE	DK350150
X(INDEX)=X(K)	DK350160
X(K)=SMALL	DK350170
SAVE=Y(K)	DK350180
Y(K)=Y(INDEX)	DK350190
Y(INDEX)=SAVE	DK350200
SAVEZ=Z(K)	DK350210
Z(K)=Z(INDEX)	DK350220
Z(INDEX)=SAVEZ	DK350230
K=K+1	DK350240
IF(K=N)101,103,101	DK350250
103 RETURN	DK350260
END	DK350270
FOR DECK36,DECK36	DK360000
SUBROUTINE HLOW(TABLE,NPTS,XH1,XL0)	DK360010
DIMENSION TABLE(1)	DK360020
XL0=TABLE(1)	DK360030
XH1=TABLE(1)	DK360040
DO 10 I=2,NPTS	DK360050
XL0=AMIN1(XL0,TABLE(I))	DK360060
10 XH1=AMAX1(XH1,TABLE(I))	DK360070
RETURN	DK360080
END	DK360090
FOR DECK37,DECK37	DK370000
SUBROUTINE TIDENI (JJ,KK,IMTHRU,TELL1,TELL2,NSATP,NCJJ,NCKK)	DK370010
*** TIDENI CONSTRUCTS SENTENCES TELL1 AND TELL2 WHICH DESCRIBE	DK370020
*** TEMPERATURE CURVES	DK370030
DIMENSION TELL1(7),TELL2(7),IMTHRU(200)	DK370040
DATA I1/0050505050505/	DK370050
DATA T5/6H CURVE /	DK370060
DATA T6/6H1 = EL/	DK370070
DATA T7/5HEMENT /	DK370080
DATA T8/5HAFIER /	DK370090
DATA T9/6HORBITS/	DK370100
DATA T10/6H2 = EL/	DK370110
IF(IMTHRU(JJ))10,10,20	DK370120
10 DO 15 I=1,7	DK370130
15 TELL1(I)=I1	DK370140
DO TO 30	DK370150
20 TELL1(1)=T5	DK370160
TELL1(2)=T6	DK370170
TELL1(3)=T7	DK370180
TELL1(5)=T8	DK370190
TELL1(7)=T9	DK370200
30 IF(IMTHRU(KK))40,40,50	DK370210
40 DO 45 I=1,7	DK370220
45 TELL2(I)=I1	DK370230
DO TO 60	DK370240
50 TELL2(1)=T5	DK370250
TELL2(2)=T10	DK370260
TELL2(3)=T7	DK370270
TELL2(5)=T8	DK370280
TELL2(7)=T9	DK370290
60 IF(IMTHRU(JJ)+IMTHRU(KK))61,70,61	DK370300
61 HORBIT=I1	DK370310
NELMT=JJ	DK370320

57 IF (IMTHRU(NELMT)) 63,63,62	DK370330
52 ORBIT=IM(HKU(NELMT))	DK370340
CALL BINDEC (ORBIT,NC,HORBIT,DUMMY)	DK370350
HORBIT=FUTA(HORBIT)	DK370360
53 IF (NELMT=JJ) 65,65,66	DK370370
55 TELL1(6)=HORBIT	DK370380
NCJJ=NC	DK370390
NELMT=KK	DK370400
HORBIT=I1	DK370410
GO TO 67	DK370420
6 TELL2(6)=HORBIT	DK370430
NCKK=NC	DK370440
70 RETURN	DK370450
END	DK370460
FOR DECK38,DECK38	DK380000
INTEGER FUNCTION FUTA(WORD)	DK380010
INTEGER WORD	DK380020
DATA MASK/00000000000077/	DK380030
J = 0	DK380040
DO 1 I=1,6	DK380050
N = WORD/2**(36-6*I)	DK380060
N = AND(N,MASK)	DK380070
IF (N.LE. 9) GO TO 2	DK380080
GO TO 5	DK380090
2 N = N + 48	DK380100
GO TO 6	DK380110
3 IF (N.EQ. 27) N= 61	DK380120
IF (N.EQ.32) N=33	DK380130
IF (N.EQ. 48) N=5	DK380140
4 J = OR(J*2**6,N)	DK380150
1 CONTINUE	DK380160
FUTA = J	DK380170
RETURN	DK380180
END	DK380190

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